

A1-F18AC-741-500

1 June 1992

Change 3 - 1 June 2002

TECHNICAL MANUAL

**ORGANIZATIONAL MAINTENANCE
SYSTEM SCHEMATICS**

MISSION COMPUTER SYSTEM

**NAVY MODEL
F/A-18A AND F/A-18B
161353 AND UP**

N68936-01-D-0007

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NATEC ELECTRONIC MANUAL

NUMERICAL INDEX OF EFFECTIVE WORK PACKAGES/PAGES

List of Current Changes

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Only those work packages/pages assigned to the manual are listed in this index. Insert Change 3, dated 1 June 2002. Dispose of superseded work packages/pages. Superseded classified work packages/pages shall be destroyed in accordance with applicable security regulations. If changed pages are issued to a work package, insert the changed pages in the applicable work package. The portion of text affected in a change or revision is indicated by change bars or the change symbol "R" in the outer margin of each column of text. Changes to illustrations are indicated by pointing hands, change bars, or MAJOR CHANGE symbols. Changes to diagrams may be indicated by shaded borders.

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1. The TPDRs listed below have been incorporated in this issue.

IDENTIFICATION NUMBER/ QA SEQUENCE NUMBER	LOCATION
NONE	

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SYSTEM SCHEMATICS
MISSION COMPUTER SYSTEM

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INTRODUCTION
ORGANIZATIONAL MAINTENANCE
SYSTEM SCHEMATICS
MISSION COMPUTER SYSTEM

This WP supersedes WP002 00, dated 1 June 1992.

1. PURPOSE.

2. This manual has system schematics to give information about the system and allow signal tracing through the system. The system schematics support on-aircraft maintenance of mechanical, pneudraulic, electrical, and electronic functions. These functions are integrated on the schematics for ease of troubleshooting a complete system.

3. REQUISITION AND AUTOMATIC DISTRIBUTION OF NAVAIR TECHNICAL MANUALS.

4. Procedures to be used by Naval activities and other Department of Defense activities requiring NAVAIR technical manuals are defined in NAVAIR 00-25-100.

5. CONTENT.

6. Each system is supported by schematics and a component locator.

7. **COMPONENT LOCATOR.** The component locator shows aircraft location, nomenclature, and reference designation number of each system component. The illustration shows the technicians view when possible.

8. **SCHEMATICS.** Simplified schematics, and detailed schematics provide direct support for testing and troubleshooting. All schematics are shown with electrical power off, switches in off position, and relays in deenergized position unless noted on schematic.

9. **Simplified Schematics.** Simplified schematics consist primarily of blocks connected by single lines with limited use of symbols and pictorial drawings

of units. These schematics simplify system functions as much as possible.

10. **Detailed Schematics.** Detailed schematics integrate applicable electrical, pneudraulic, and mechanical functions of the system. Detailed schematics show component location, connector pin letters and numbers, in line connectors, test points, and enough data to trace signals through the components within the system. Operational information next to components provides more data as required.

11. SCHEMATIC HIGHLIGHTS.

12. For schematic highlights see figure 1.

13. MANUAL ISSUE DATE.

14. The date on the title page is the copy freeze date. No additions, deletions, or changes are made after the manual issue date except last minute safety of flight or required maintenance changes. Data collected after the manual issue date will be included in later changes or revisions of the manual.

15. EFFECTIVITIES.

16. Effectivity notes on manual title pages, work package title pages, and within a work package indicate the aircraft or software program to which the data applies. If no effectivity note appears on the work package title page, the work package has the same effectivity as shown on the manual title page. The effectivity notes may use:

NOTE

Aircraft with model designator F/A-18B are the same type and model as TF/A-18A.

a. Type, model, and series

- b. Bureau number (tail number)
- c. Combination of type, model, series, and bureau numbers
- d. Part number or serial number
- e. Technical directive number
- f. Configuration/identification number
17. The table below shows examples of effectivity notes and their meanings:

Effectivity Note Examples

Effectivity Note	Definition
160777 AND UP	Applicable to all F/A-18A, F/A-18B, F/A-18C, and F/A-18D for bureau numbers listed.
F/A-18A, F/A-18B	Applicable to all F/A-18A and F/A-18B.
F/A-18C, F/A-18D	Applicable to all F/A-18C and F/A-18D.
F/A-18A	Applicable to all F/A-18A, but not F/A-18B, F/A-18C, and F/A-18D.
F/A-18B	Applicable to all F/A-18B, but not F/A-18A, F/A-18C, and F/A-18D.
F/A-18C	Applicable to all F/A-18C, but not F/A-18A, F/A-18B, and F/A-18D.
F/A-18D	Applicable to all F/A-18D, but not F/A-18A, F/A-18B, and F/A-18C.
F/A-18A, F/A-18C	Applicable to all F/A-18A and F/A-18C, but not to F/A-18B and F/A-18D.
F/A-18B, F/A-18D	Applicable to all F/A-18B and F/A-18D, but not to F/A-18A and F/A-18C.
F/A-18A 160775, 160777 THRU 160782	Only applicable to some bureau numbers of F/A-18A. Not applicable to any F/A-18B, even if an F/A-18B bureau number is within the numbers listed.
F/A-18C 163427, 163430 THRU 163456	Only applicable to some bureau numbers of F/A-18C. Not applicable to any F/A-18D, even if an F/A-18D bureau number is within the numbers listed.
F/A-18B 160784 AND UP	Only applicable to some bureau numbers of F/A-18B. Not applicable to any F/A-18A, even if an F/A-18A bureau number is within the numbers listed.
F/A-18D 163434 THRU 163457	Only applicable to some bureau numbers of F/A-18D. Not applicable to any F/A-18C, even if an F/A-18C bureau number is within the numbers listed.
160775 THRU 160785 BEFORE F/A-18 AFC 772	Applicable to F/A-18A and F/A-18B for bureau numbers listed, before modification by technical directive.
161213 AND UP; ALSO 160775 THRU 160785 AFTER F/A-18 AFC 772	Applicable to aircraft modified during production; also applicable when affected aircraft have been modified by technical directive.

Effectivity Note Examples (Continued)

Effectivity Note	Definition
160775 THRU 160785; WHEN NO. 2 CONTROL PANEL P/N XXXX-X IS INSTALLED	Applicable to F/A-18A and F/A-18B for bureau numbers listed if panel P/N XXXX-X is installed. (Configuration before AVC)
161213 AND UP; ALSO 160775 THRU 160785; WHEN NO. 2 CONTROL PANEL P/N XXXX-Y (AVC-102) IS INSTALLED	Applicable to aircraft modified during production; also applicable to aircraft components modified to the production configuration by technical directive. (Configuration after AVC)
P/N MBEU65101-9, MBEU65101-10 & MBEU65105-3	Applicable to assemblies which are interchangeable between aircraft.
ENGINE NO. 215101 THRU 215109	Applicable to assemblies which are interchangeable between aircraft, but configurations can not be identified by part number.
CONFIG/IDENT NUMBER 84A	The CONFIG/IDENT Number is the program load identification number which identifies the software program loaded in specific programmable units. Refer to A1-F18AC-SCM-000 for CONFIG/IDENT Number tables.

18. TECHNICAL DIRECTIVES.

19. Technical directives are documents which direct the accomplishment, and recording of a retrofit configuration or inspection to delivered aircraft, or aircraft components.

20. AIRFRAME CHANGE (AFC) AND AIRBORNE TACTICAL SOFTWARE CHANGE (ASC).

Technical directives which change configuration of aircraft structure or equipment installation, i.e. AFC, will list aircraft bureau numbers in effectivity notes and show before and after the AFC. Technical directives which change configuration of operational flight programs (OPF), i.e. ASC, will list the OPF CONFIG/IDENT NUMBER in effectivity notes and show the latest two authorized OPF programs. See AFC and ASC effectivity examples in Effectivity Note Example Table.

21. AIRCRAFT COMPONENT CHANGES. Technical directives which change configuration of aircraft components, i.e. AAC, ACC, AVC, AYC, and PPC will list part numbers in the effectivities. See AVC effectivity examples in Effectivity Note Example table.

22. RECORD OF APPLICABLE TECHNICAL DIRECTIVES.

23. The technical directives affecting this manual are listed in the Record of Applicable Technical Directives of each affected work package. Because an ASC directs all aircraft be modified within 30 days, ASC's are not listed. When all affected aircraft are modified, the before configuration is removed from the manual, and the technical directive entry is removed from the Record of Applicable Technical Directives.

24. TECHNICAL PUBLICATIONS DEFICIENCY REPORTS (TPDR).

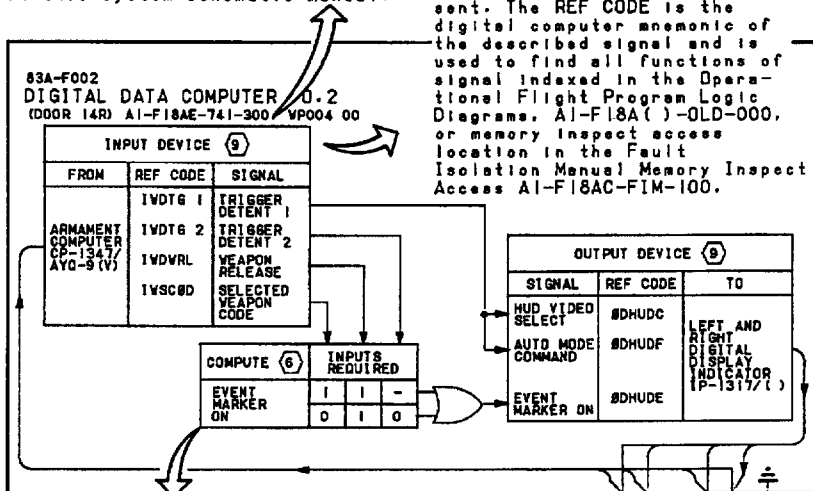
25. The TPDR (OPNAV FORM 4790/66) is the form for reporting errors and suspected omissions in the technical manuals. Reporting procedures are in OPNAVINST 4790.2 SERIES.

26. NAVY (AN) STANDARD/COMMON NAME NOMENCLATURE.

27. When an item has both NAVY (AN) standard and common name nomenclature assigned, the common name nomenclature will be used in text and on illustrations. Full NAVY (AN) standard nomenclature will be used in the Illustrated Parts Breakdown (IPB).

AI-F18AC-741-300. WP004 00 is a reference to manual which contains component maintenance procedures. When no reference appears, the system maintenance for the component is contained in the -300 series system manual being covered in this system schematic manual.

INPUT OR OUTPUT DEVICE describes the signal, tells where signal comes from or to what component signal is sent. The REF CODE is the digital computer mnemonic of the described signal and is used to find all functions of signal indexed in the Operational Flight Program Logic Diagrams. AI-F18A()-OLD-000, or memory inspect access location in the Fault Isolation Manual Memory Inspect Access AI-F18AC-FIM-100.



COMPUTER MATRIX shows the computer operational flight program in a truth table form. A hexagon symbol is placed in the computer matrix and is a reference to the **LEGEND** for an explanation of matrix.

83P-F002D is the reference designator for an electrical disconnect. The reference designator is used as the entry point into the Wiring Diagram Manual. AI-F18A()-VDM-000 or Wiring Repair Manual. AI-F18AC-VRM-000. It may also be used to get the part number of the item by cross referencing in the Ref Des Section of the Parts List Index. AI-F18AC-IPB-450.

HEXAGON SYMBOL contains a number. This symbol and number are used to reference the notes contained in the **LEGEND**.

SQUARE SYMBOL contains a number that refers to a sheet of the schematic where the circuitry is continued.

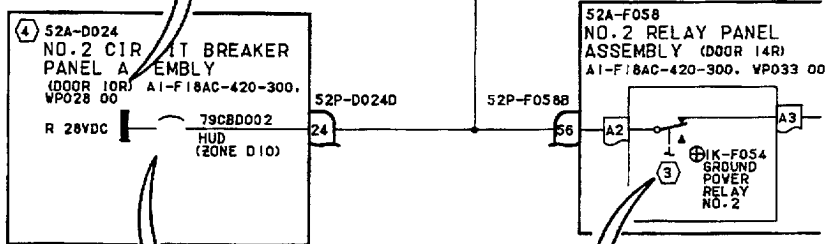
Figure 1. Schematic Highlights (Sheet 1)

Aircraft electrical disconnects are shown on schematics. The disconnect reference designator and aircraft location are shown on schematic. These disconnects may be used as test points for signal tracing.

FLAG SYMBOL contains a number. This symbol and number are used to reference the notes contained in the LEGEND.

Alternate aircraft wiring hookups for different aircraft configurations are shown using this symbol.

(DOOR 10R) Indicates component location on aircraft



Information pertinent to circuit breaker is shown on schematics as listed below:

- R 28VDC is the aircraft bus which supplies voltage to circuit breaker.
- 79CBD002 is the reference designator of circuit breaker and is located next to breaker on rear of panel.
- HUD is the name of circuit breaker and is located next to breaker on front of panel.
- (ZONE D10) is the location of circuit breaker on the circuit breaker panel. The letter D is the vertical location and number 10 is the horizontal location.

DEENERGIZED WHEN GROUND POWER 2 SWITCH IS IN B ON. EXTERNAL ELECTRICAL POWER IS NOT APPLIED, OR APU IS NOT IN GROUND MAINTENANCE MODE.

Operation highlights give pertinent information about the operation of the circuit, for ease of signal tracing.

Figure 1. Schematic Highlights (Sheet 2)

The legend contains all notes pertinent to the schematic as listed below:

- NUMBER listed with no symbol is general information about the schematic.
- NONSTANDARD SYMBOLS appearing on schematic are shown or referenced with an explanation.
- ABBREVIATIONS appearing on schematic are shown or referenced with an explanation.
- HEXAGON SYMBOL refers to another schematic or manual for continuation of a circuit or an explanation of data contained on schematic.
- FLAG SYMBOL indicates limited aircraft application.



LEGEND

1. CONTINUITY TESTS:

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A(1)-VDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.

2. NONSTANDARD SYMBOLS:

- \oplus IDENTIFIES RELAY USED TO SWITCH TO LOW LEVEL CURRENT. SEE NOTE 1.

③ GROUND POWER SWITCHING SCHEMATIC. A1-F18AC-420-500. VP005 00.

④ POWER DISTRIBUTION SCHEMATIC. A1-F18AC-420-500. VP004 00.

⑤ EXPLANATION OF MATRIX:

- A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.
- B. INPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUT.
- C. THE SIGNAL OUTPUT IS READ HORIZONTALLY. EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.
- D. INTERPRET MATRIX TABLE AS INDICATED:
 - (1) ONE (1) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.
 - (2) ZERO (0) INDICATES THE INPUT AS NAMED MUST NOT BE THERE TO GET THE OUTPUT.
 - (3) DASH (-) INDICATES THE OUTPUT DOES NOT DEPEND ON THIS INPUT.

⑥ AVIONIC MUX CHANNEL 1 SCHEMATIC. A1-F18AC-741-500. VP003 00.

⑦ AVIONIC MUX CHANNEL 2 SCHEMATIC. A1-F18AC-741-500. VP003 00.

⑧ FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO A1-F18AC-OLD-000. FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REF CODE, REFER TO A1-F18AC-FIM-100.

⑨ F/A-18A.

⑩ F/A-18B.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

ABBREVIATIONS AND NONSTANDARD SYMBOLS

MISSION COMPUTER SYSTEM

Reference Material

None

Alphabetical Index

Subject	Page No.
Abbreviations, Table 1	2
Introduction	1
Nonstandard Symbols, Figure 1	3

Record of Applicable Technical Directives

None

1. INTRODUCTION.

2. This work package supports illustrations contained in this manual.
3. Table 1 contains abbreviations used in this manual.

4. Figure 1 contains the nonstandard symbols used in this manual.

Table 1. Abbreviations

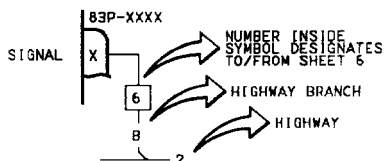
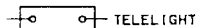
Abbreviation	Definition
ADC	Air Data Computer
ADF	Automatic Direction Finder
APU	Auxiliary Power unit
AUG	Augmented
BIT	Built-In Test
CC	Control-Converter
CHAN	Channel
CLC	Command Launch Computer
COMM	Communication
CSC	Control-Converter
CTR	Center
D/A	Digital-to-Analog
D/L	Data Link
DDC	Digital Data Computer (MC)
DDI	Digital Display Indicator
DSPLY	Display
EEC	Electronic Equipment Control (UFC)
FCCA	Flight Control Computer A
FCCB	Flight Control Computer B
FLIR	Forward Looking Infrared System
GND	Ground
HI	Horizontal Indicator IP-1350A
HOTAS	Hands On Throttle And Stick
HUD	Head Up Display
IBIT	Initiated Built-In Test

Table 1. Abbreviations (Continued)

Abbreviation	Definition
ICS	Intercommunication Amplifier-Control System
IFF	Identify Friend or Foe
INS	Inertial Navigation System
INTR	Interior
MAINT	Maintenance
MC	Mission Computer (Digital Data Computer)
MDG	Multi-Display Group
MEM	Memory
MUX	Multiplex Bus
NAV	Navigation
OFP	Operational Flight Program
PWR	Power
RCVR	Receiver
S/W	Software
SMS	Stores Management Set
STBY	Standby
SYS	System
TACTS	Tactical Aircrew Combat Training System
TDC	Throttle Designator Control
Temp	Temperature
UFC	Up Front Control (EEC)
WRA	Weapon Replaceable Assembly

LEGEND

1. NONSTANDARD SYMBOLS:



⊕ IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT. SEE NOTE 2.

⊗ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTORS. SEE NOTE 2.

Figure 1. Nonstandard Symbols

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

COMPONENT LOCATOR

MISSION COMPUTER SYSTEM

This WP supersedes WP003 00, dated 1 June 1992.

Title	Work Package
Circuit Breakers	003 01
Component Locator, (F/A-18A/B)	003 02
Component Locator (F/A-18A/B AFTER AFC 225 AND AFC 231)	003 03
Component Locator (F/A-18A AFTER AFC 253 OR AFC 292; ALSO AFTER AFC 231 PART 2 OR AFC 231 PART 3)	003 04

ORGANIZATIONAL MAINTENANCE
SYSTEM SCHEMATICS
CIRCUIT BREAKERS
MISSION COMPUTER SYSTEM

Reference Material

None

Alphabetical Index

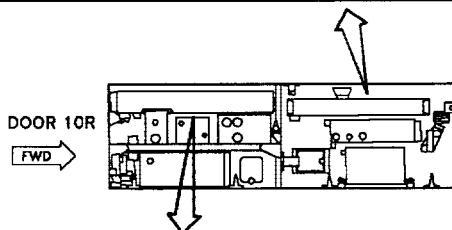
Subject	Page No.
Circuit Breakers, Figure 1	2

Record of Applicable Technical Directives

None

2 52A-D024 NO.2 CIRCUIT BREAKER PANEL ASSEMBLY			
ZONE	REF DES	NOMENCLATURE	BUS
A11	80C8D007	MFD	R115VAC2A
A12	83C8D006	DIGITAL DATA COMPUTER NO.2	R115VAC2A
A20	71C8D002	ADF	R26VAC2C
B11	80C8D008	MFD	R115VAC2B
B12	83C8D010	DIGITAL DATA COMPUTER NO.2	R115VAC2B
C11	80C8D009	MFD	R115VAC2C
C12	83C8D011	DIGITAL DATA COMPUTER NO.2	R115VAC2C

3 52A-D024 NO.2 CIRCUIT BREAKER PANEL ASSEMBLY			
ZONE	REF DES	NOMENCLATURE	BUS
A11	82C8D002	CONTROL CONVERTER	R115VAC2A
A17	80C8D007	MFD	R115VAC2A
A18	83C8D006	DIGITAL DATA COMPUTER NO.2	R115VAC2A
B11	82C8D003	CONTROL CONVERTER	R115VAC2B
B17	80C8D008	MFD	R115VAC2B
B18	83C8D010	DIGITAL DATA COMPUTER NO.2	R115VAC2C
C11	82C8D004	CONTROL CONVERTER	R26VAC2C
D6	71C8D002	ADF	R115VAC2C
D7	80C8D009	MFD	R115VAC2C
D8	83C8D011	DIGITAL DATA COMPUTER NO.2	R115VAC2C

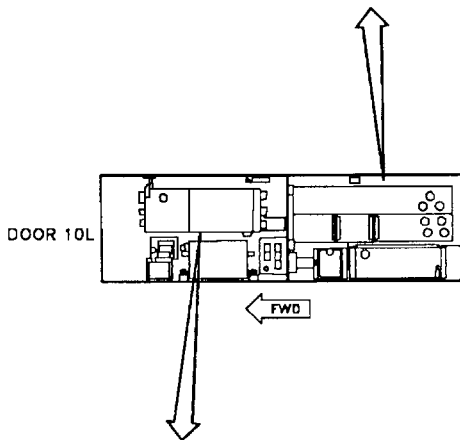


2 52A-D026 NO.4 CIRCUIT BREAKER PANEL ASSEMBLY			
ZONE	REF DES	NOMENCLATURE	BUS
B12	82C8D005	CONTROL CONVERTER	R26VDC
C7	82C8D004	CONTROL CONVERTER	R115VAC2C
C8	82C8D003	CONTROL CONVERTER	R115VAC2B
C9	82C8D002	CONTROL CONVERTER	R115VAC2A

3 52A-D026 NO.4 CIRCUIT BREAKER PANEL ASSEMBLY			
ZONE	REF DES	NOMENCLATURE	BUS
B3	82C8D005	CONTROL CONVERTER	R26VDC

Figure 1. Circuit Breakers (Sheet 1)

52A-C159 NO.8 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY			
ZONE	REF DES	NOMENCLATURE	BUS
D2	850BC004	MSDRS	MAINT 24/28VDC
D12	80C8C006	MMD	L115VAC/2C
E12	80C8C005	MMD	L115VAC/2B
F12	80C8C004	MMD	L115VAC/2A



52A-C057 NO.8 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY			
ZONE	REF DES	NOMENCLATURE	BUS
A20	83CBC006	DIGITAL DATA COMPUTER NO.1	L115VAC/2A
B6	10BCD48	GND PWR CONT	L28VDC
B20	83CBC007	DIGITAL DATA COMPUTER NO.1	L115VAC/2B
C20	83CBC008	DIGITAL DATA COMPUTER NO.1	L115VAC/2C

LEGEND

1. AIRCRAFT CONNECTOR LOCATIONS
ARE SHOWN IN A1-F18()-WDM-000.

② > 161353 THRU 161359.

③ > 161360 AND UP.

Figure 1. Circuit Breakers (Sheet 2)

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****COMPONENT LOCATOR****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A/B**

Reference Material

None

Alphabetical Index**Subject****Page No.**

Component Locator, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 48	-	Automatic AC Bus Isolation (ECP MDA-F/A-18-00121)	1 Jun 92	-

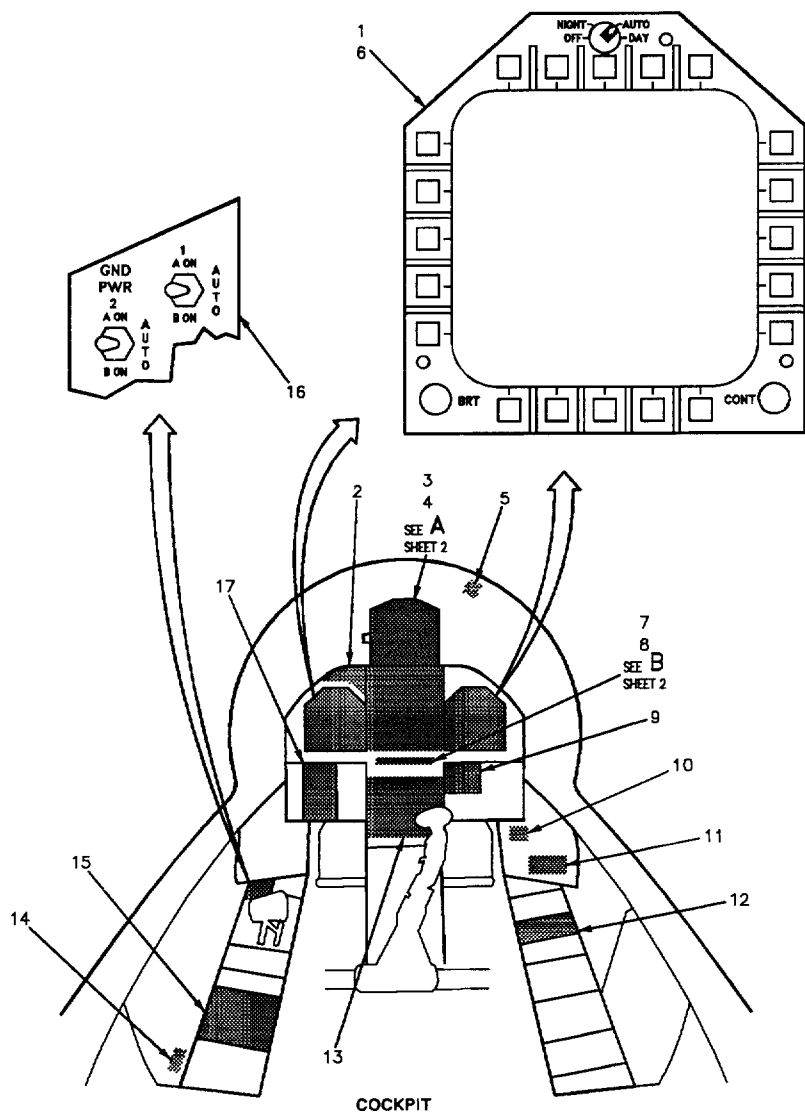
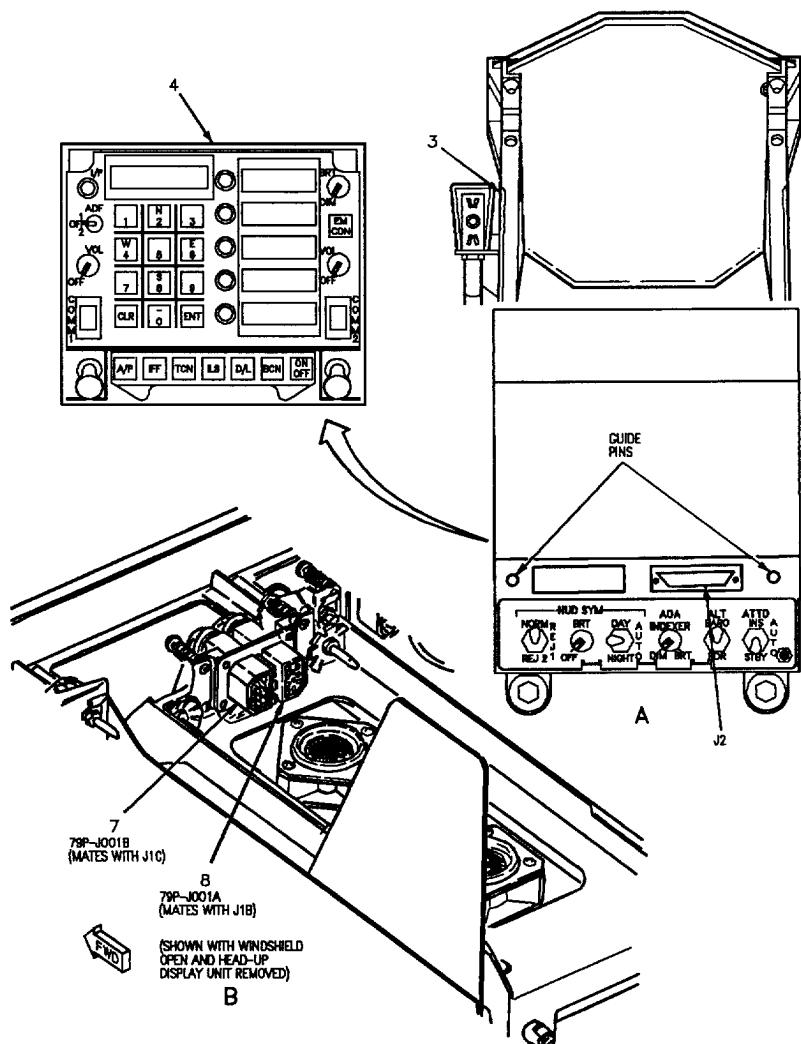


Figure 1. Component Locator (Sheet 1)



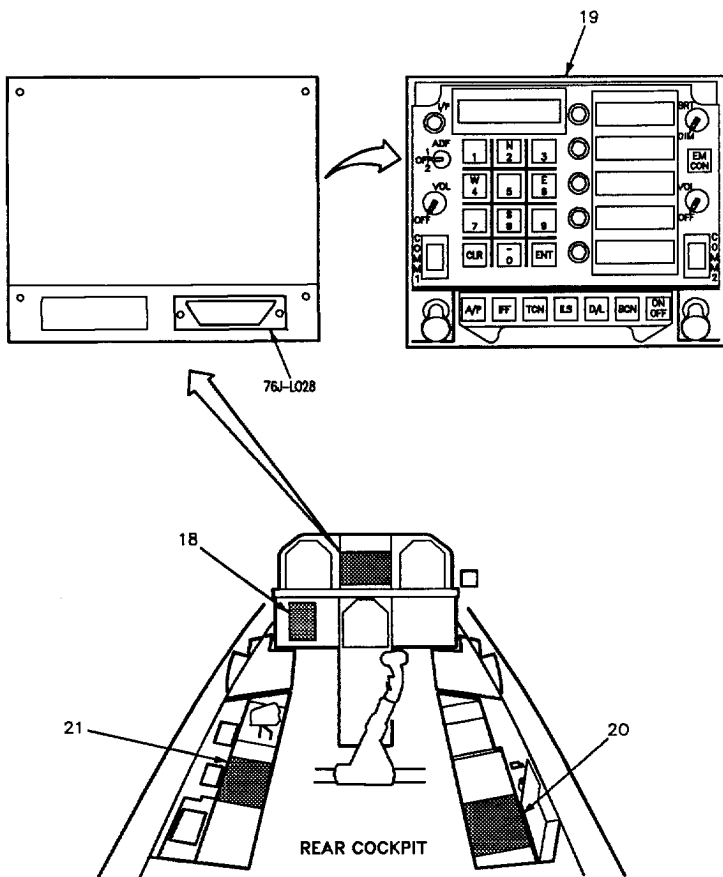


Figure 1. Component Locator (Sheet 3)

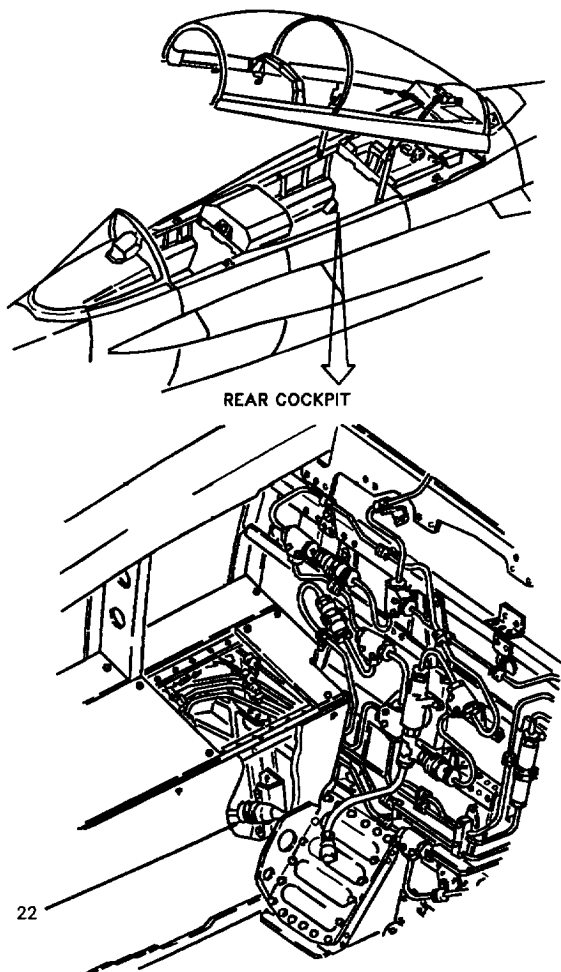


Figure 1. Component Locator (Sheet 4)

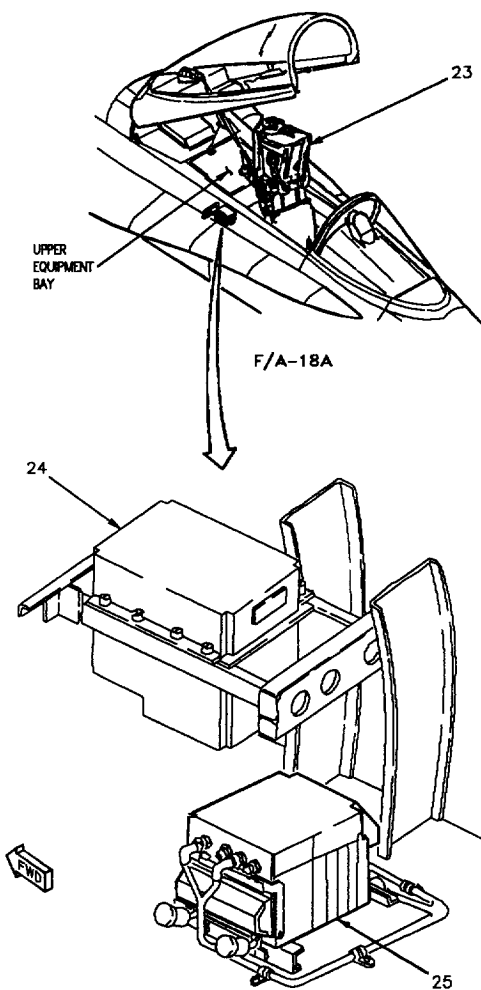


Figure 1. Component Locator (Sheet 5)

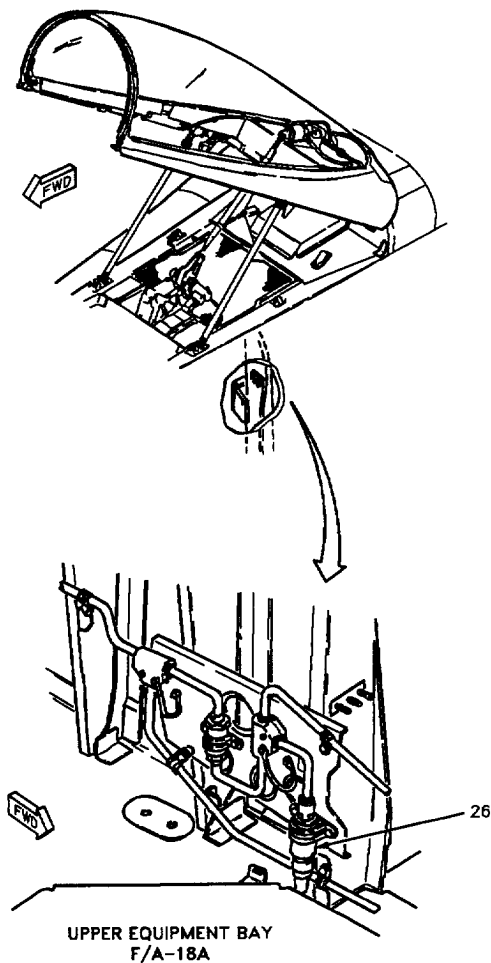


Figure 1. Component Locator (Sheet 6)



Figure 1. Component Locator (Sheet 7)

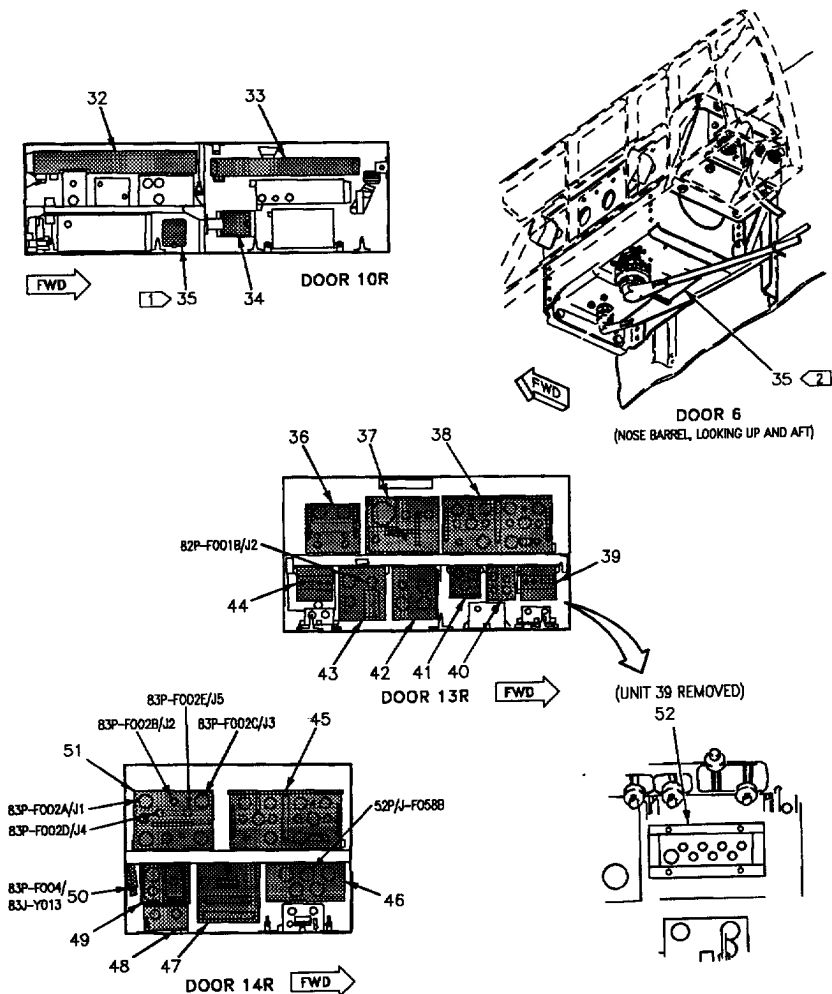


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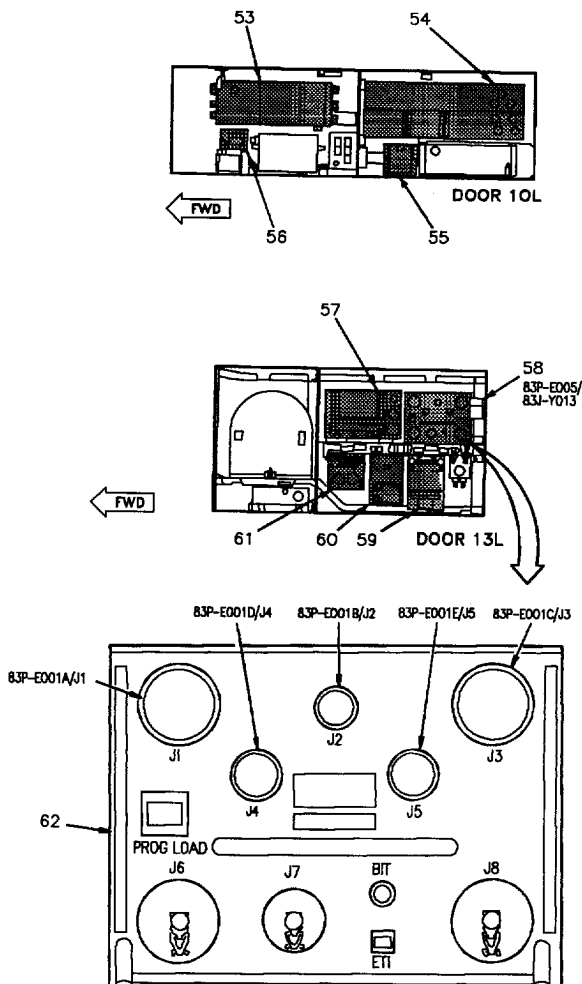


Figure 1. Component Locator (Sheet 9)

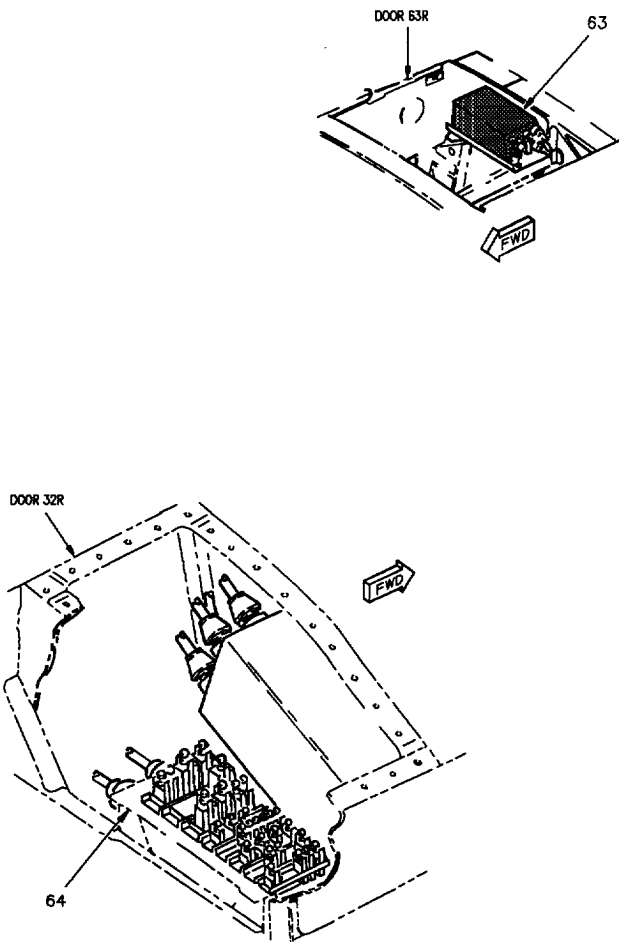


Figure 1. Component Locator (Sheet 10)

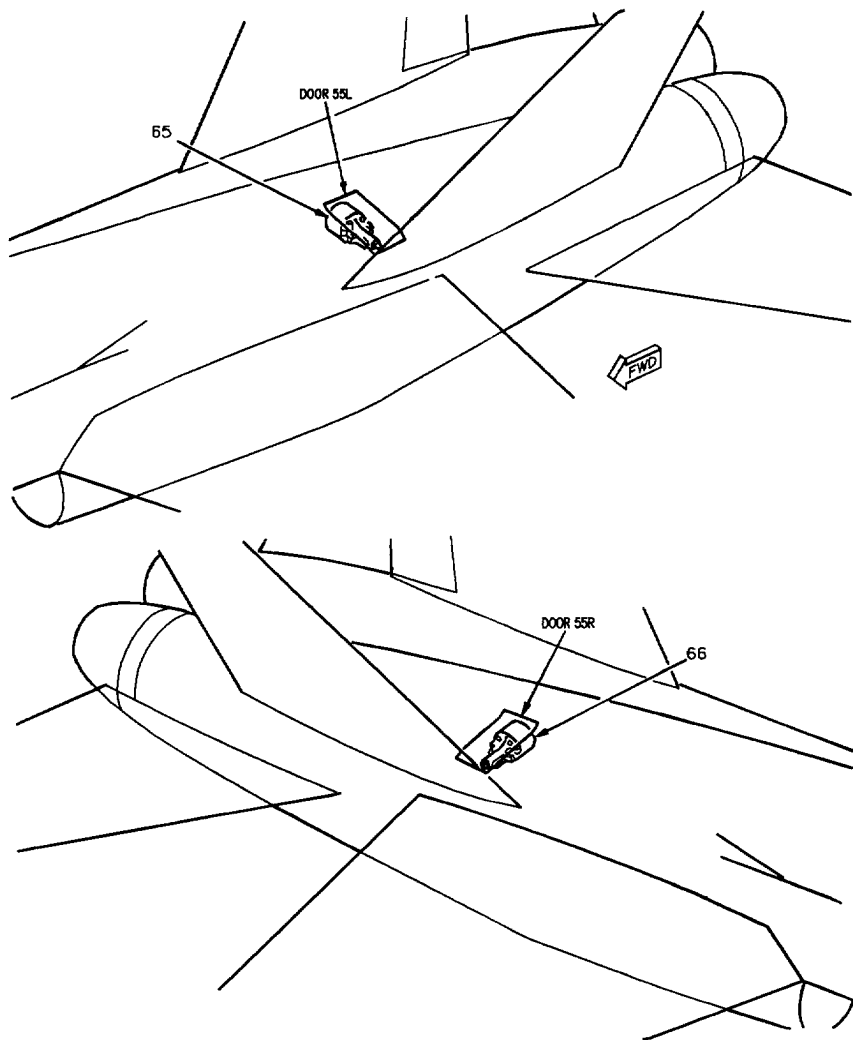


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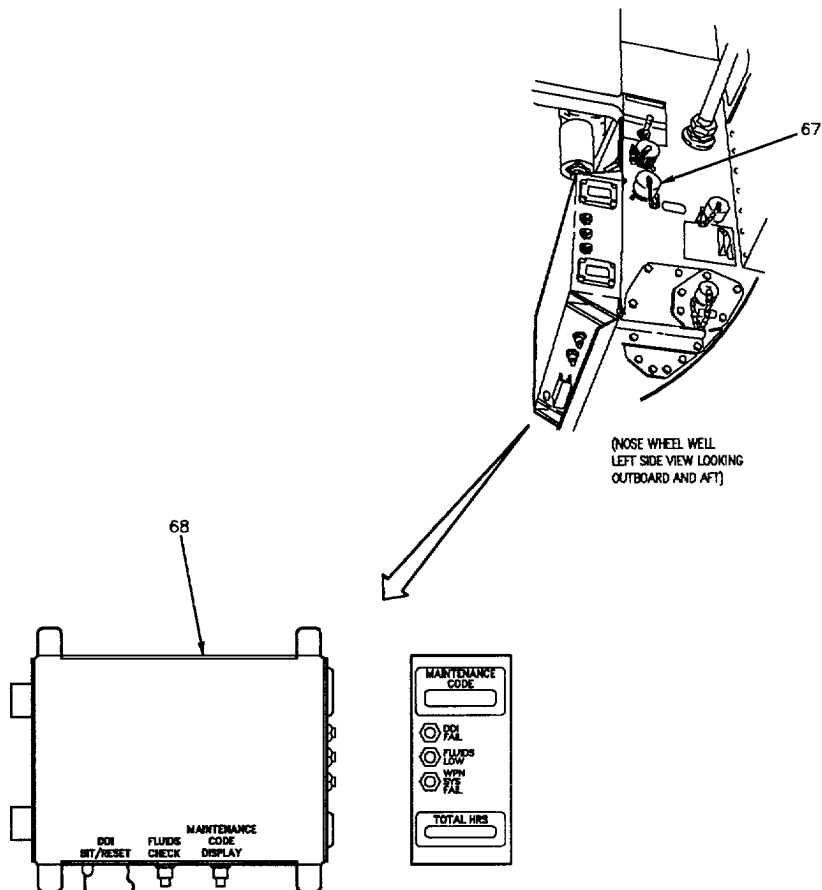


Figure 1. Component Locator (Sheet 12)

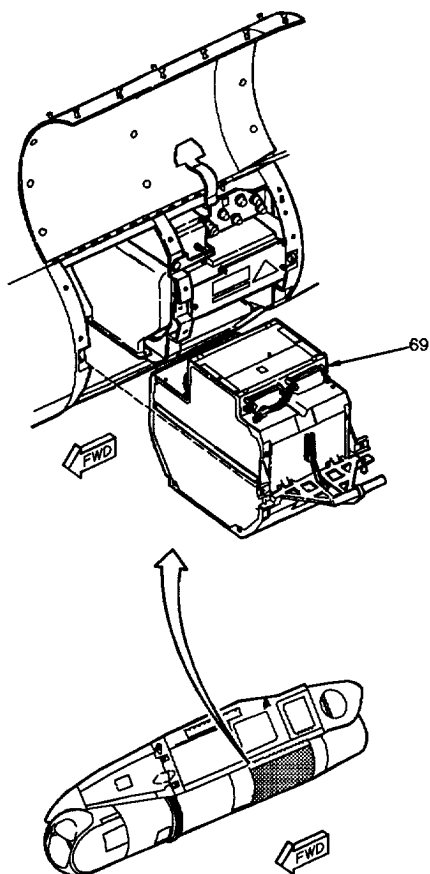
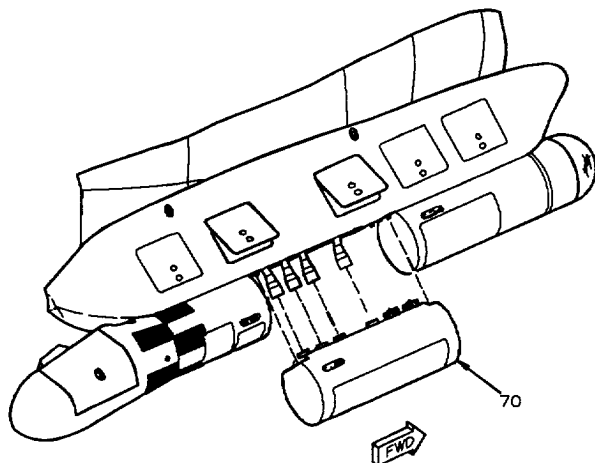


Figure 1. Component Locator (Sheet 13)



LEGEND

1. AIRCRAFT CONNECTOR LOCATIONS
ARE SHOWN IN A1-F18()-WDM-000.

Figure 1. Component Locator (Sheet 14)

Nomenclature	Index No.	Ref Des
AIRCRAFT EJECTION SEAT	23	25A-H002
AIR DATA COMPUTER	42	70A-F001
ANTENNA SELECTOR SA-2292/A	52	76S-F004
ARMAMENT COMPUTER CP-1342/AYQ-9(V)	47	61A-F001
ATTITUDE REFERENCE INDICATOR ARU-48/A	9	33M-J015
CAUTION LIGHT INDICATOR PANEL	11	8A-J042
3 COCKPIT ELECTRIC LIGHT CONTROL	24	8A-L001
COMMAND LAUNCH COMPUTER	37	61A-F010
COMPUTER-POWER SUPPLY CP-1325/APG-65	27	60A-A505
COMPUTER-TRANSPONDER KIT-1A/TSEC	59	78A-E003
CONTROL-CONVERTER	43	82A-F001
CONTROLLER-PROCESSOR C-10661/AAS-38	69	61A-P520
COUNTERMEASURES COMPUTER CP-1293/ALR-67(V)	60	62A-E006
CREW STATION ENGINE MONITOR INDICATOR AEU-12/A	17	3M-H001
DIRECTION FINDER OA-8697/ARD	30	71ARB001
DIGITAL DATA COMPUTER NO. 1	62	83A-E001
DIGITAL DATA COMPUTER NO. 2	51	83A-F002
DIGITAL DISPLAY INDICATOR ID-2150/ASM-612	68	85A-G003
ELECTRONIC EQUIPMENT CONTROL	4	79A-J006
EXTERNAL POWER CONTACTOR	56	1K-C022
FUEL QUANTITY GAGING INTERMEDIATE DEVICE	48	5A-F014
GND PWR CONTROL PANEL ASSEMBLY	16	1A-H004
HEAD-UP DISPLAY	3	79A-J001
HEAD-UP DISPLAY DISCONNECT	7	79P-J001B
HEAD-UP DISPLAY DISCONNECT	8	79P-J001A
HEIGHT INDICATOR ID-2163/A	10	67A-J002

Figure 1. Component Locator (Sheet 15)

Nomenclature	Index No.	Ref Des
HORIZONTAL INDICATOR IP-1350/A	13	80A-J003
3 IFF SWITCH	26	78S-K005
4 IFF SWITCH	22	78S-L005
INERTIAL NAVIGATION UNIT	57	68A-E001
INTERCOMMUNICATION AMPLIFIER-CONTROL AM-6979/A OR AM-7360/A	15	76A-H009
INTERCONNECTING BOX J-3656/ASQ-173	70	61ARR510
INTERFERENCE BLANKER MX-9965/A	40	A-F001
INTR LT CONTROL BOX PANEL ASSEMBLY	12	8A-J002
1 LANDING GEAR CONTROL UNIT	35	12A-D004
2 LANDING GEAR CONTROL UNIT	35	12A-A004
LEFT DIGITAL DISPLAY INDICATOR IP-1317()	1	80A-H001
LEFT MUX BUS IMPEDANCE MATCHING NETWORK	58	83A-Y013
LEFT POWER CONTACTOR	55	1K-C007
LH ADVISORY AND THREAT WARNING INDICATOR PANEL	2	52A-H073
LOCK/SHOOT LIGHT ASSEMBLY	5	8DSJ150
MC/HYD ISOL CONTROL PANEL ASSEMBLY	14	52A-H081
MUX TEST CONNECTOR	67	83J-G003
NO. 2 CIRCUIT BREAKER PANEL ASSEMBLY	33	52A-D024
NO. 2 RELAY PANEL ASSEMBLY	46	52A-F058
NO. 4 CIRCUIT BREAKER PANEL ASSEMBLY	32	52A-D026
NO. 4 RELAY PANEL ASSEMBLY	64	52A-N118
NO. 7 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY	54	52A-C057
NO. 8 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY	53	52A-C159
PULSE DECODER KY-651()/ARA-63	41	74A-F002
RADAR RECEIVER R-1623/APN	29	72REB001
RADAR RECEIVER-TRANSMITTER RT-1028/APN-202	31	72A-A002
RADIO RECEIVER R-1379()/ARA-63	28	74REB001
4 REAR COCKPIT ELECTRIC LIGHT CONTROL	20	8A-L097
4 REAR CREW STATION ENGINE MONITOR INDICATOR AEU-12/A	18	3M-K002
4 REAR ELECTRONIC EQUIPMENT CONTROL	19	76A-L028

Figure 1. Component Locator (Sheet 16)

Nomenclature		Index No.	Ref Des
	RECEIVER-TRANSMITTER RT-1015()/APN-194(V)	63	67A-T001
	RECEIVER-TRANSMITTER RT-1157()/APX-100(V)	61	78A-E001
	RECEIVER-TRANSMITTER RT-1159/A	36	69A-F001
	RECEIVER-TRANSMITTER RT-1250()/ARC NO. 1	39	76A-F001
	RECEIVER-TRANSMITTER RT-1250()/ARC NO. 2	44	76A-F002
4	RECEIVER-TRANSMITTER PROCESSOR RT-1379()/ASW	21	77A-K001
3	RECEIVER-TRANSMITTER PROCESSOR RT-1379()/ASW	25	77A-L001
	RESERVOIR	65	10HPP006
	RESERVOIR	66	10HPR007
	RIGHT DIGITAL DISPLAY INDICATOR IP-1317()	6	80A-J002
	RIGHT MUX BUS IMPEDANCE MATCHING NETWORK	50	83A-Y013
	RIGHT POWER CONTACTOR	34	1K-D008
	ROLL-PITCH-YAW COMPUTER (FCCA)	38	84A-F001
	ROLL-PITCH-YAW COMPUTER (FCCB)	45	84A-F002
	SIGNAL DATA RECORDER RO-508/ASM-612	49	85A-F001
LEGEND			
1	161353 THRU 161987 BEFORE F/A-18 AFC 48.		
2	162394 AND UP; ALSO 161353 THRU 161987 AFTER F/A-18 AFC 48.		
3	F/A-18A.		
4	F/A-18B.		

Figure 1. Component Locator (Sheet 17)

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****COMPONENT LOCATOR****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AND F/A-18B AFTER F/A-18 AFC 225 AND F/A-18 AFC 231**

Reference Material

None

Alphabetical Index**Subject****Page No.**

Component Locator, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 48	-	Automatic AC Bus Isolation (ECP MDA-F/A-18-00121)	1 Jun 92	-
F/A-18 AFC 225	-	Five (5) Avionics Multiplex Bus Upgrade, Incorporation of (ECP MDA-F/A-18 0529)	1 Jun 02	-
F/A-18 AFC 231	-	Embedded Global Positioning System (GPS)/Inertial Navigation System (INS) (EGI), Incorporation of (ECP MDA-F/A-18 0521)	1 Jun 02	-

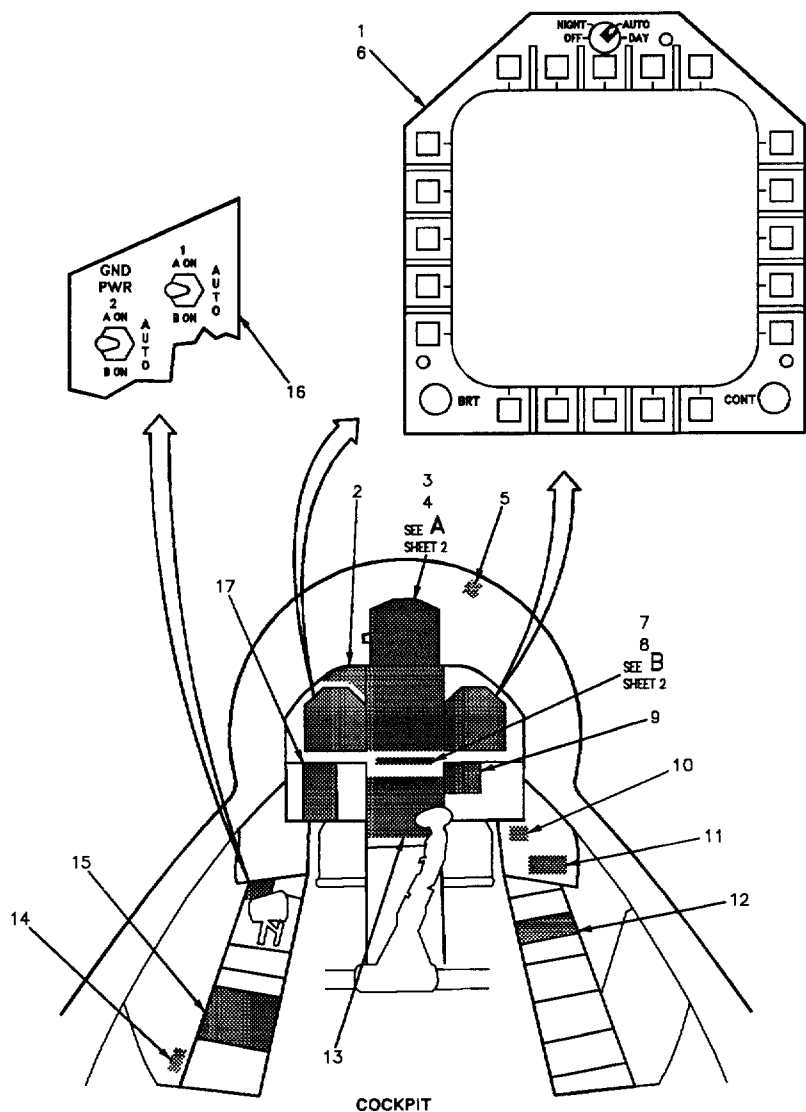


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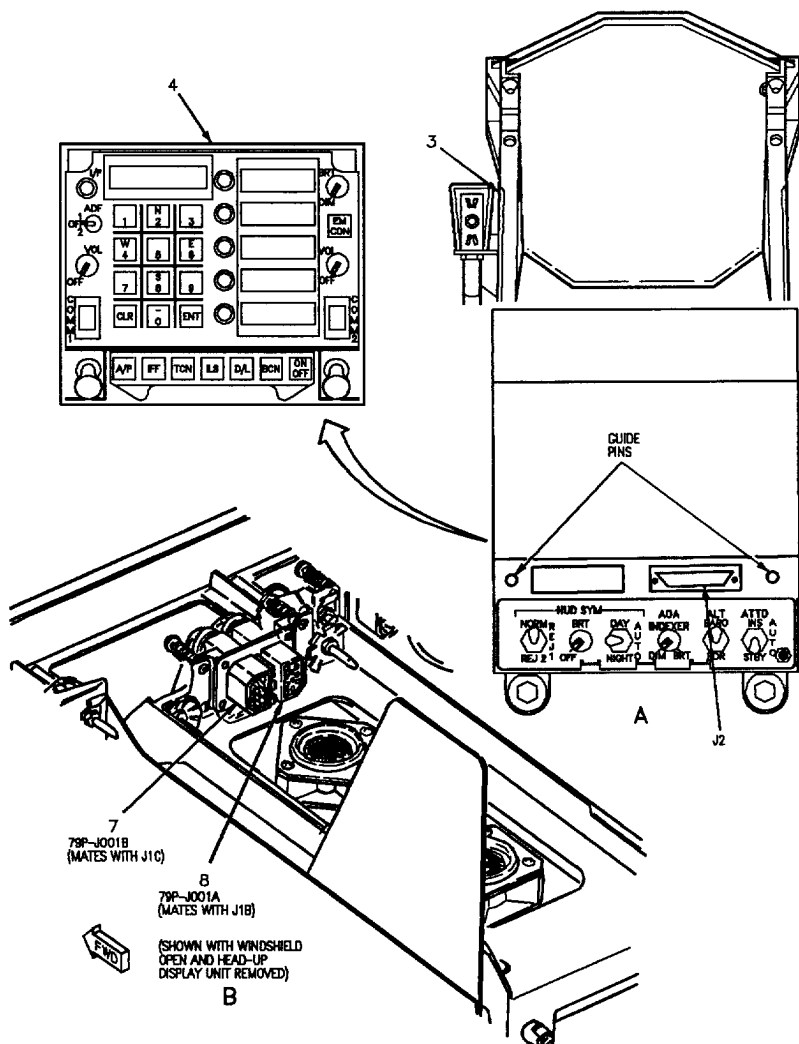


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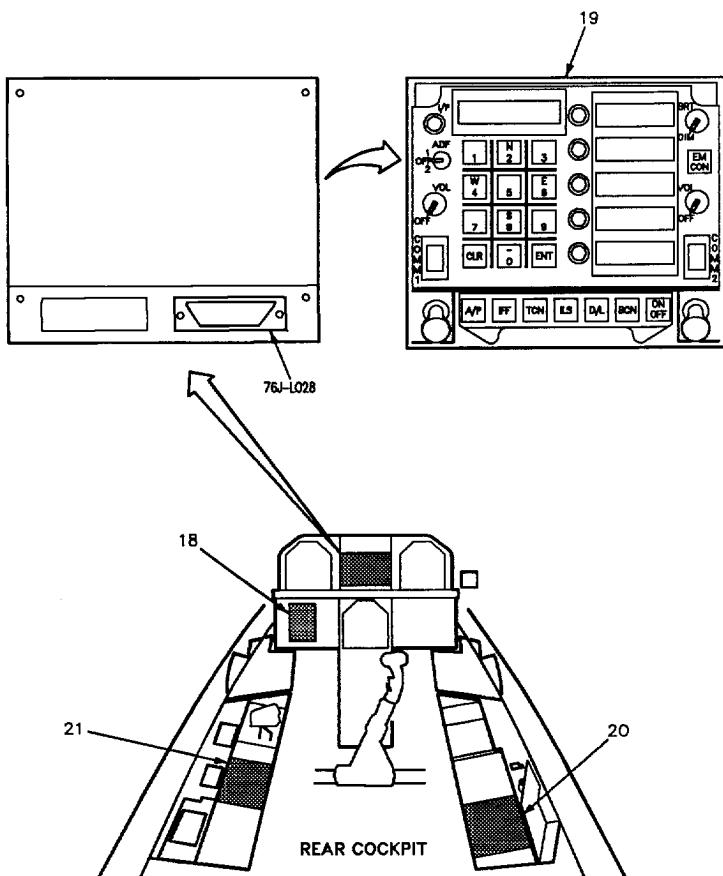


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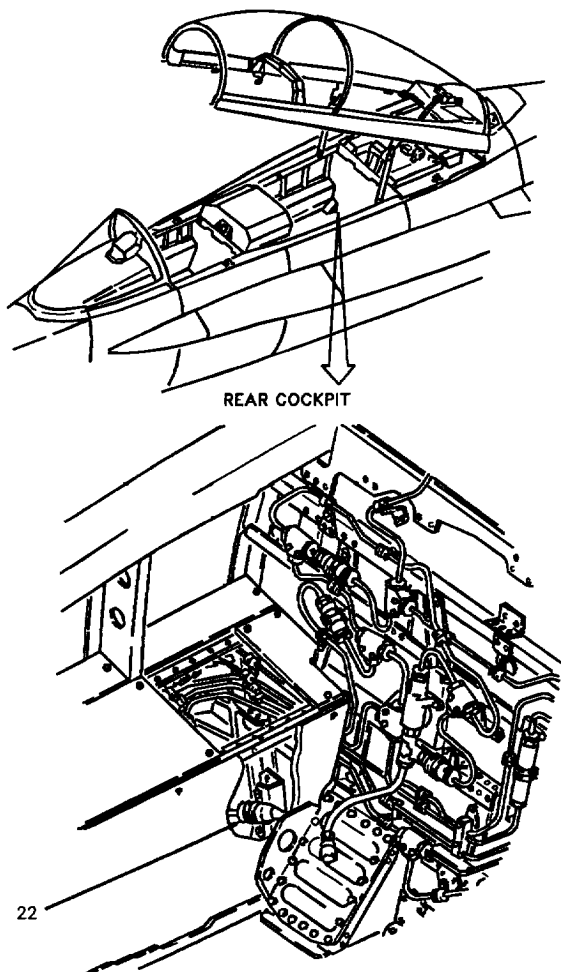


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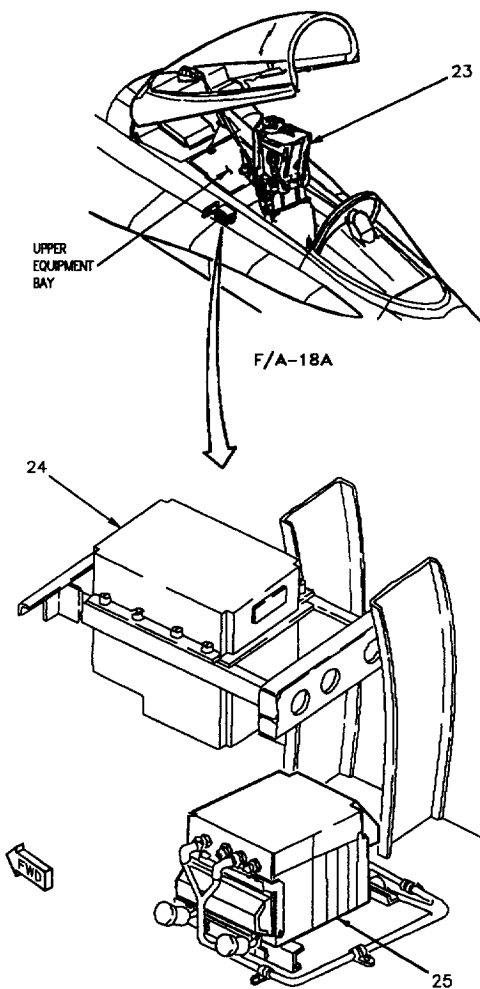


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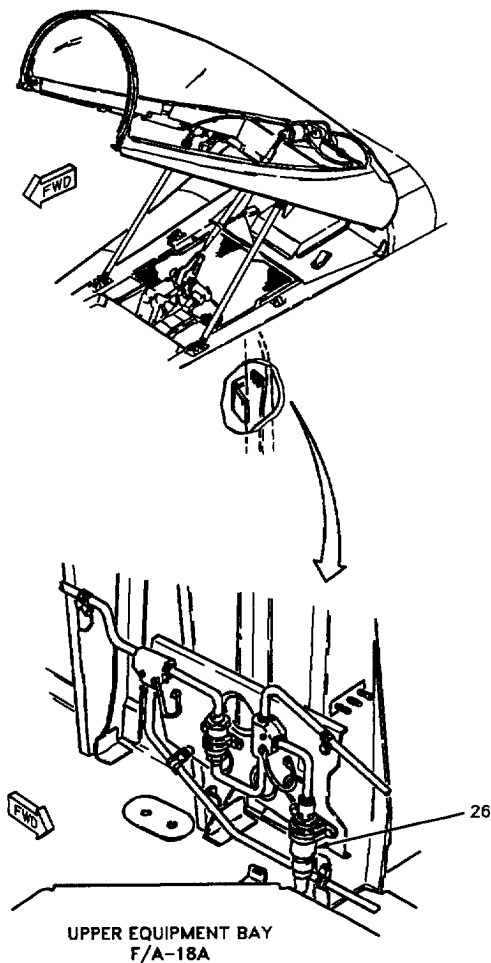


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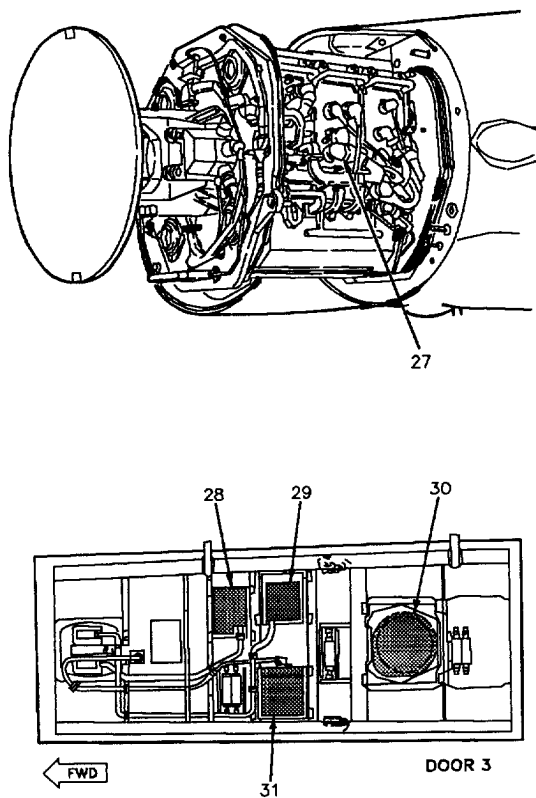


Figure 1. Component Locator (Sheet 7)

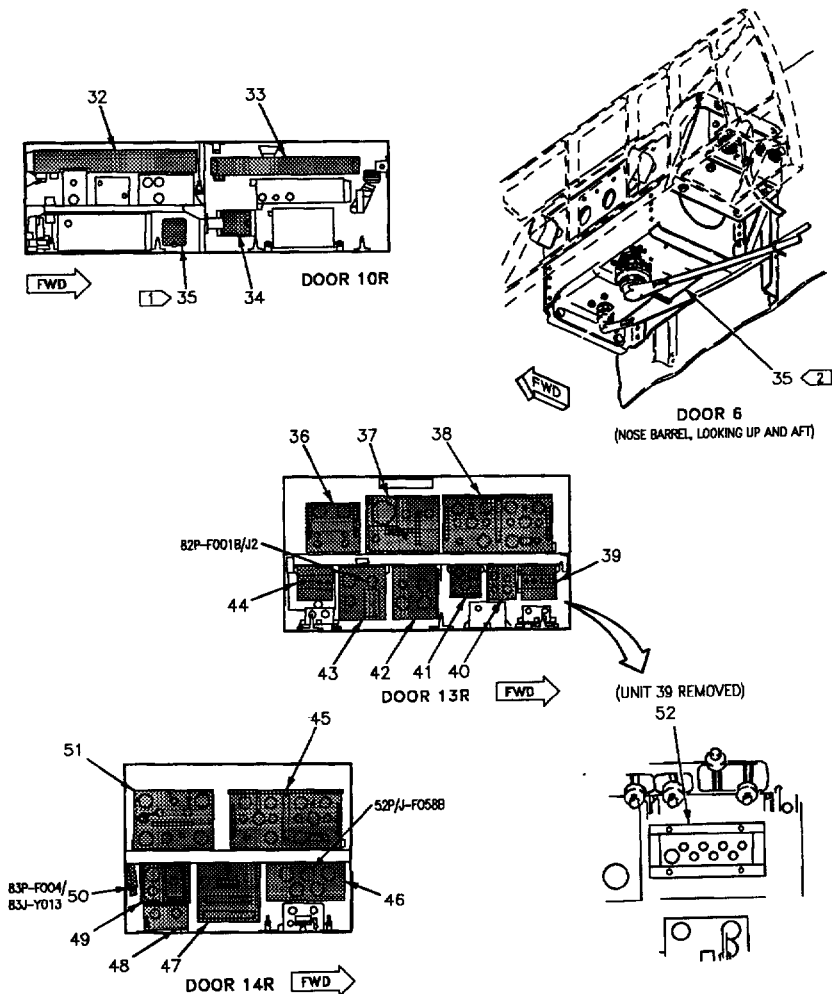


Figure 1. Component Locator (Sheet 8)

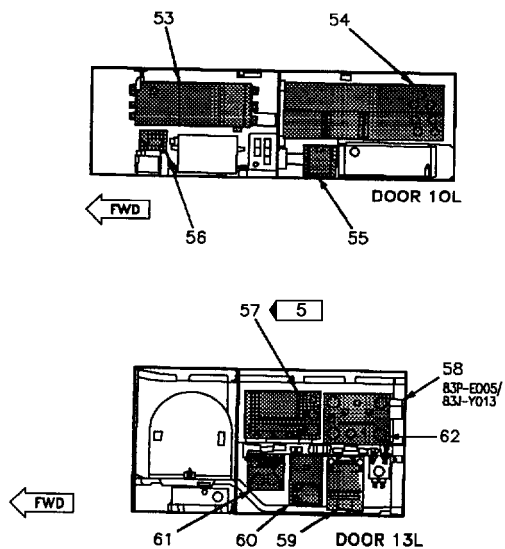
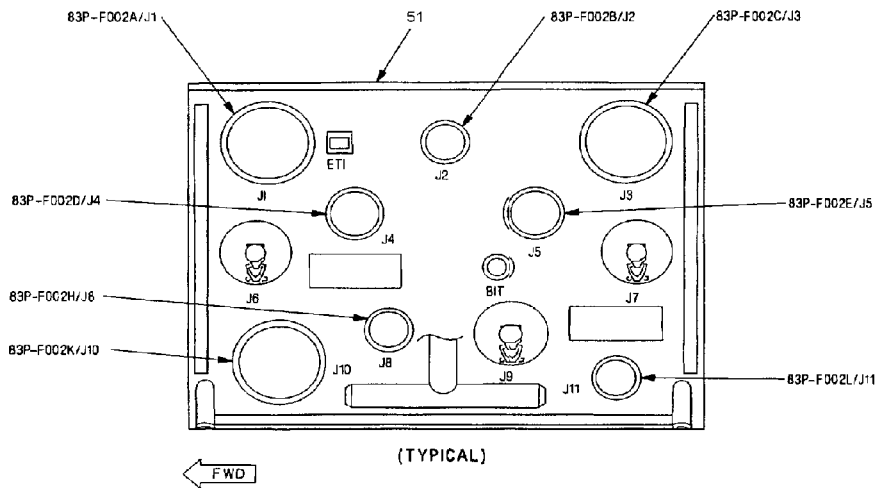
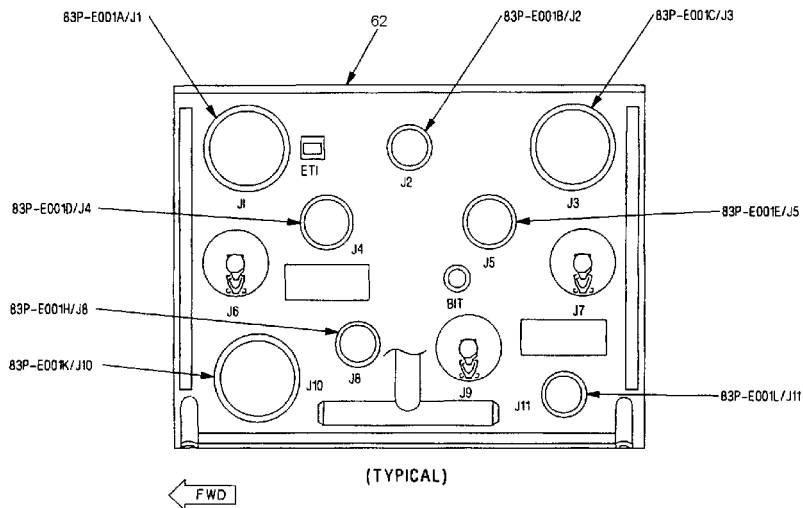


Figure 1. Component Locator (Sheet 9)



03030110

Figure 1. Component Locator (Sheet 10)



03030111

Figure 1. Component Locator (Sheet 11)

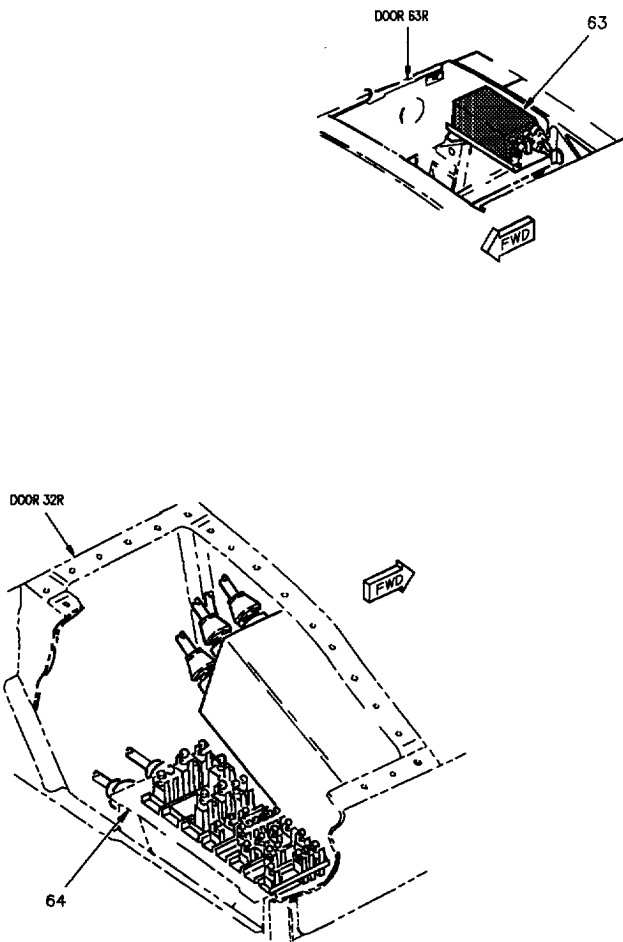
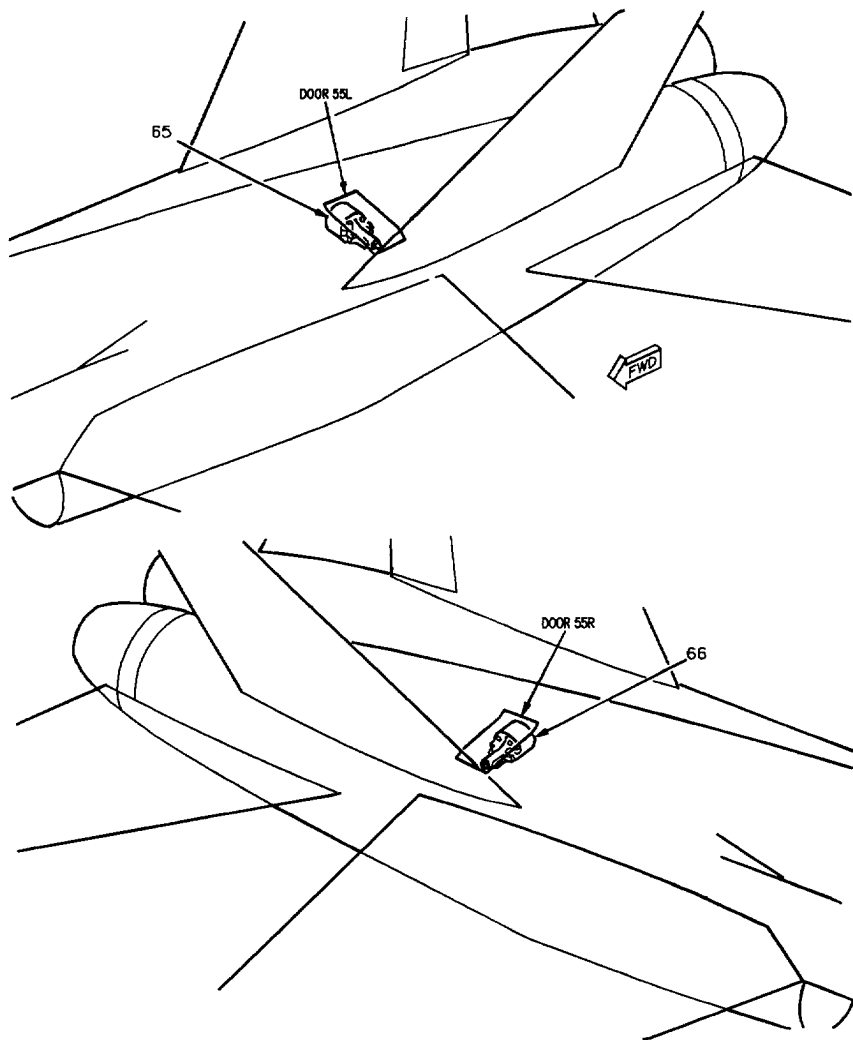


Figure 1. Component Locator (Sheet 12)



03030113

Figure 1. Component Locator (Sheet 13)

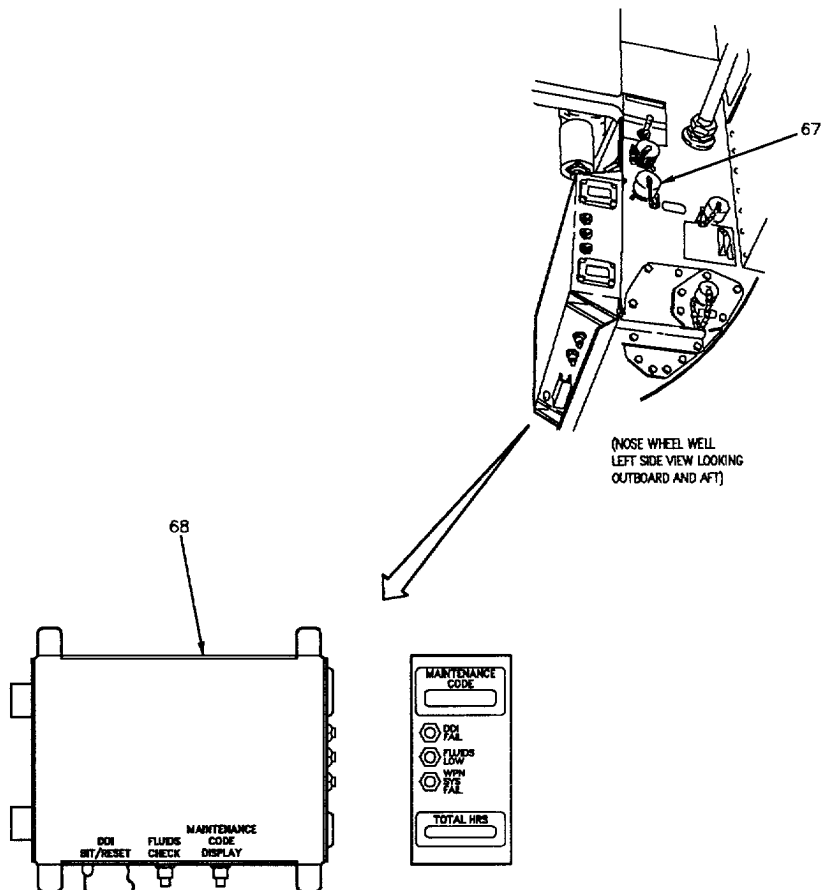


Figure 1. Component Locator (Sheet 14)

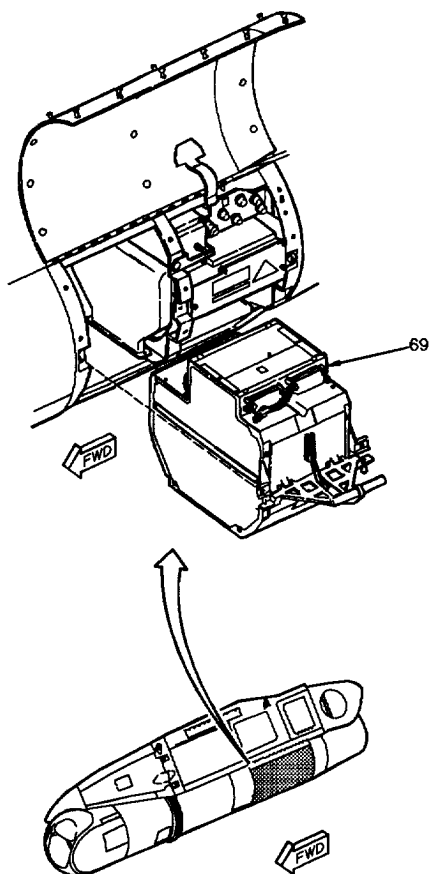
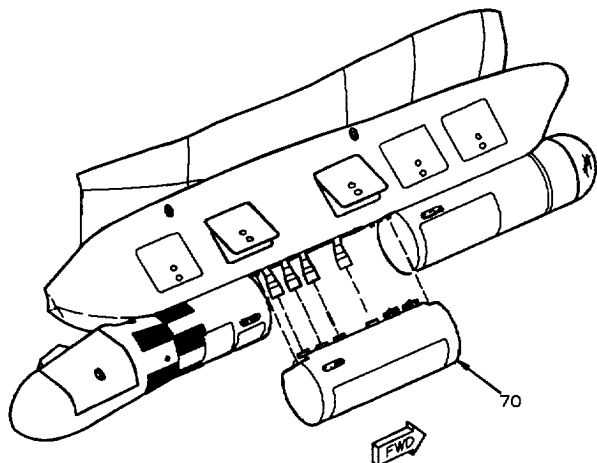


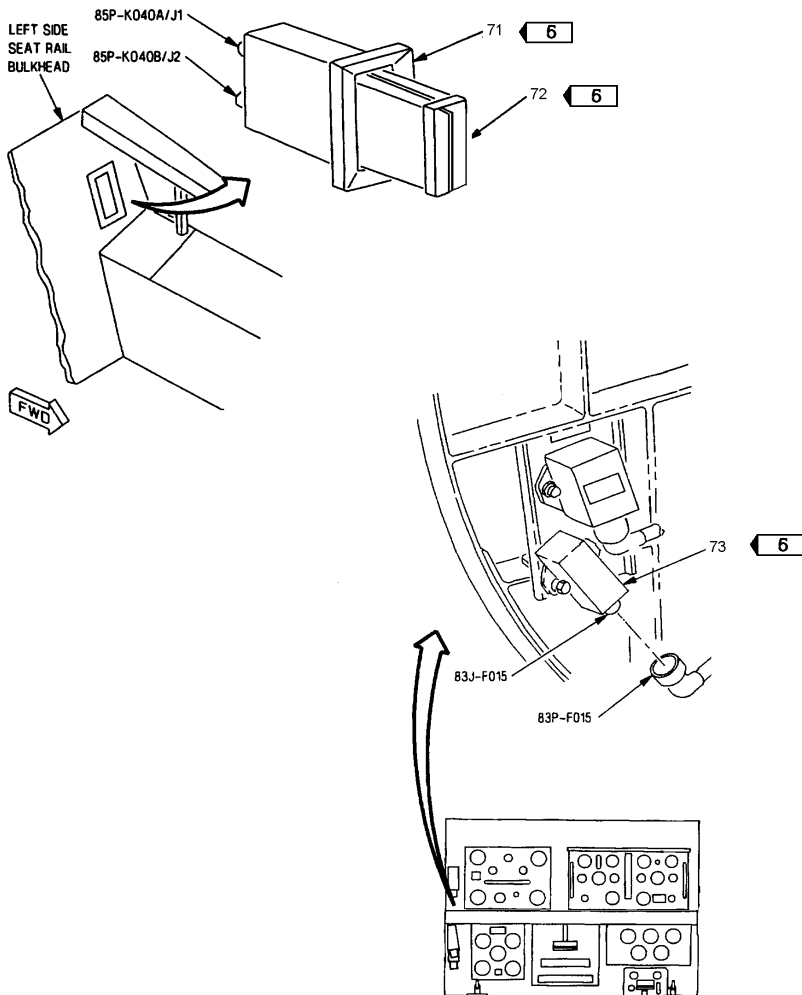
Figure 1. Component Locator (Sheet 15)



LEGEND

1. AIRCRAFT CONNECTOR LOCATIONS
ARE SHOWN IN A1-F18()-WDM-000.

Figure 1. Component Locator (Sheet 16)



D00R 14R

03030117

Figure 1. Component Locator (Sheet 17)

Nomenclature	Index No.	Ref Des
AIRCRAFT EJECTION SEAT	23	25A-H002
AIR DATA COMPUTER	42	70A-F001
ANTENNA SELECTOR SA-2292/A	52	76S-F004
ARMAMENT COMPUTER CP-1342/AYQ-9(V)	47	61A-F001
ATTITUDE REFERENCE INDICATOR ARU-48/A	9	33M-J015
6 AVIONIC MUX BUS IMPEDANCE MATCHING NETWORK	73	83A-F015
CAUTION LIGHT INDICATOR PANEL	11	8A-J042
3 COCKPIT ELECTRIC LIGHT CONTROL	24	8A-L001
COMMAND LAUNCH COMPUTER	37	61A-F010
COMPUTER-POWER SUPPLY CP-1325/APG-65	27	60A-A505
COMPUTER-TRANSPONDER KIT-1A/TSEC	59	78A-E003
CONTROL-CONVERTER	43	82A-F001
CONTROLLER-PROCESSOR C-10661/AAS-38	69	61A-P520
COUNTERMEASURES COMPUTER CP-1293/ALR-67(V)	60	62A-E006
CREW STATION ENGINE MONITOR INDICATOR AEU-12/A	17	3M-H001
DIRECTION FINDER OA-8697/ARD	30	71ARB001
DIGITAL DATA COMPUTER NO. 1	62	83A-E001
DIGITAL DATA COMPUTER NO. 2	51	83A-F002
DIGITAL DISPLAY INDICATOR ID-2150/ASM-612	68	85A-G003
ELECTRONIC EQUIPMENT CONTROL	4	79A-J006
5 EMBEDDED GPS/INS (EGI) UNIT	57	68A-E011
EXTERNAL POWER CONTACTOR	56	1K-C022
FUEL QUANTITY GAGING INTERMEDIATE DEVICE	48	5A-F014
GND PWR CONTROL		
HORIZONTAL INDICATOR IP-1350/A	13	80A-J003
3 IFF SWITCH	26	78S-K005
4 IFF SWITCH	22	78S-L005
INTERCOMMUNICATION AMPLIFIER-CONTROL AM-6979/A OR AM-7360/A	15	76A-H009

Figure 1. Component Locator (Sheet 18)

Nomenclature	Index No.	Ref Des
INTERCONNECTING BOX J-3656/ASQ-173	70	61ARR510
INTERFERENCE BLANKER MX-9965/A	40	A-F001
INTR LT CONTROL BOX PANEL ASSEMBLY	12	8A-J002
1 LANDING GEAR CONTROL UNIT	35	12A-D004
2 LANDING GEAR CONTROL UNIT	35	12A-A004
LEFT DIGITAL DISPLAY INDICATOR IP-1317()	1	80A-H001
LEFT MUX BUS IMPEDANCE MATCHING NETWORK	58	83A-Y013
LEFT POWER CONTACTOR	55	1K-C007
LH ADVISORY AND THREAT WARNING INDICATOR PANEL	2	52A-H073
LOCK/SHOOT LIGHT ASSEMBLY	5	8DSJ150
MC/HYD ISOL CONTROL PANEL ASSEMBLY	14	52A-H081
6 MISSION DATA LOADER MOUNT	71	85A-K503
6 MISSION DATA LOADER	72	85A-K040
MUX TEST CONNECTOR	67	83J-G003
NO. 2 CIRCUIT BREAKER PANEL ASSEMBLY	33	52A-D024
NO. 2 RELAY PANEL ASSEMBLY	46	52A-F058
NO. 4 CIRCUIT BREAKER PANEL ASSEMBLY	32	52A-D026
NO. 4 RELAY PANEL ASSEMBLY	64	52A-N118
NO. 7 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY	54	52A-C057
NO. 8 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY	53	52A-C159
PULSE DECODER KY-651()/ARA-63	41	74A-F002
RADAR RECEIVER R-1623/APN	29	72REB001
RADAR RECEIVER-TRANSMITTER RT-1028/APN-202	31	72A-A002
RADIO RECEIVER R-1379()/ARA-63	28	74REB001
4 REAR COCKPIT ELECTRIC LIGHT CONTROL	20	8A-L097
4 REAR CREW STATION ENGINE MONITOR INDICATOR AEU-12/A	18	3M-K002
4 REAR ELECTRONIC EQUIPMENT CONTROL	19	76A-L028
RECEIVER-TRANSMITTER RT-1015()/APN-194(V)	63	67A-T001
RECEIVER-TRANSMITTER RT-1157()/APX-100(V)	61	78A-E001
RECEIVER-TRANSMITTER RT-1159/A	36	69A-F001
RECEIVER-TRANSMITTER RT-1250()/ARC NO. 1	39	76A-F001
RECEIVER-TRANSMITTER RT-1250()/ARC NO. 2	44	76A-F002
4 RECEIVER-TRANSMITTER PROCESSOR RT-1379()/ASW	21	77A-K001
3 RECEIVER-TRANSMITTER PROCESSOR RT-1379()/ASW	25	77A-L001
RESERVOIR	65	10HPP006

Figure 1. Component Locator (Sheet 19)

Nomenclature	Index No.	Ref Des
RESERVOIR	66	10HPR007
RIGHT DIGITAL DISPLAY INDICATOR IP-1317()	6	80A-J002
RIGHT MUX BUS IMPEDANCE MATCHING NETWORK	50	83A-Y013
RIGHT POWER CONTACTOR	34	1K-D008
ROLL-PITCH-YAW COMPUTER (FCCA)	38	84A-F001
ROLL-PITCH-YAW COMPUTER (FCCB)	45	84A-F002
SIGNAL DATA RECORDER RO-508/ASM-612	49	85A-F001
LEGEND		
1 161353 THRU 161987 BEFORE F/A-18 AFC 48.		
2 162394 AND UP; ALSO 161353 THRU 161987 AFTER F/A-18 AFC 48.		
3 F/A-18A.		
4 F/A-18B.		
5 AFTER F/A-18 AFC 231.		
6 AFTER F/A-18 AFC 225.		

Figure 1. Component Locator (Sheet 20)

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

COMPONENT LOCATOR

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292;
AND AFTER F/A-18 AFC 231 PART 2 OR PART 3

Reference Material

None

Alphabetical Index

Subject

Page No.

Component Locator, Figure 1

2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 48	-	Automatic AC Bus Isolation (ECP MDA-F/A-18-00121)	1 Jun 92	-
F/A-18 AFC 253	-	U.S. Naval Reserves A+ Avionics Upgrade, In- corporation of (ECP MDA-F/A-18 0560R1)	1 Jun 02	-
F/A-18 AFC 292	-	U.S Marine Corps Reserves A+ Avionics Up- grade, Incorporation of (ECP MDA-F/A-18 0583)	1 Jun 02	-
F/A-18 AFC 231 Part 2 or Part 3	-	Embedded Global Positioning System (GPS)/In- ertial Navigation System (INS) (EGI), Incorpora- tion of (ECP MDA-F/A-18 0521)	1 Jun 02	-

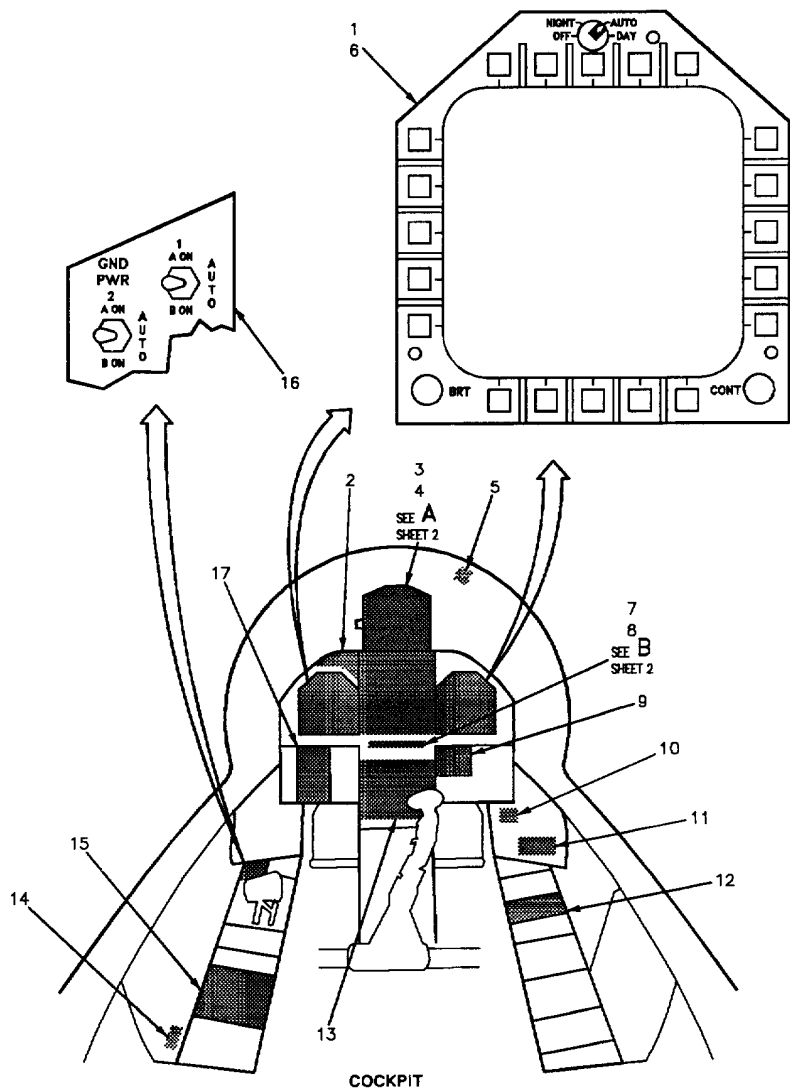


Figure 1. Component Locator (Sheet 1)

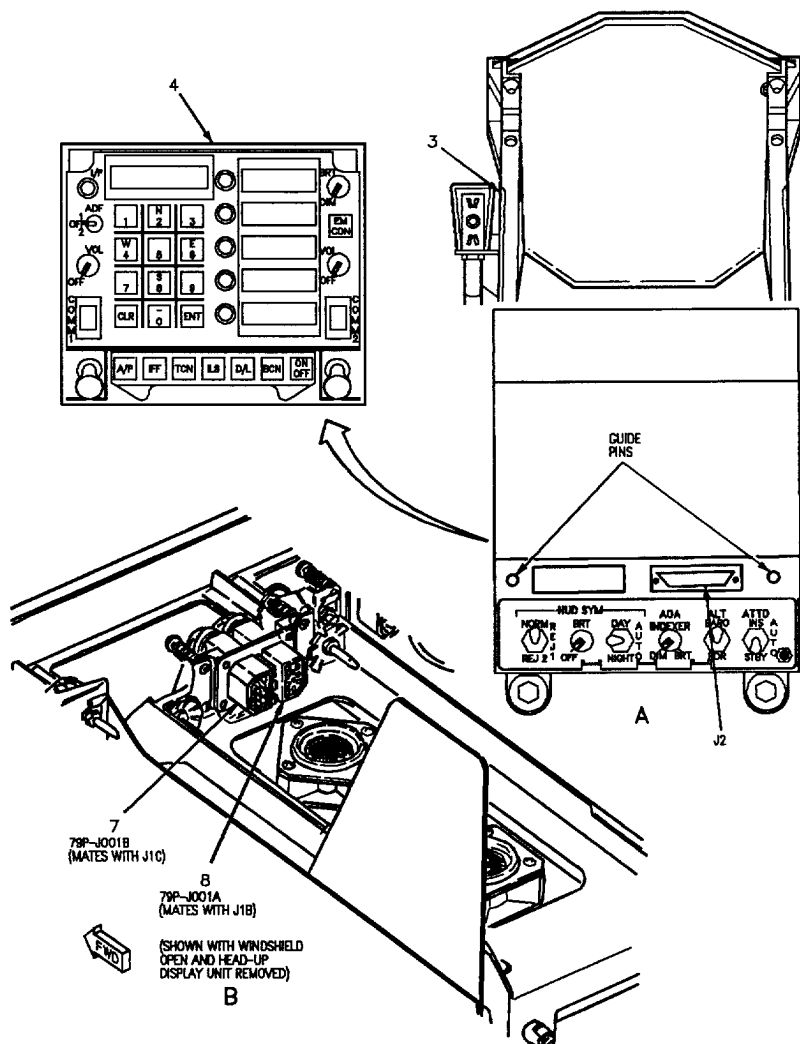


Figure 1. Component Locator (Sheet 2)

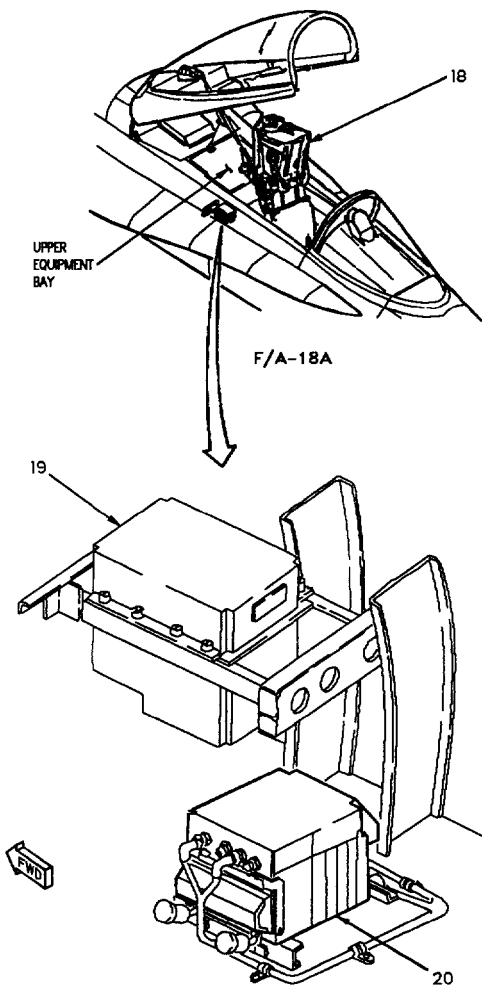


Figure 1. Component Locator (Sheet 3)

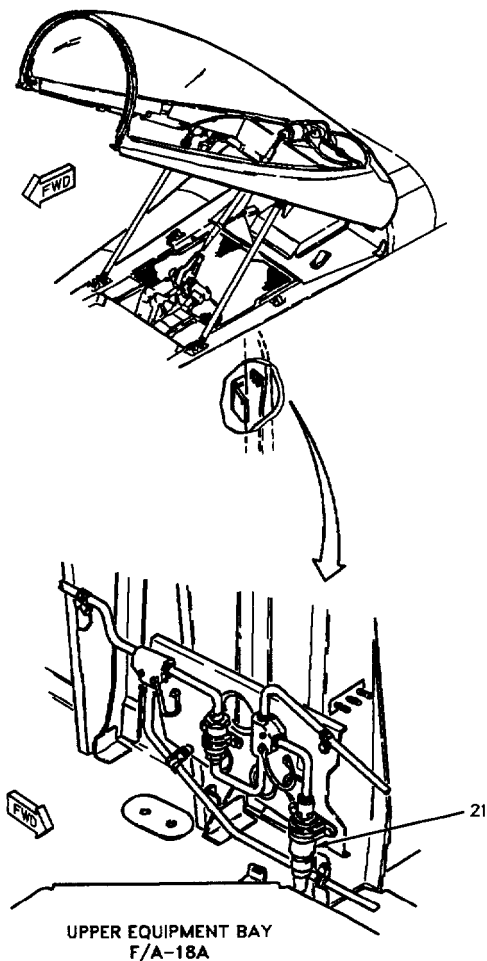


Figure 1. Component Locator (Sheet 4)

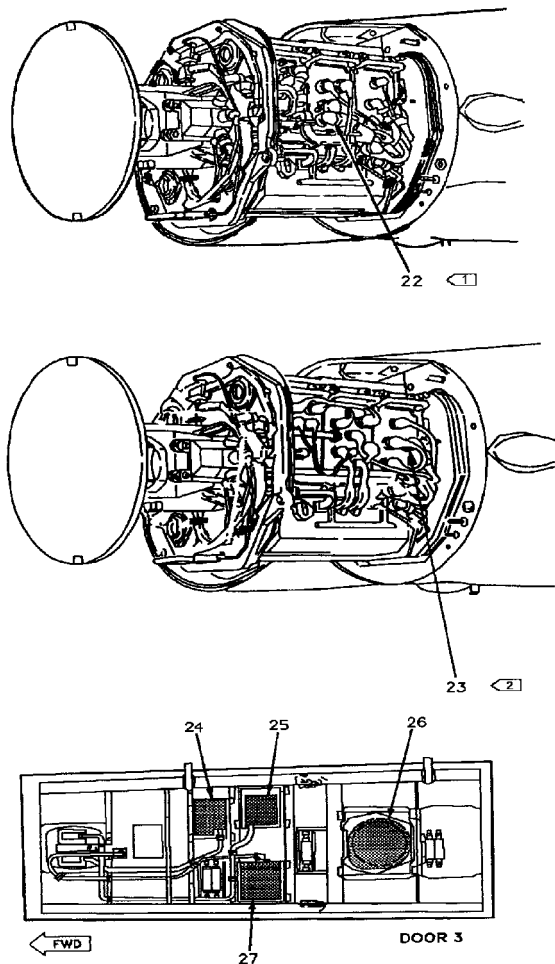


Figure 1. Component Locator (Sheet 5)

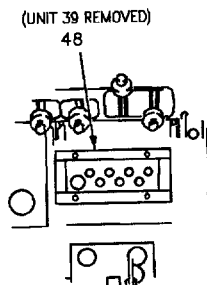
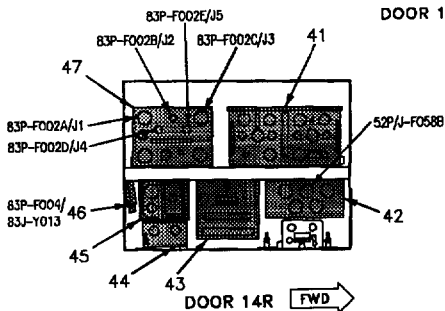
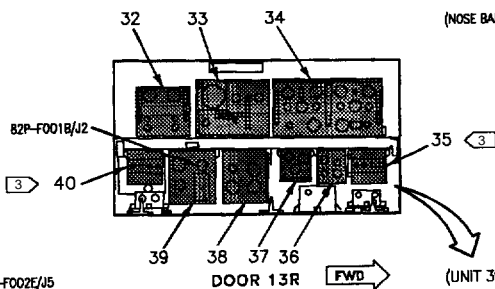
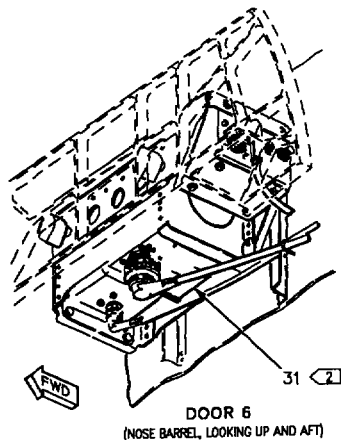
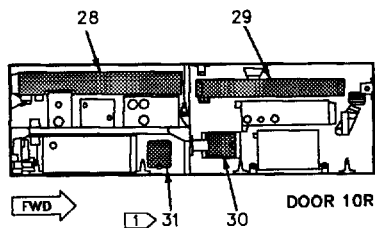
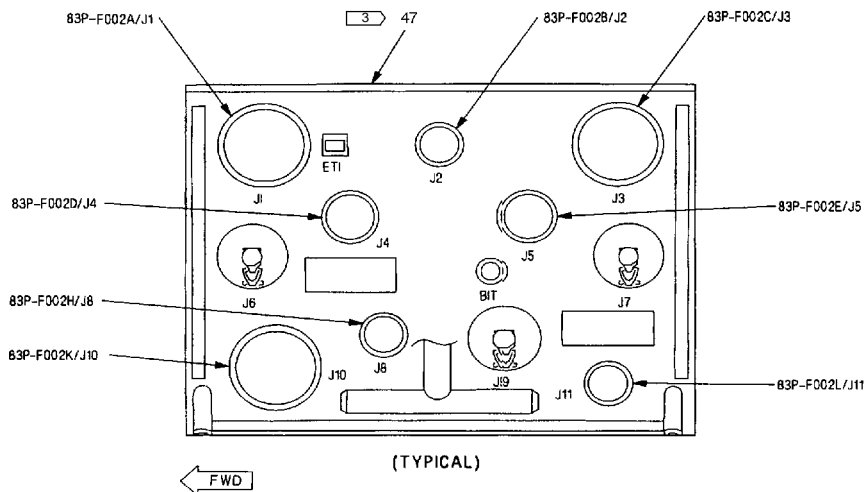


Figure 1. Component Locator (Sheet 6)



03040107

Figure 1. Component Locator (Sheet 7)

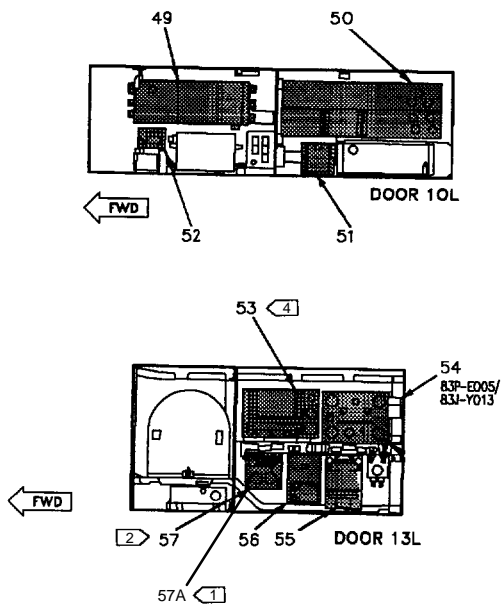


Figure 1. Component Locator (Sheet 8)



03040109

Figure 1. Component Locator (Sheet 9)

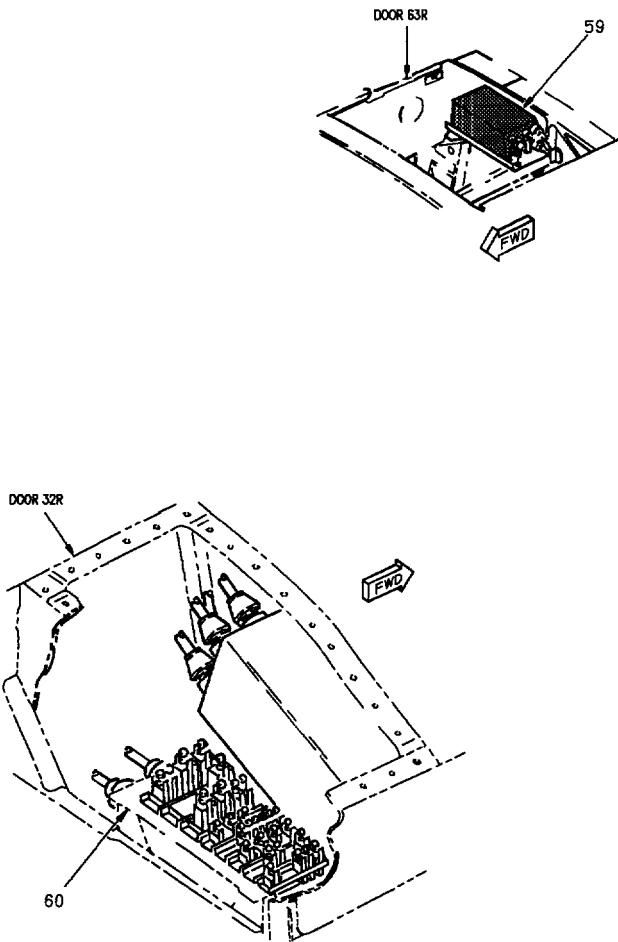


Figure 1. Component Locator (Sheet 10)

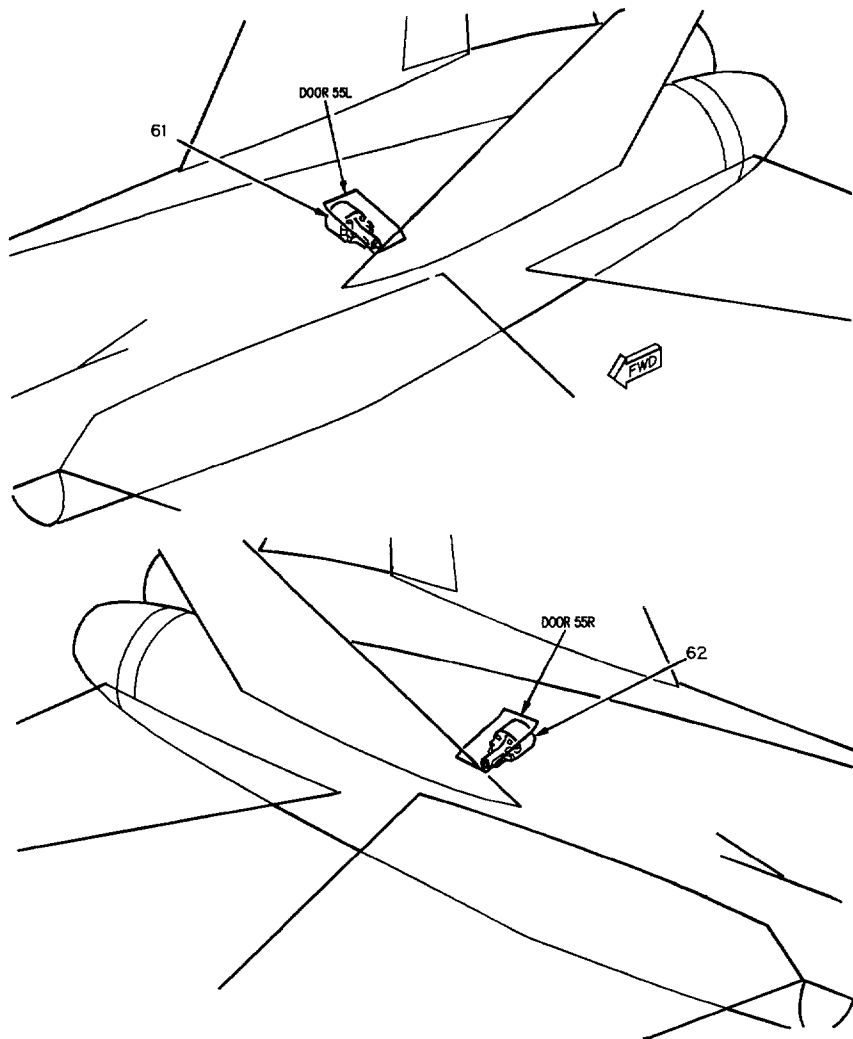


Figure 1. Component Locator (Sheet 11)

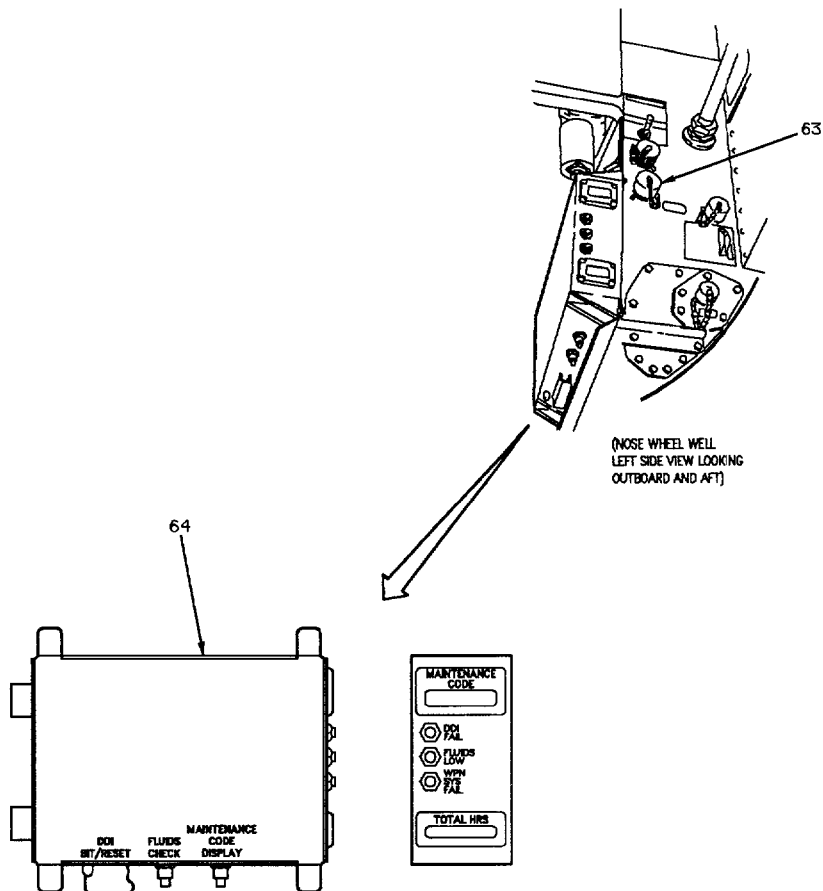


Figure 1. Component Locator (Sheet 12)

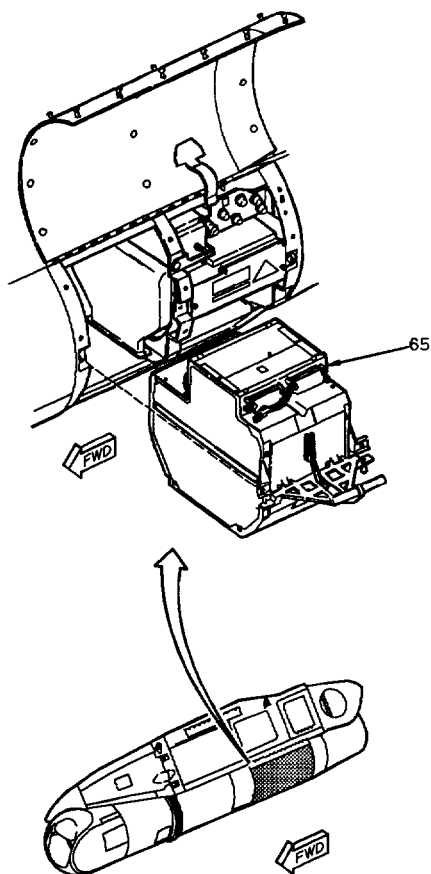
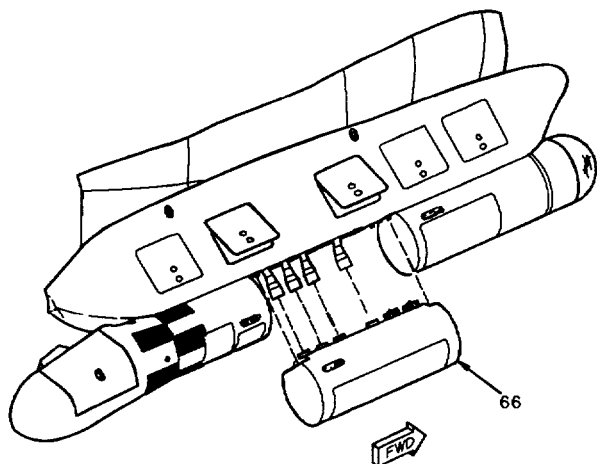


Figure 1. Component Locator (Sheet 13)



LEGEND

1. AIRCRAFT CONNECTOR LOCATIONS
ARE SHOWN IN A1-F18()-WDM-000.

Figure 1. Component Locator (Sheet 14)

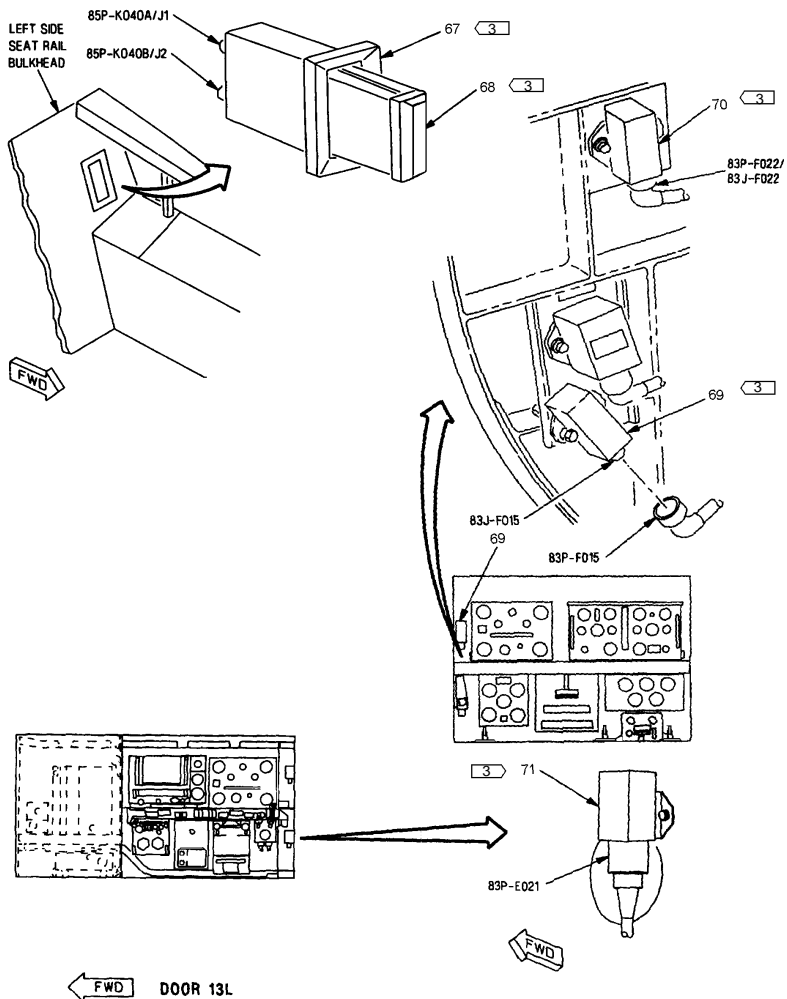


Figure 1. Component Locator (Sheet 15)

Nomenclature	Index No.	Ref Des
AIRCRAFT EJECTION SEAT	18	25A-H002
AIR DATA COMPUTER	38	70A-F001
ANTENNA SELECTOR SA-2292/A	48	76S-F004
ARMAMENT COMPUTER CP-1342/AYQ-9(V)	43	61A-F001
ATTITUDE REFERENCE INDICATOR ARU-48/A	9	33M-J015
3 AVIONIC MUX BUS IMPEDANCE MATCHING NETWORK	69	83A-F015
3 AVIONIC MUX BUS IMPEDANCE MATCHING NETWORK	70	83A-F022
3 AVIONIC MUX BUS IMPEDANCE MATCHING NETWORK	71	83A-E021
CAUTION LIGHT INDICATOR PANEL	11	8A-J042
COCKPIT ELECTRIC LIGHT CONTROL	19	8A-L001
COMMAND LAUNCH COMPUTER	33	61A-F010
1 COMPUTER-POWER SUPPLY CP-1325/APG-65	22	60A-A505
1 COMPUTER-TRANSPONDER KIT-1A/TSEC	55	78A-E003
CONTROL-CONVERTER	39	82A-F001
CONTROLLER-PROCESSOR C-10661/AAS-38	65	61A-P520
COUNTERMEASURES COMPUTER CP-1293/ALR-67(V)	56	62A-E006
CREW STATION ENGINE MONITOR INDICATOR AEU-12/A	17	3M-H001
DIRECTION FINDER OA-8697/ARD	26	71ARB001
3 DIGITAL DATA COMPUTER NO. 1	58	83A-E001
3 DIGITAL DATA COMPUTER NO. 2	47	83A-F002
DIGITAL DISPLAY INDICATOR ID-2150/ASM-612	64	85A-G003
ELECTRONIC EQUIPMENT CONTROL	4	79A-J006
4 EMBEDDED GPS/INS (EGI) UNIT	53	68A-E011
EXTERNAL POWER CONTACTOR	52	1K-C022
FUEL QUANTITY GAGING INTERMEDIATE DEVICE	44	5A-F014
GND PWR CONTROL PANEL ASSEMBLY	16	1A-H004
HEAD-UP DISPLAY	3	79A-J001
HEAD-UP DISPLAY DISCONNECT	7	79P-J001B
HEAD-UP DISPLAY DISCONNECT	8	79P-J001A
HEIGHT INDICATOR ID-2163/A	10	67A-J002
HORIZONTAL INDICATOR IP-1350/A	13	80A-J003
IFF SWITCH	21	78S-K005

Figure 1. Component Locator (Sheet 16)

	Nomenclature	Index No.	Ref Des
	INTERCOMMUNICATION AMPLIFIER-CONTROL AM-6979/A OR AM-7360/A	15	76A-H009
	INTERCONNECTING BOX J-3656/ASQ-173	66	61ARR510
	INTERFERENCE BLANKER MX-9965/A	36	A-F001
	INTR LT CONTROL BOX PANEL ASSEMBLY	12	8A-J002
2	LANDING GEAR CONTROL UNIT	31	12A-D004
	LANDING GEAR CONTROL UNIT	31	12A-A004
	LEFT DIGITAL DISPLAY INDICATOR IP-1317()	1	80A-H001
	LEFT MUX BUS IMPEDANCE MATCHING NETWORK	54	83A-Y013
	LEFT POWER CONTACTOR	51	1K-C007
	LH ADVISORY AND THREAT WARNING INDICATOR PANEL	2	52A-H073
	LOCK/SHOOT LIGHT ASSEMBLY	5	8DSJ150
	MC/HYD ISOL CONTROL PANEL ASSEMBLY	14	52A-H081
3	MISSION DATA LOADER MOUNT	67	85A-K503
3	MISSION DATA LOADER	68	85A-K040
	MUX TEST CONNECTOR	63	83J-G003
	NO. 2 CIRCUIT BREAKER PANEL ASSEMBLY	29	52A-D024
	NO. 2 RELAY PANEL ASSEMBLY	42	52A-F058
	NO. 4 CIRCUIT BREAKER PANEL ASSEMBLY	28	52A-D026
	NO. 4 RELAY PANEL ASSEMBLY	60	52A-N118
	NO. 7 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY	50	52A-C057
	NO. 8 CIRCUIT BREAKER/RELAY PANEL ASSEMBLY	49	52A-C159
	PULSE DECODER KY-651()/ARA-63	37	74A-F002
2	RADAR DATA PROCESSOR CP-2062/APG-73	23	60A-A503
	RADAR RECEIVER R-1623/APN	25	72REB001
	RADAR RECEIVER-TRANSMITTER RT-1028/APN-202	27	72A-A002
	RADIO RECEIVER R-1379()/ARA-63	24	74REB001
2	RADIO RECEIVER-TRANSMITTER RT-1763/APX-111(V)	57	78A-E016
1	RADIO RECEIVER-TRANSMITTER RT-1157A/APX-100(V)	57A	78A-E001
	RECEIVER-TRANSMITTER RT-1015()/APN-194(V)	59	67A-T001
	RECEIVER-TRANSMITTER RT-1159/A	32	69A-F001
	RECEIVER-TRANSMITTER RT-1250()/ARC NO. 1	35	76A-F001
	RECEIVER-TRANSMITTER RT-1250()/ARC NO. 2	40	76A-F002
	RECEIVER-TRANSMITTER PROCESSOR RT-1379()/ASW	20	77A-L001

Figure 1. Component Locator (Sheet 17)

Nomenclature	Index No.	Ref Des
RESERVOIR	61	10HPP006
RESERVOIR	62	10HPR007
RIGHT DIGITAL DISPLAY INDICATOR IP-1317()	6	80A-J002
RIGHT MUX BUS IMPEDANCE MATCHING NETWORK	46	83A-Y013
RIGHT POWER CONTACTOR	30	1K-D008
ROLL-PITCH-YAW COMPUTER (FCCA)	34	84A-F001
ROLL-PITCH-YAW COMPUTER (FCCB)	41	84A-F002
SIGNAL DATA RECORDER RO-508/ASM-612	45	85A-F001
3 VHF/UHF RECEIVER-TRANSMITTER NO. 1	35	76A-F041
3 VHF/UHF RECEIVER-TRANSMITTER NO. 2	40	76A-F042
LEGEND		
1 F/A-18A AFTER AFC 253 (USNR A+)		
2 F/A-18A AFTER AFC 292 (USMCRA+)		
3 F/A-18A AFTER AFC 253 OR AFC 292		
4 F/A-18A AFTER AFC 231 PART 2 OR AFC 231 PART 3		

Figure 1. Component Locator (Sheet 18)

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - AVIONIC MUX CHANNEL 1

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292 AND F/A-18B

This WP supersedes WP004 00, dated 1 June 1992.

Reference Material

None

Alphabetical Index

Subject	Page No.
Avionic Mux Channel 1 Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 48	-	Automatic AC Bus Isolation (ECP MDA-F/A-18-00121)	1 Jun 92	-

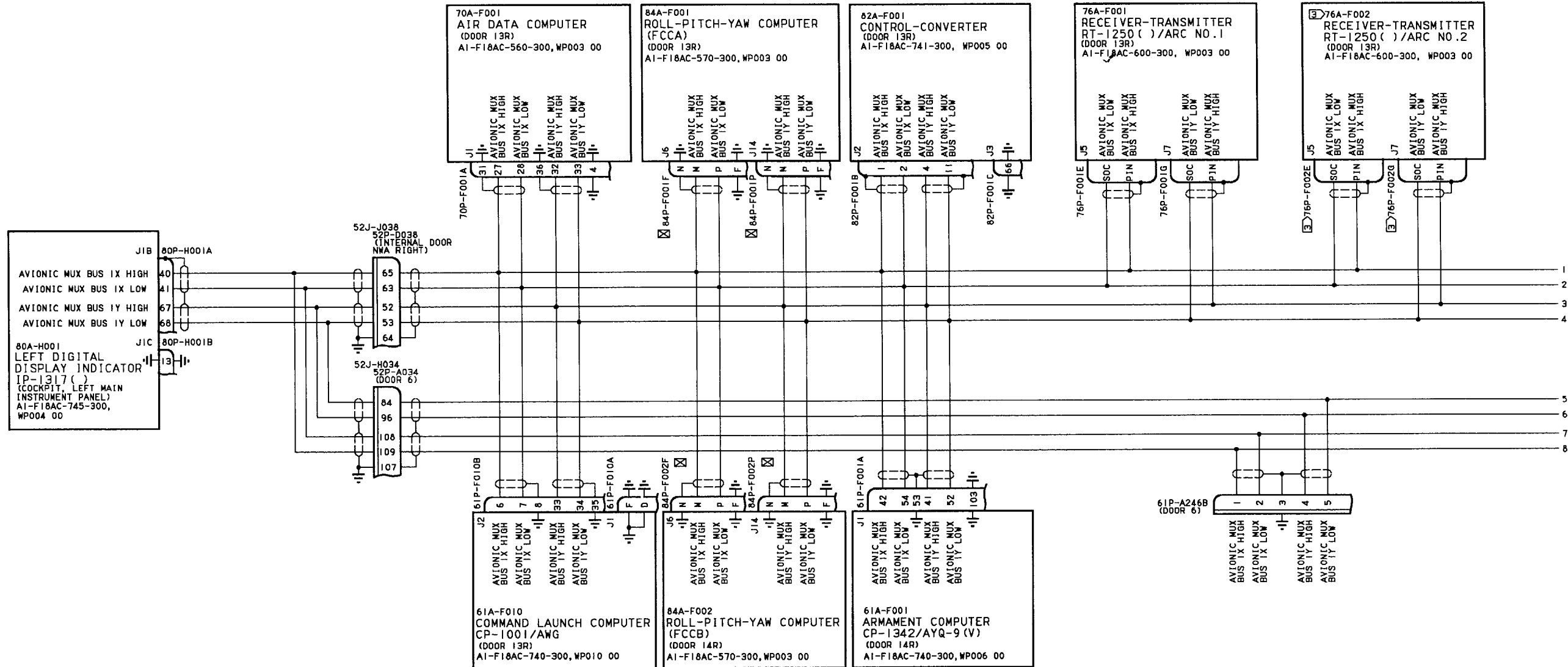


Figure 1.

Figure 1. Avionic Mux Channel 1 Schematic (Sheet 1)

Figure 1.

Change 2

Page 3



Figure 1. Avionic Mux Channel 1 Schematic (Sheet 2)

LEGEND**1. CONTINUITY TESTS:**

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

3 161353 THRU 161528.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - AVIONIC MUX CHANNEL 1

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

Alphabetical Index

Subject	Page No.
Avionic Mux Channel 1 Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jan 01	-

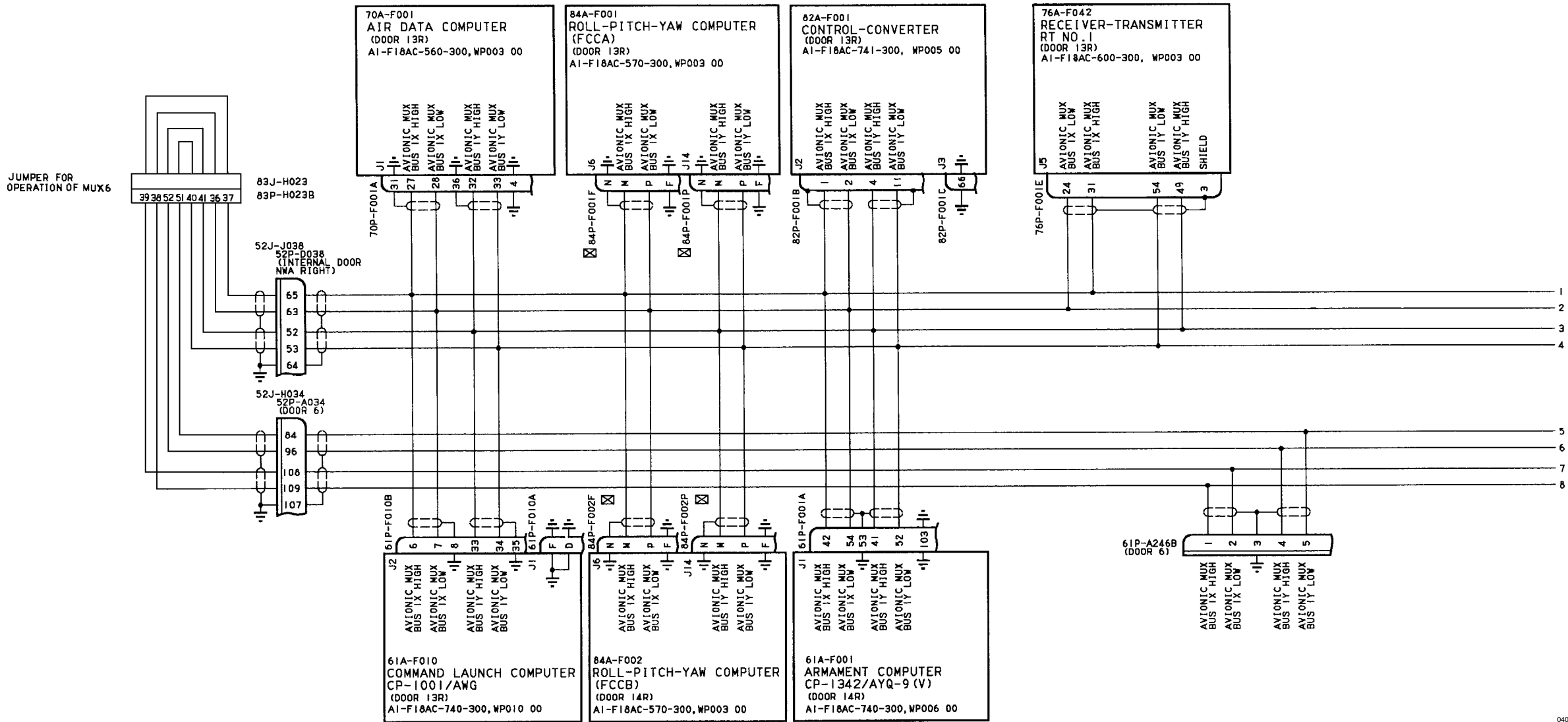


Figure 1.

Figure 1. Avionic Mux Channel 1 Schematic (Sheet 1)

Figure 1.



Figure 1. Avionic Mux Channel 1 Schematic (Sheet 2)

LEGEND**1. CONTINUITY TESTS:**

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

3

 DELETED.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - AVIONIC MUX CHANNEL 2

MISSION COMPUTER SYSTEM

This WP supersedes WP 005 00, dated 1 January 2001.

Title	Work Package
Schematic - Avionic MUX Channel 2 (F/A-18A/B)	005 01
Schematic - Avionic MUX Channel 2 (AFTER F/A-18 AFC 225 AND F/A-18 AFC 231)	005 02
Schematic - Avionic MUX Channel 2 (AFTER F/A-18 AFC 253 OR F/A-18 AFC 292; AND AFTER AFC 231 PART 2 OR PART 3)	005 03

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - AVIONIC MUX CHANNEL 2

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A/B

This WP supersedes WP 005 01, dated 1 January 2001.

Reference Material

None

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Record of Applicable Technical Directives

None

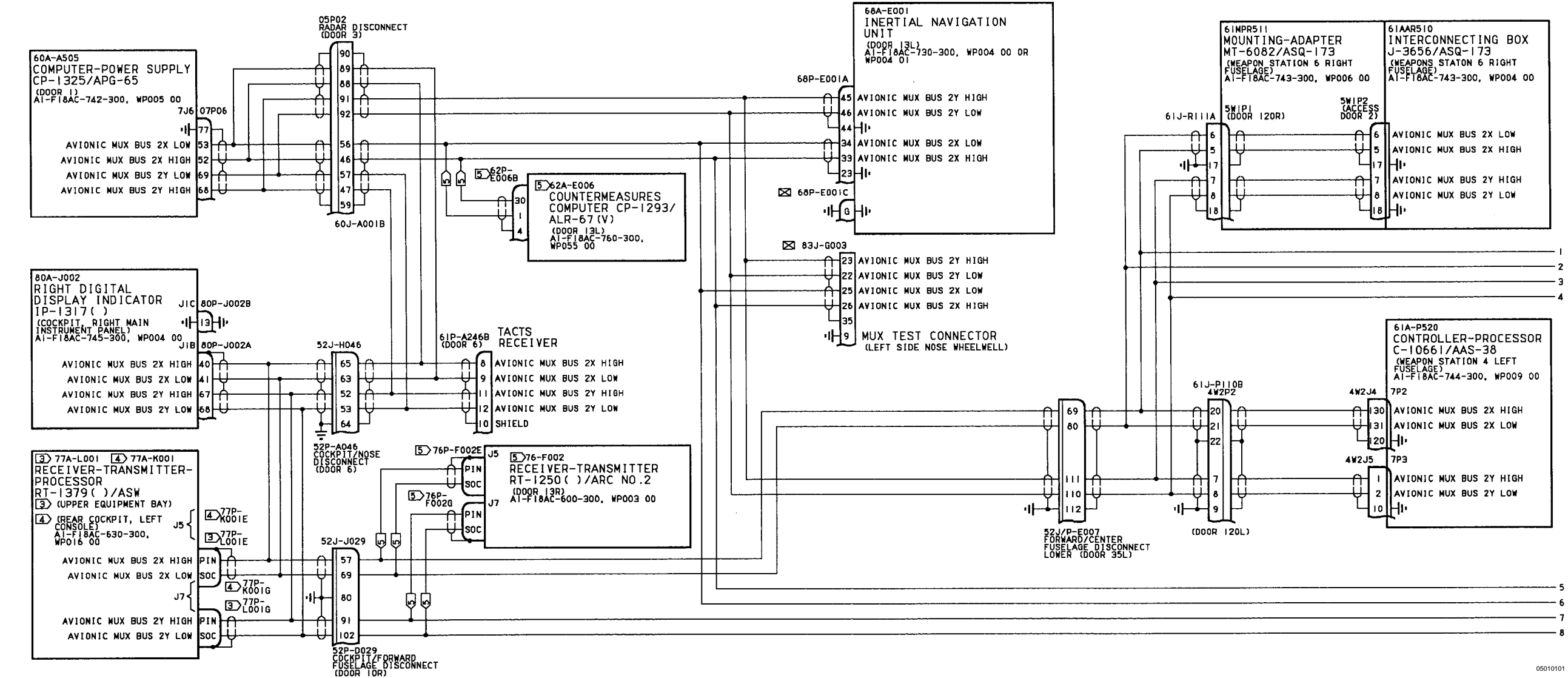


Figure 1. Avionic Mux Channel 2 Schematic (Sheet 1)

Figure 1.

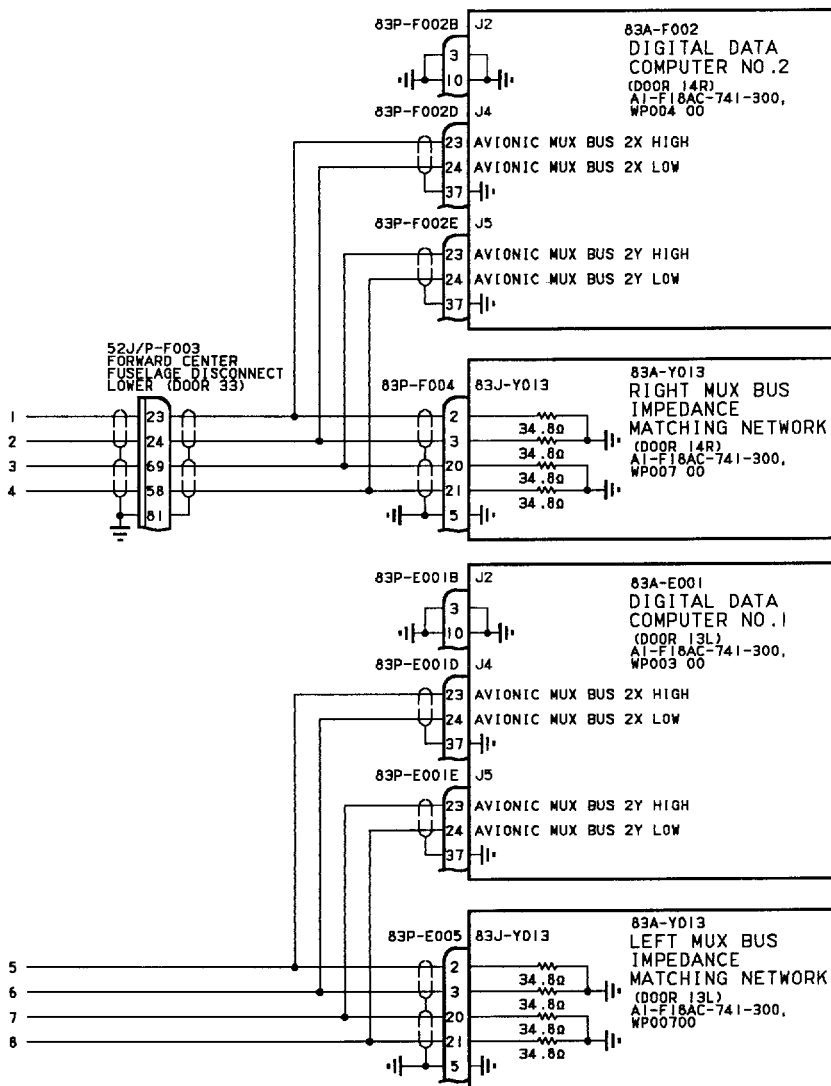


Figure 1. Avionic Mux Channel 2 Schematic (Sheet 2)

LEGEND**1. CONTINUITY TESTS:**

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.**3** F/A-18A.**4** F/A-18B.**5** 161702 AND UP.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - AVIONIC MUX CHANNEL 2****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AND F/A-18B AFTER F/A-18 AFC 225 OR F/A-18 AFC 231**

Reference Material

None

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Avionic Mux Channel 2 Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 225	-	Five (5) Avionics Multiplex Bus Upgrade Incorporation of (ECP MDA-F/A-18 0529)	1 Jun 02	-
F/A-18 AFC 231	-	Embedded Global Positioning System (GPS)/In- ertial Navigation System (INS) (EGI), Incorpora- tion of (ECP MDA-F/A-18 0521)	1 Jun 02	-

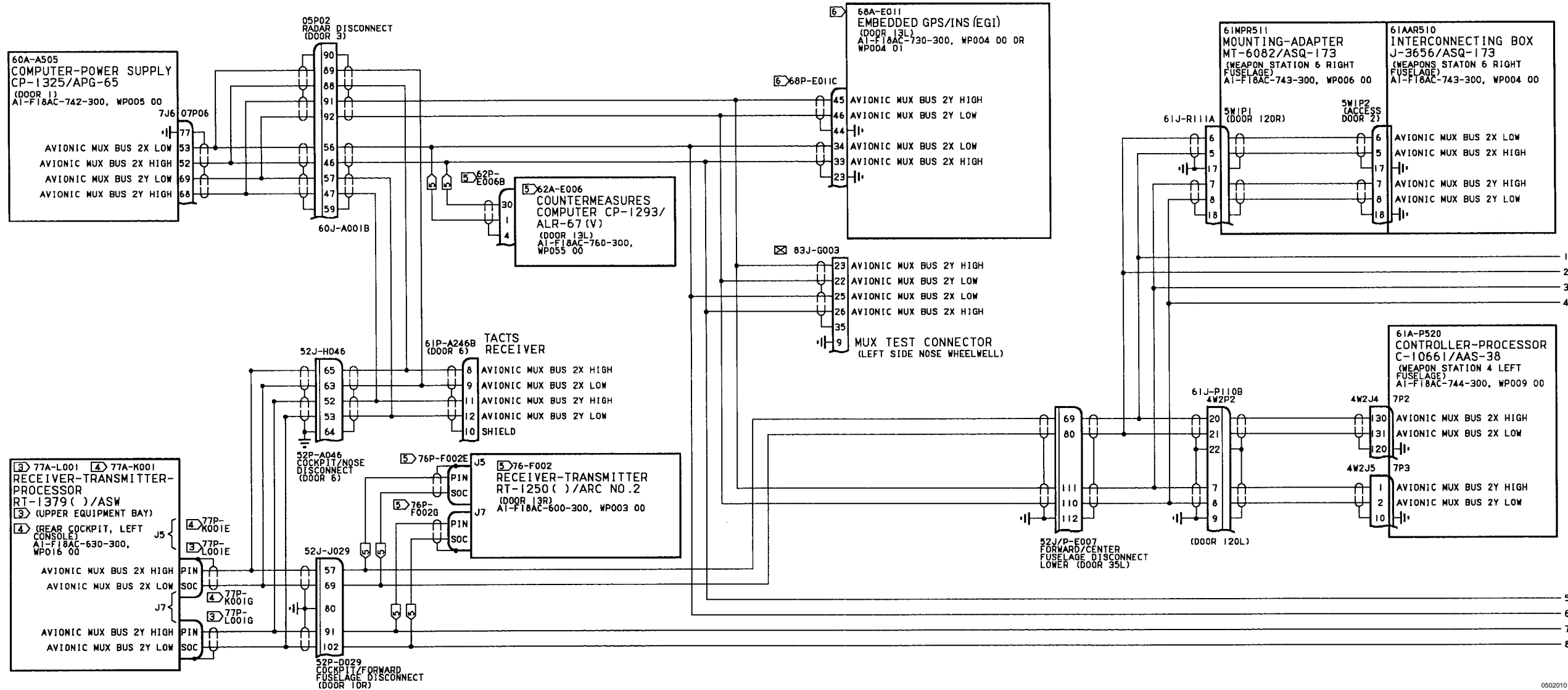


Figure 1.

Figure 1. Avionic Mux Channel 2 Schematic (Sheet 1)

Figure 1.

Change 3

Page 3

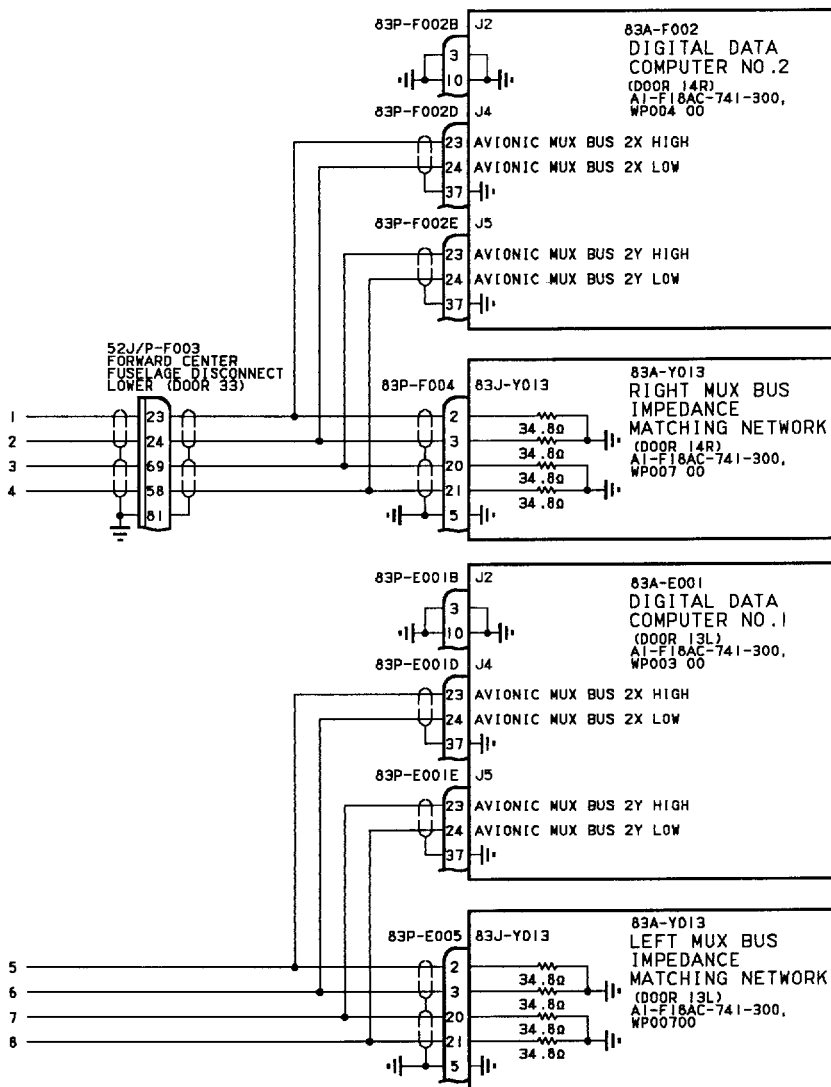


Figure 1. Avionic Mux Channel 2 Schematic (Sheet 2)

05020102

LEGEND

1. CONTINUITY TESTS:

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
 - B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
 - C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
 - E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

- 3 F/A-18A.
- 4 F/A-18B.
- 5 161702 AND UP.
- 6 F/A-18A AND F/A-18B AFTER F/A-18 AFC 231.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - AVIONIC MUX CHANNEL 2****MISSION COMPUTER SYSTEM**

**EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292; AND
AFTER F/A-18 AFC 231 PART 2 OR 3**

Reference Material

None

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Avionic Mux Channel 2 Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jan 01	-
F/A-18 AFC 231 Part 2 or Part 3	-	Embedded Global Positioning System (GPS)/Inertial Navigation System (INS) (EGI), Incorporation of (ECP MDA-F/A-18 0521)	1 Jun 02	-

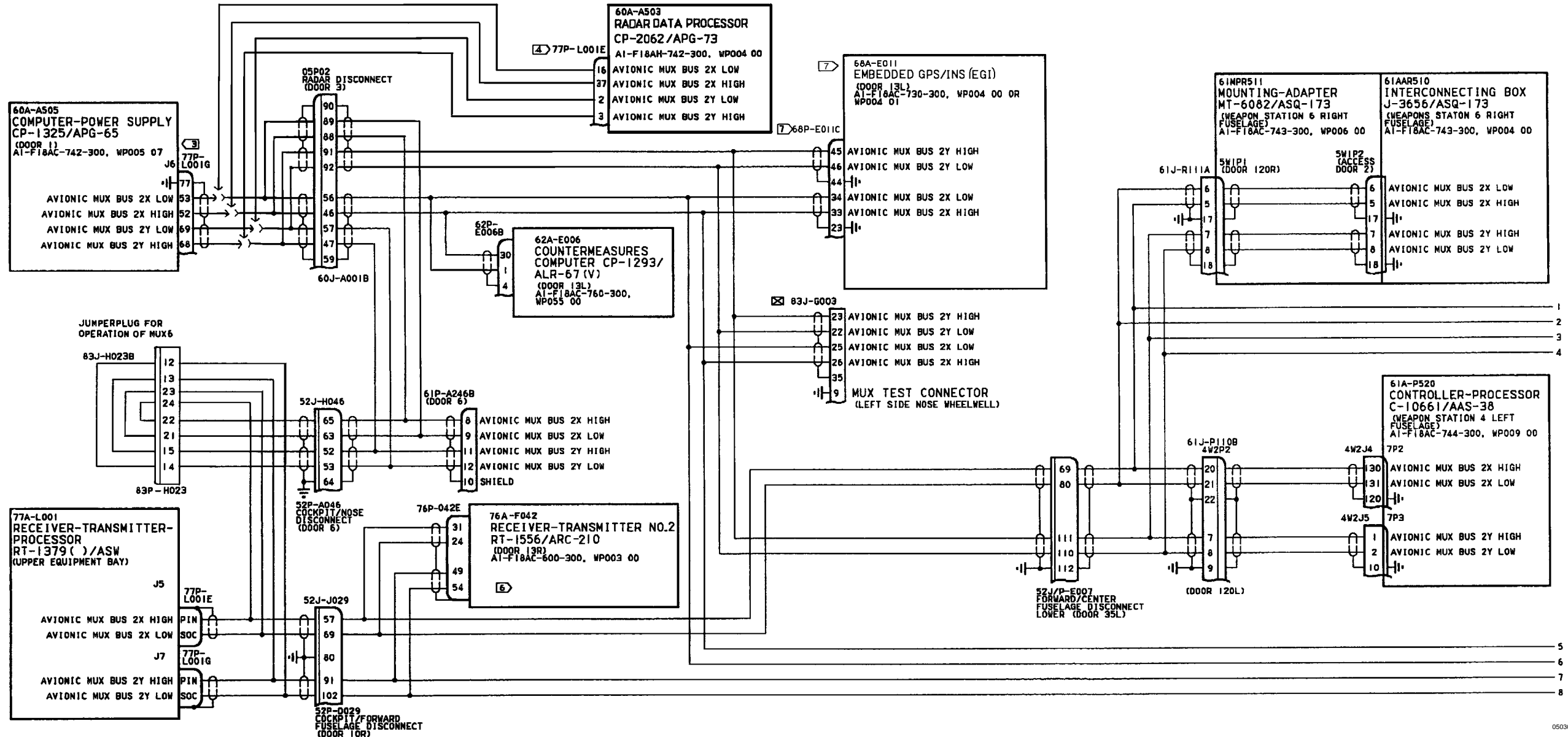


Figure 1.

Figure 1. Avionic Mux Channel 2 Schematic (Sheet 1)

Figure 1.

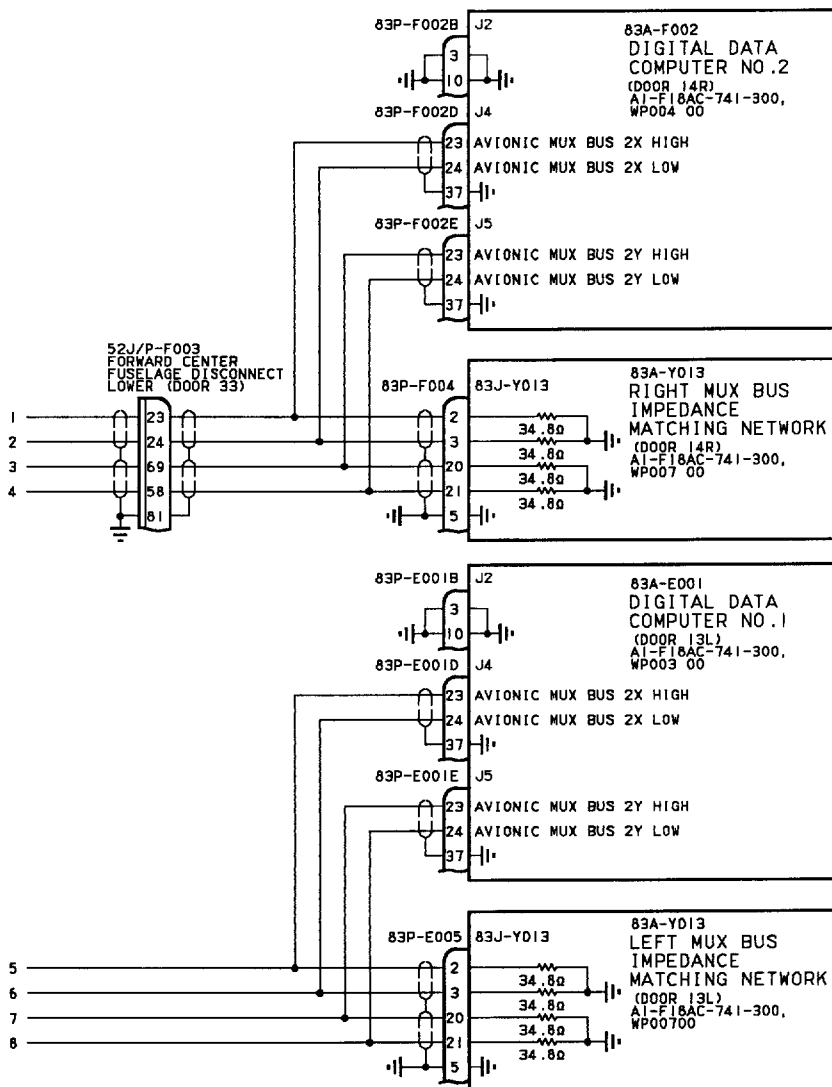


Figure 1. Avionic Mux Channel 2 Schematic (Sheet 2)

LEGEND**1. CONTINUITY TESTS:**

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

- 3** AFTER F/A-18 AFC 253.
- 4** AFTER F/A-18 AFC 292.
- 5** DELETED.
- 6** CONNECTOR STOWED WITH DCS COMM2 INSTALLED.
- 7** AFTER F/A-18 AFC 231 PART 2 OR PART 3.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - AVIONIC MUX CHANNEL 3

MISSION COMPUTER SYSTEM

Reference Material

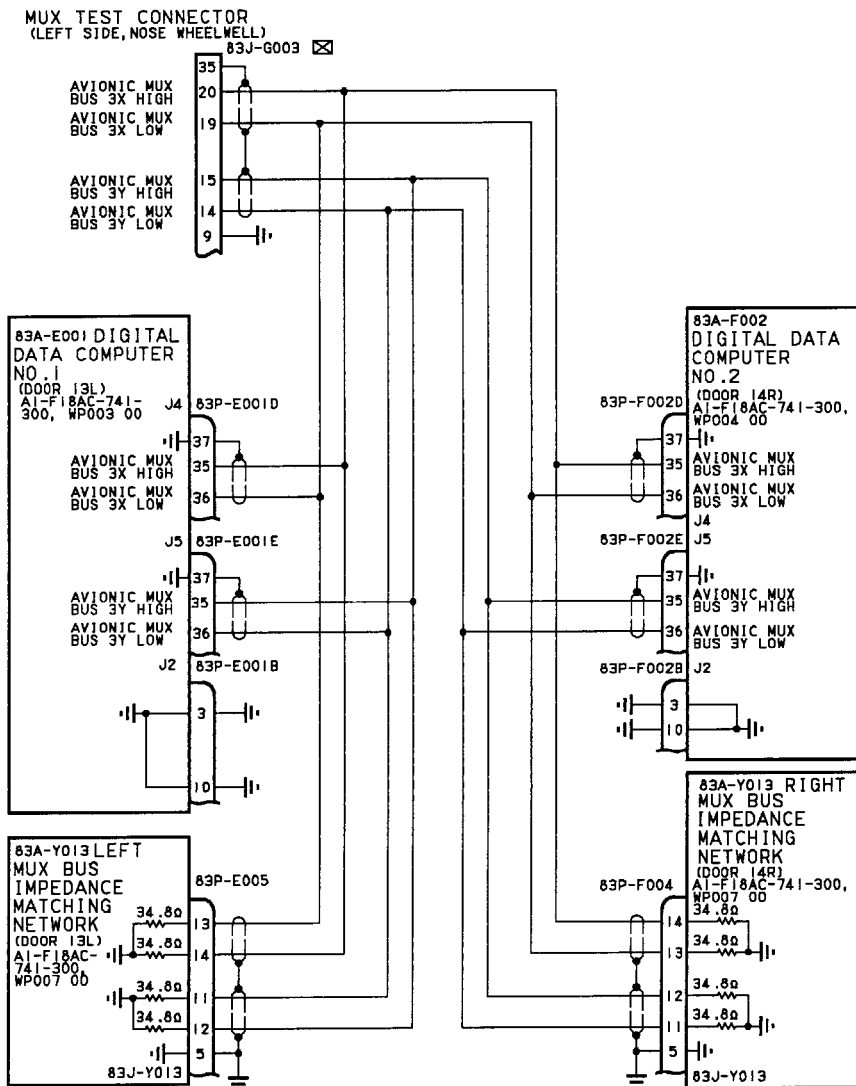
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Record of Applicable Technical Directives

None



LEGEND

1. CONTINUITY TESTS:

- ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18AC()-WDM-000.
- WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY ⊕) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON THE RXI SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RXI SCALE.
- WHEN TESTING CONTINUITY, TEST FOR:
 - SHORTS TO GROUND.
 - SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - SHIELD CONTINUITY.
- WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY ☒). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. NONSTANDARD SYMBOLS:

- ⊕ IDENTIFIES RELAY USED TO SWITCH LOW LEVEL CURRENT, SEE NOTE 1.
- ☒ IDENTIFIES 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS OF THE CONNECTOR. SEE NOTE 1.

Figure 1.

Figure 1. Avionic Mux Channel 3 Schematic

Figure 1.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

**SCHEMATIC - CONTROL-CONVERTER C-10382/A AND
ELECTRONIC EQUIPMENT CONTROL C-10380/ASQ INTERCONNECT**

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292 AND F/A-18B

This WP supersedes WP 007 00, dated 1 June 1992.

Reference Material

None

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Subject	Page No.
Control-Converter C-10382/A and Electronic Equipment Control C-10380/ASQ Interconnect Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 39	-	No. 1 Fuel Tank Interconnect Valve Replacement and Fuel Sequencing Modification (ECP MDA-F/A-18-00072C1)	1 Apr 87	-
F/A-18 AFC 48	-	Automatic AC Bus Isolation (ECP MDA-F/A-18-00121)	1 Jun 92	-

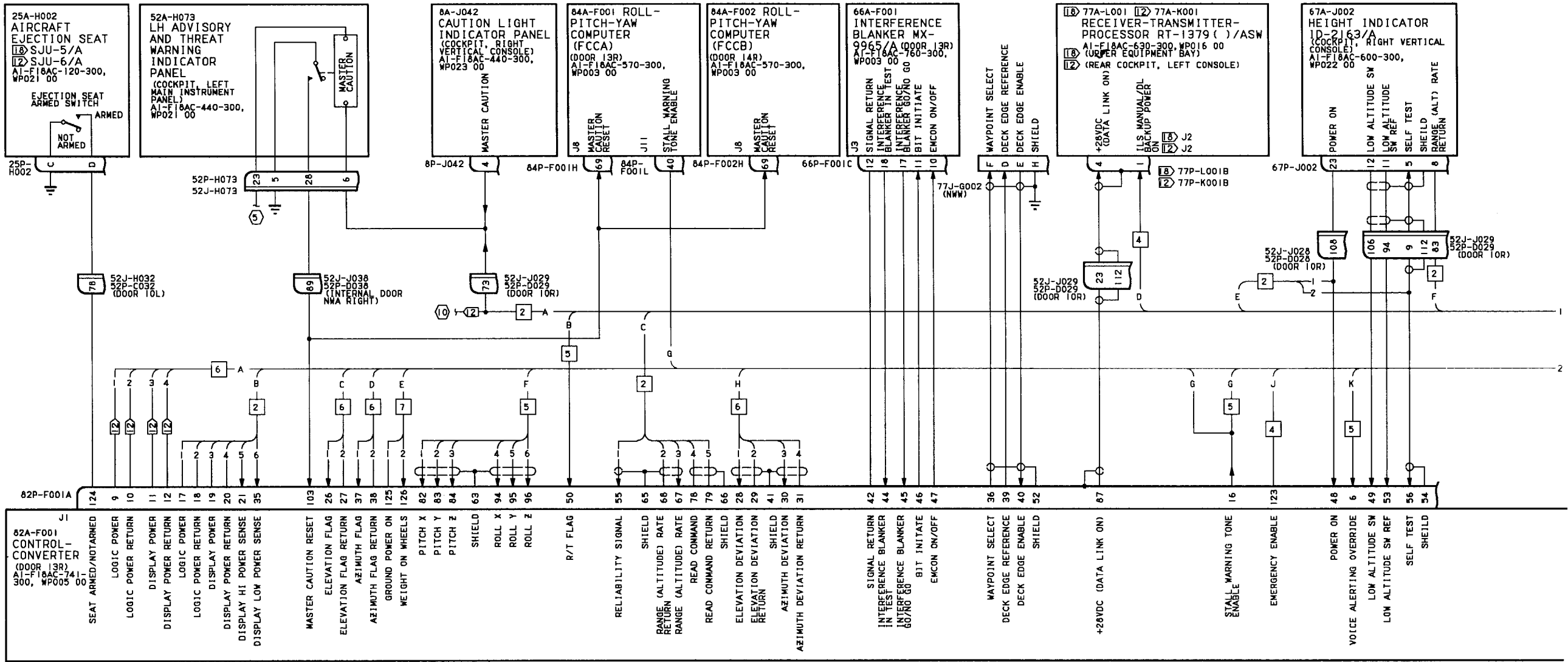


Figure 1.

Figure 1. Control-Converter C-10382/A and Electronic Equipment Control C-10380/ASQ Interconnect Schematic (Sheet 1)

Figure 1.

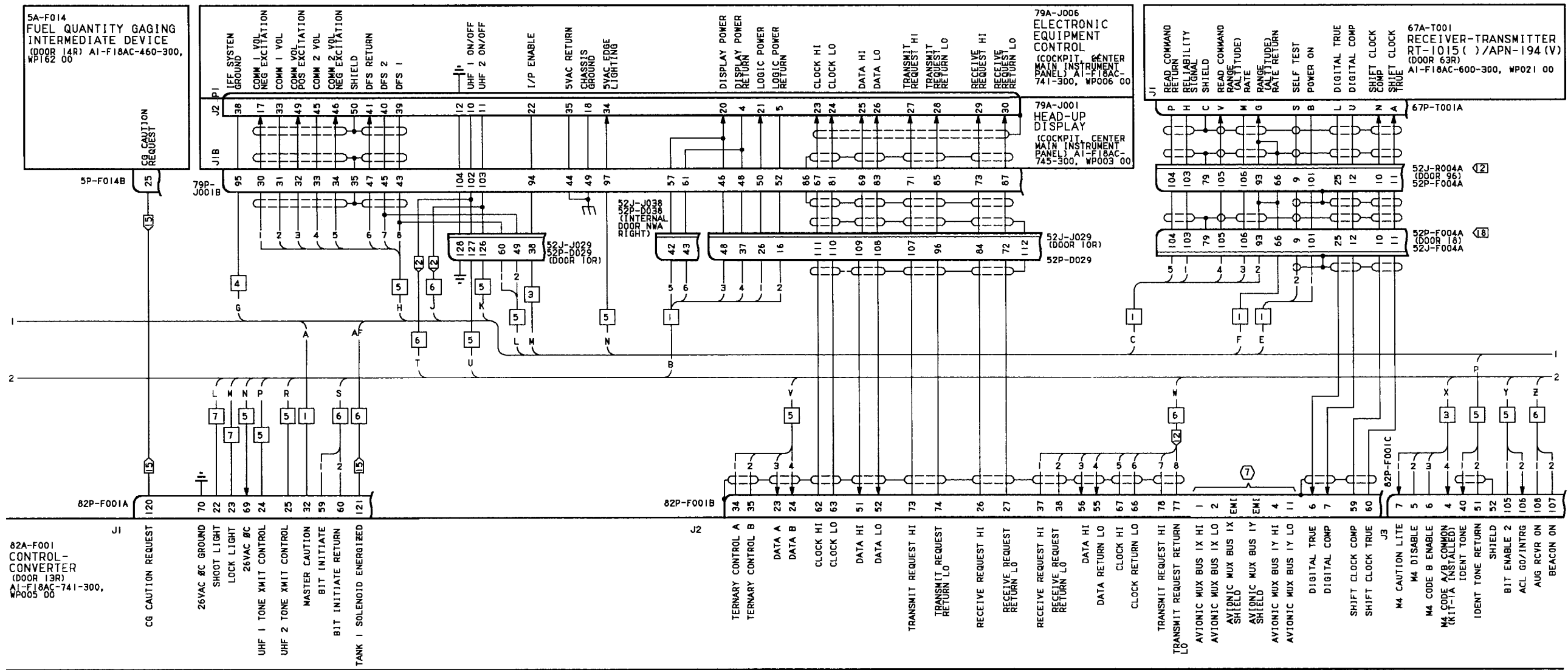


Figure 1.

Figure 1. Control Converter C-10382/A and Electronic Equipment Control C-10380/ASQ Interconnect Schematic (Sheet 2)

Figure 1.

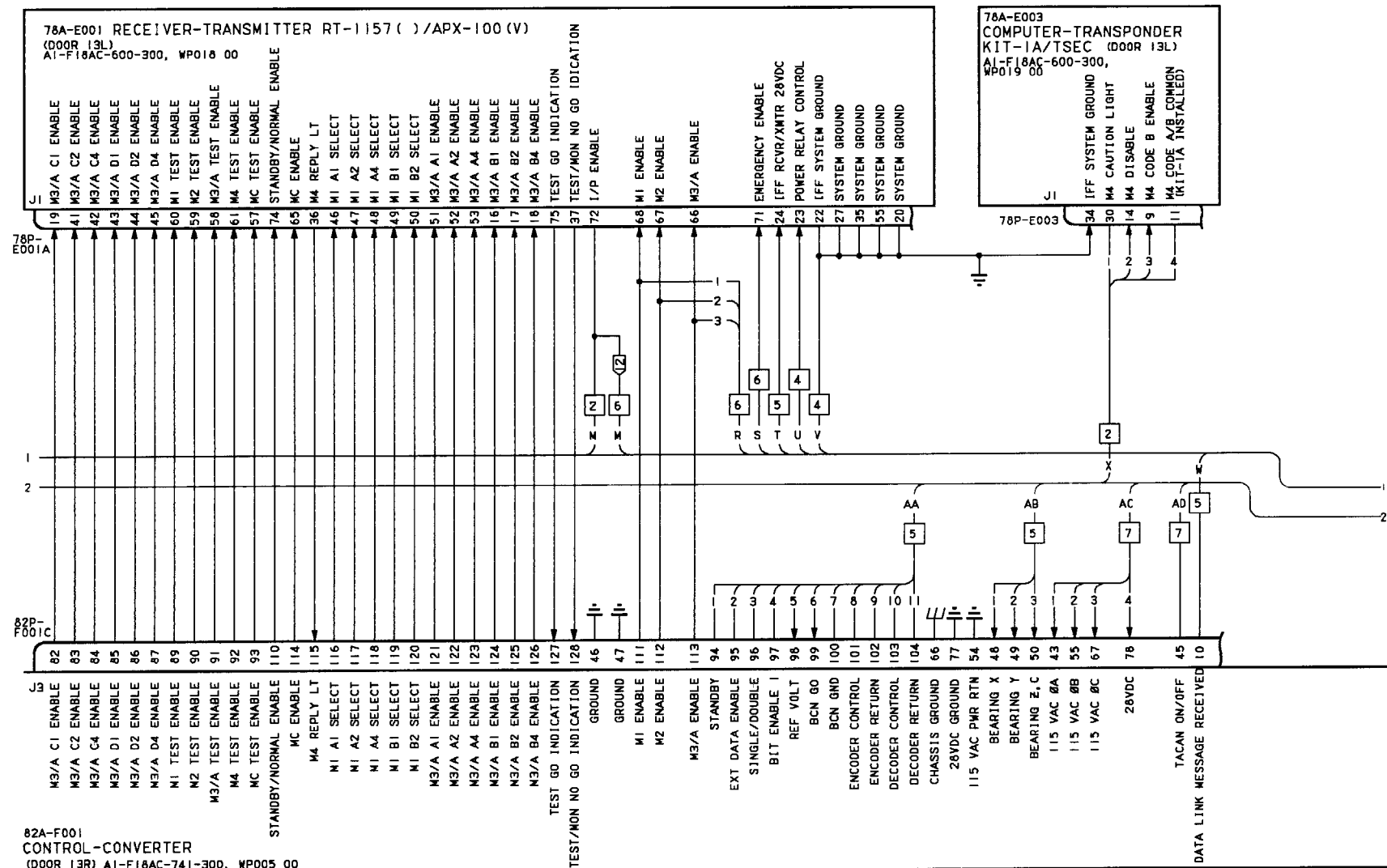


Figure 1.

Figure 1. Control-Converter C-10382/A and Electronic Equipment Control C-10380/ASQ Interconnect Schematic (Sheet 3)

Figure 1.

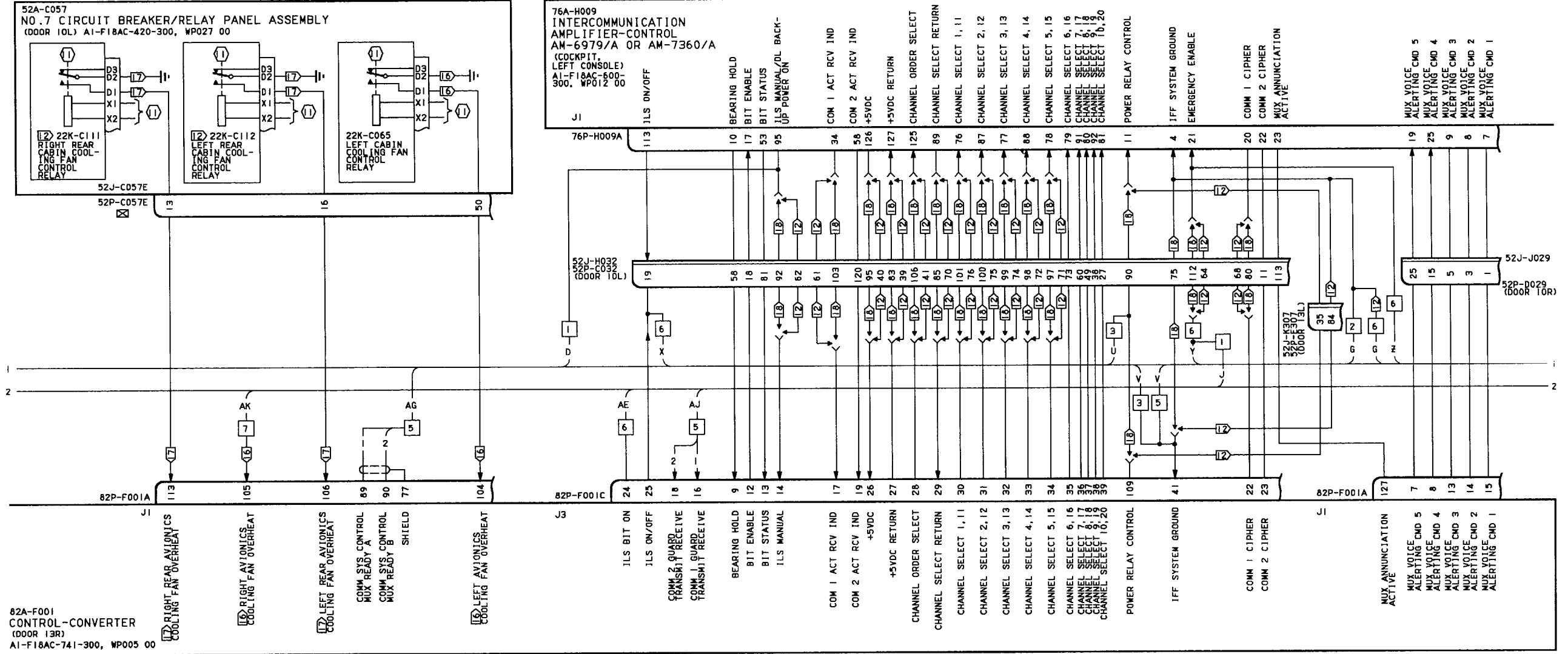


Figure 1.

Figure 1. Control-Converter C-10382/A and Electronic Equipment Control C-10380/ASQ Interconnect Schematic (Sheet 4)

Figure 1.

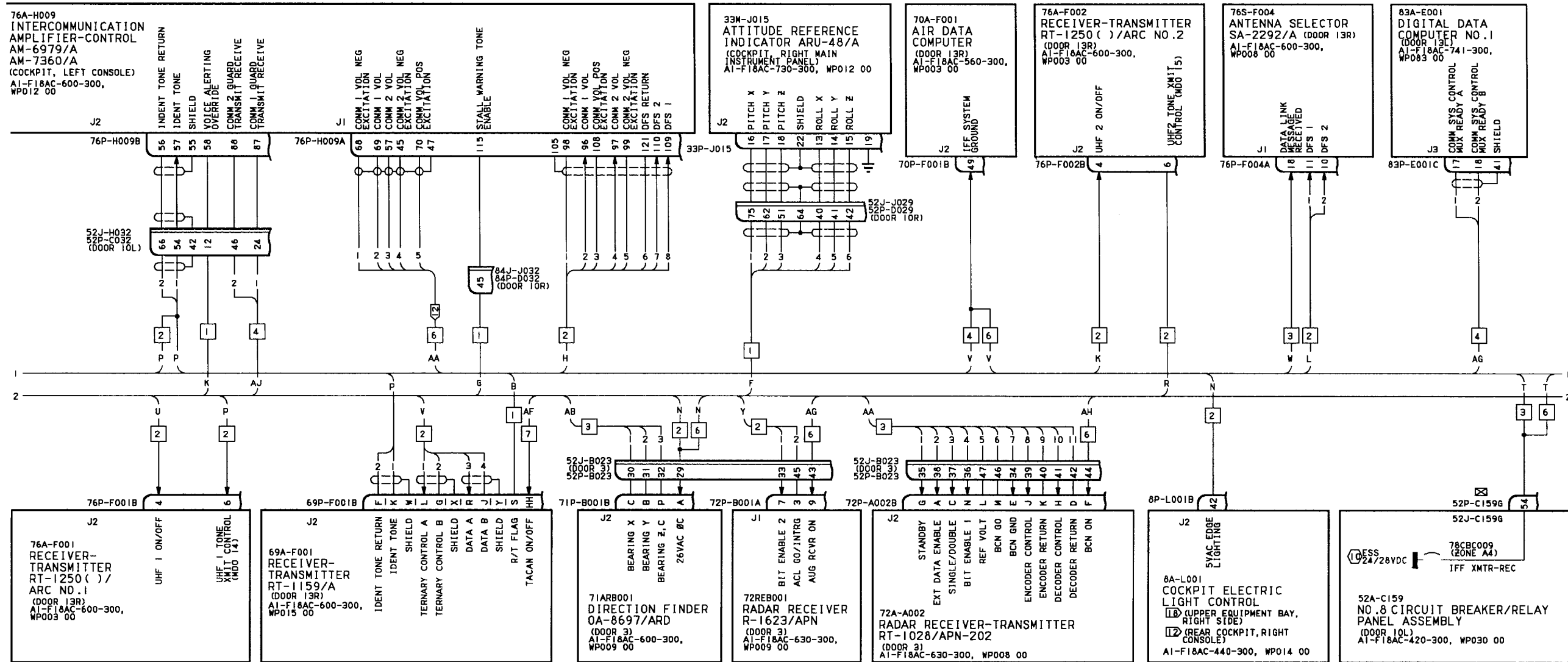


Figure 1.

Figure 1. Control-Converter C-10382/A and Electronic Equipment Control C-10380/ASQ Interconnect Schematic (Sheet 5)

Figure 1.

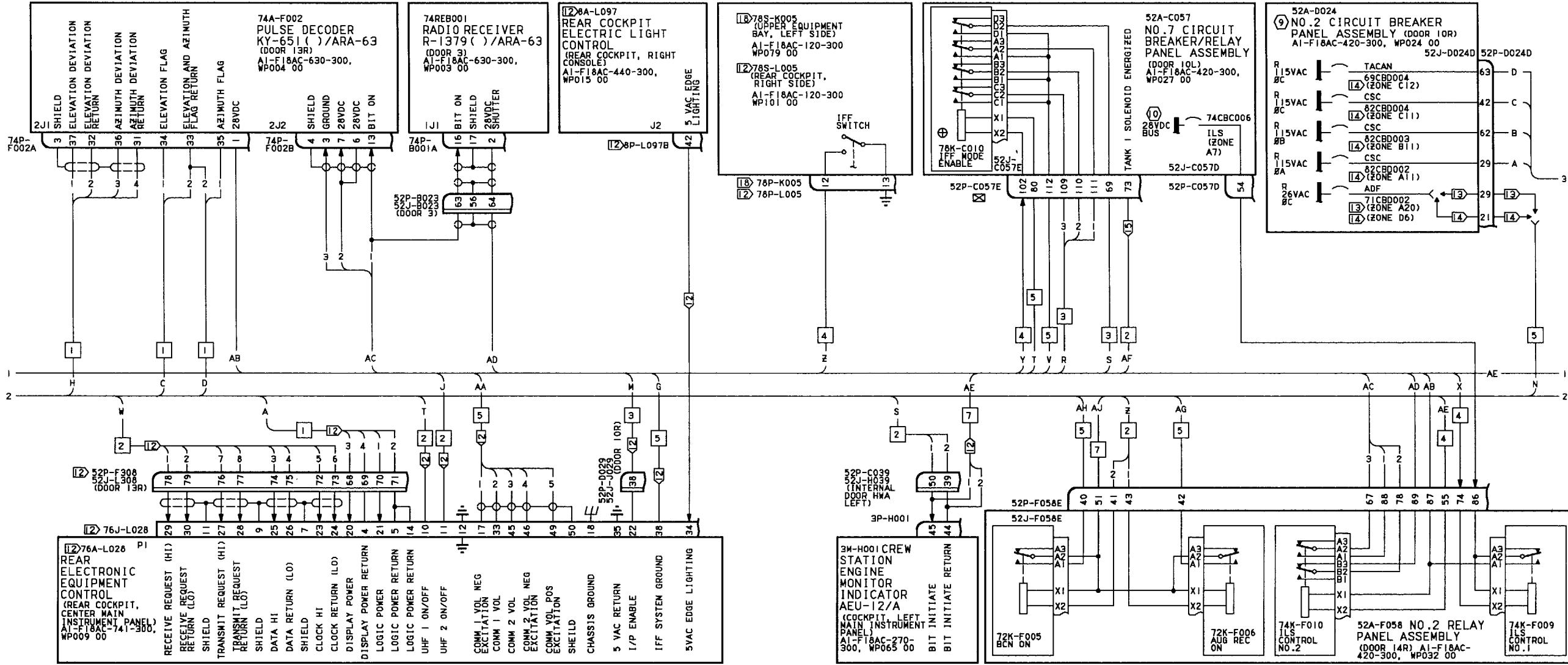


Figure 1.

Figure 1. Control-Converter C-10382/A and Electronic Equipment Control C-10380/ASQ Interconnect Schematic (Sheet 6)

Figure 1.

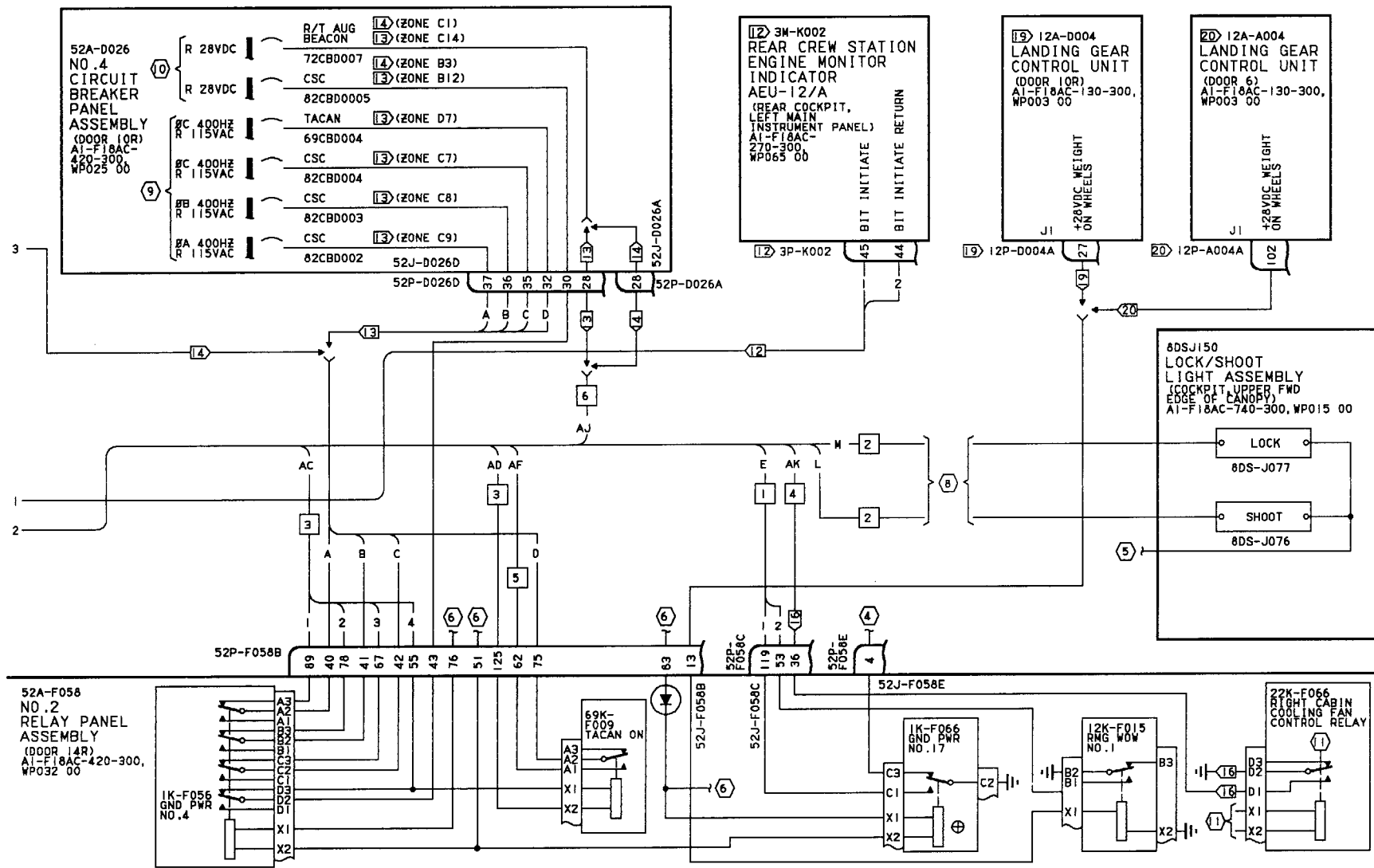


Figure 1.

Figure 1. Control-Converter C-10382/A and Electronic Equipment Control C-10380/ASQ Interconnect Schematic (Sheet 7)

Figure 1.

LEGEND

1.

CONTINUITY TESTS:

A.

ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.

B.

WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY ⊕) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.

C.

DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.

D.

WHEN TESTING CONTINUITY, TEST FOR:

(1)

SHORTS TO GROUND.

(2)

SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.

(3)

SHORTS BETWEEN SHIELD AND CONDUCTORS.

(4)

SHIELD CONTINUITY.

E.

WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY ⊠). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2.

FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.
3.

LINE UNDER LETTER (S) INDICATES LOWER CASE PIN LETTER.
- Ⓐ

AVIONIC COOLING SYSTEM SCHEMATIC, A1-F18AC-410-500, WP009 00.

Ⓟ

COCKPIT WARNING/CAUTION/ADVISORY LIGHTING SYSTEM SCHEMATIC, A1-F18AC-440-500, WP006 00.

Ⓠ

GROUND POWER SWITCHING SCHEMATIC, A1-F18AC-420-500, WP005 00.

Ⓡ

AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 00.

Ⓢ

LOCK/SHOOT LIGHT SCHEMATIC, A1-F18AC-740-500, WP020 00.

Ⓣ

AC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP003 00.

Ⓤ

DC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.

Ⓥ

COCKPIT AVIONICS COOLING SCHEMATIC-161925 AND UP, A1-F18AC-410-500, WP009 00.

12

F/A-18B.

13

161353 THRU 161359.

14

161360 AND UP.

15

161520 AND UP; ALSO 161353 THRU 161519 AFTER F/A-18 AFC 39.

16

162394 AND UP.

17

F/A-18B 162402 AND UP.

18

F/A-18A.

19

161353 THRU 161987 BEFORE F/A-18 AFC 48.

20

162394 AND UP; ALSO 161353 THRU 161987 AFTER F/A-18 AFC 48.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

**SCHEMATIC - CONTROL-CONVERTER AND ELECTRONIC EQUIPMENT CONTROL
INTERCONNECT**

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

Alphabetical Index

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Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jan 01	-

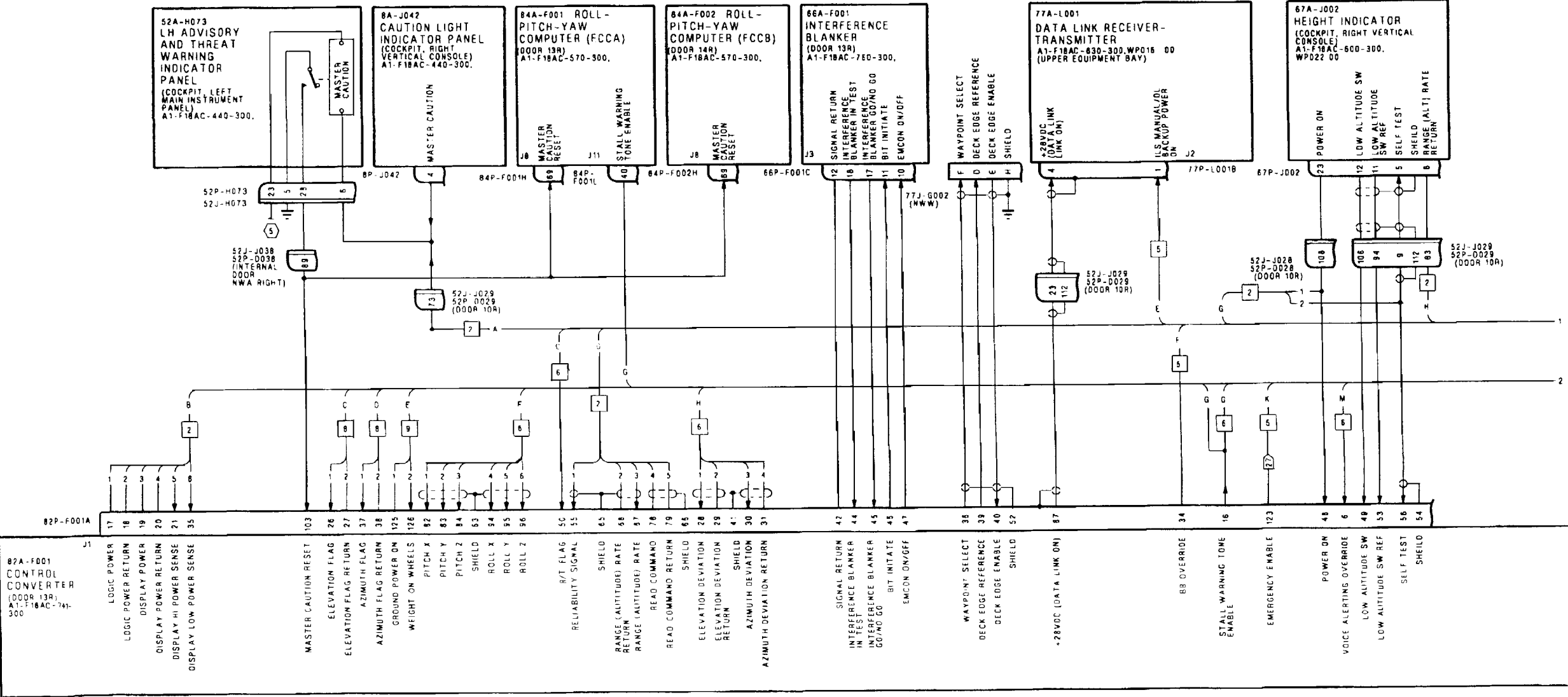


Figure 1.

Figure 1. Control-Converter and Electronic Equipment Control Interconnect Schematic (Sheet 1)

Figure 1.

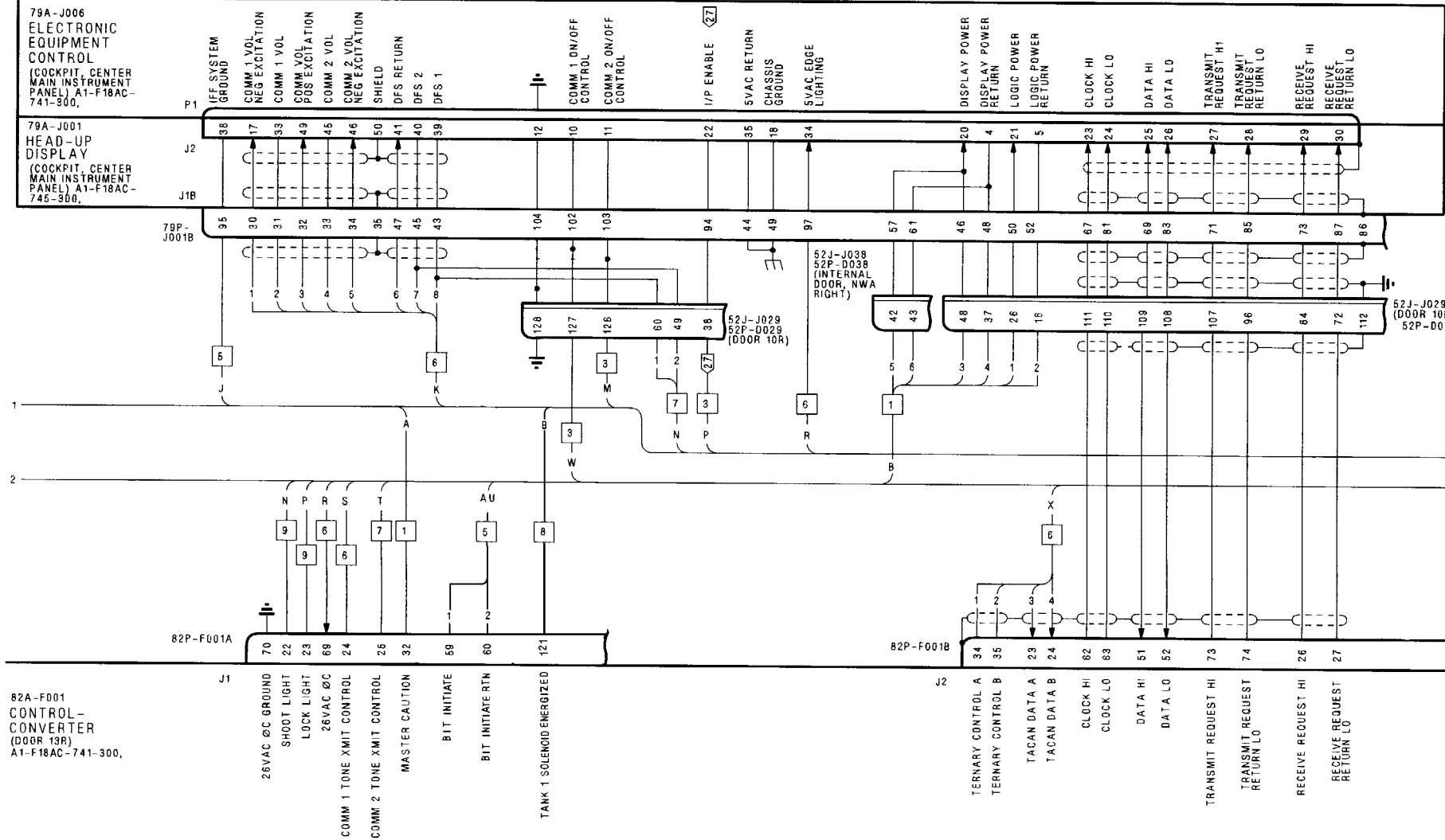


Figure 1.

Figure 1. Control Converter and Electronic Equipment Control Interconnect Schematic (Sheet 2)

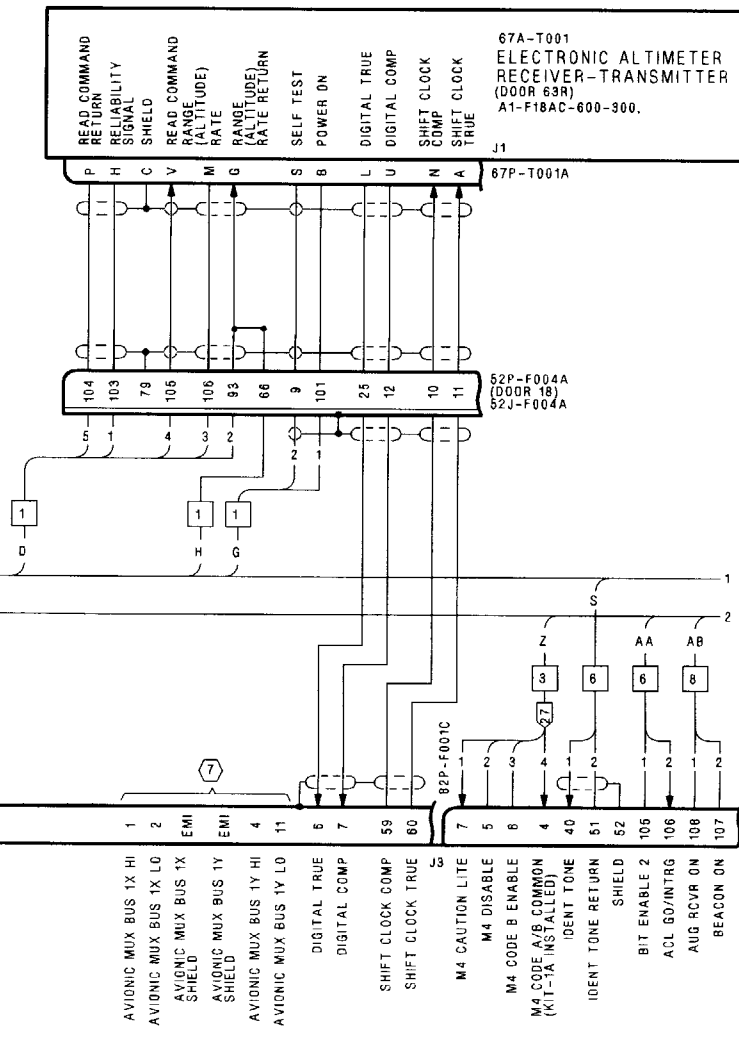


Figure 1.

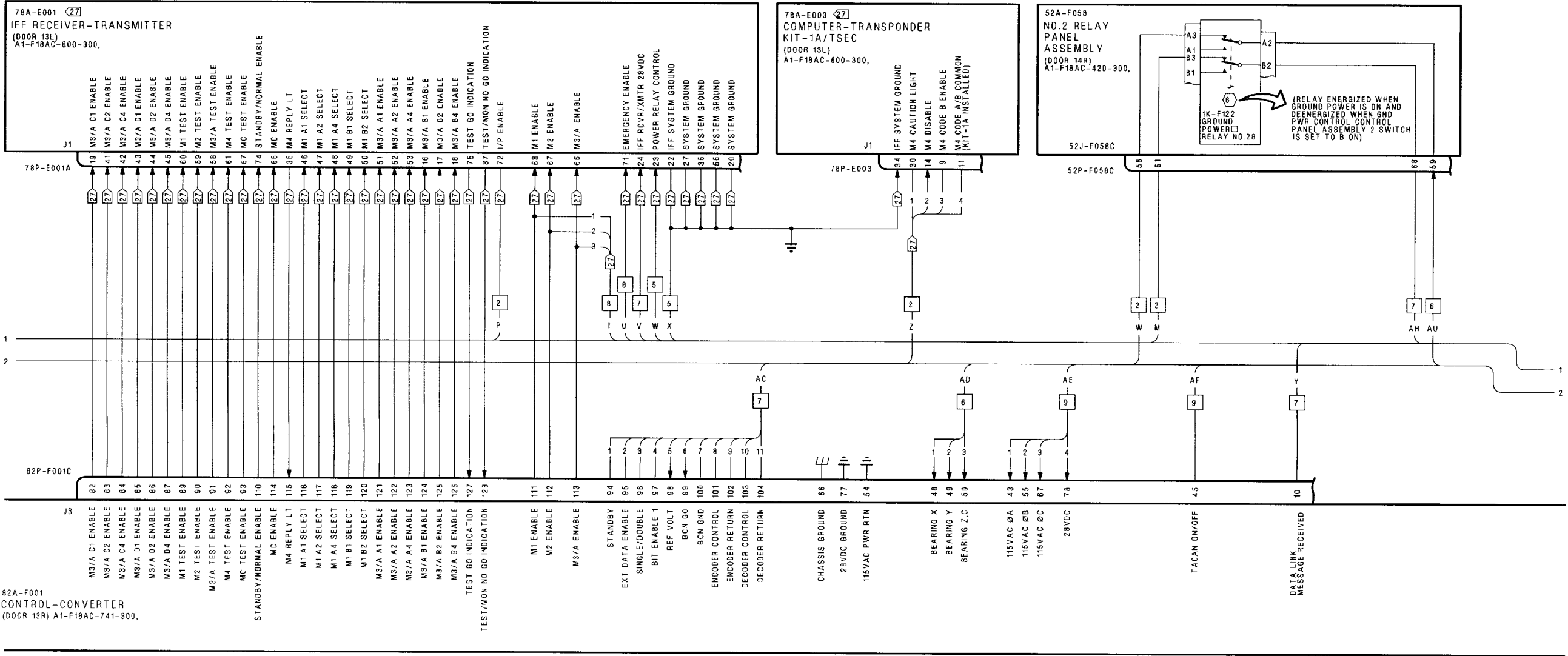


Figure 1.

Figure 1. Control Converter and Electronic Equipment Control Interconnect Schematic (Sheet 3)

Figure 1.

Change 2



Figure 1.

Figure 1. Control Converter and Electronic Equipment Control Interconnect Schematic (Sheet 4)

Figure 1.

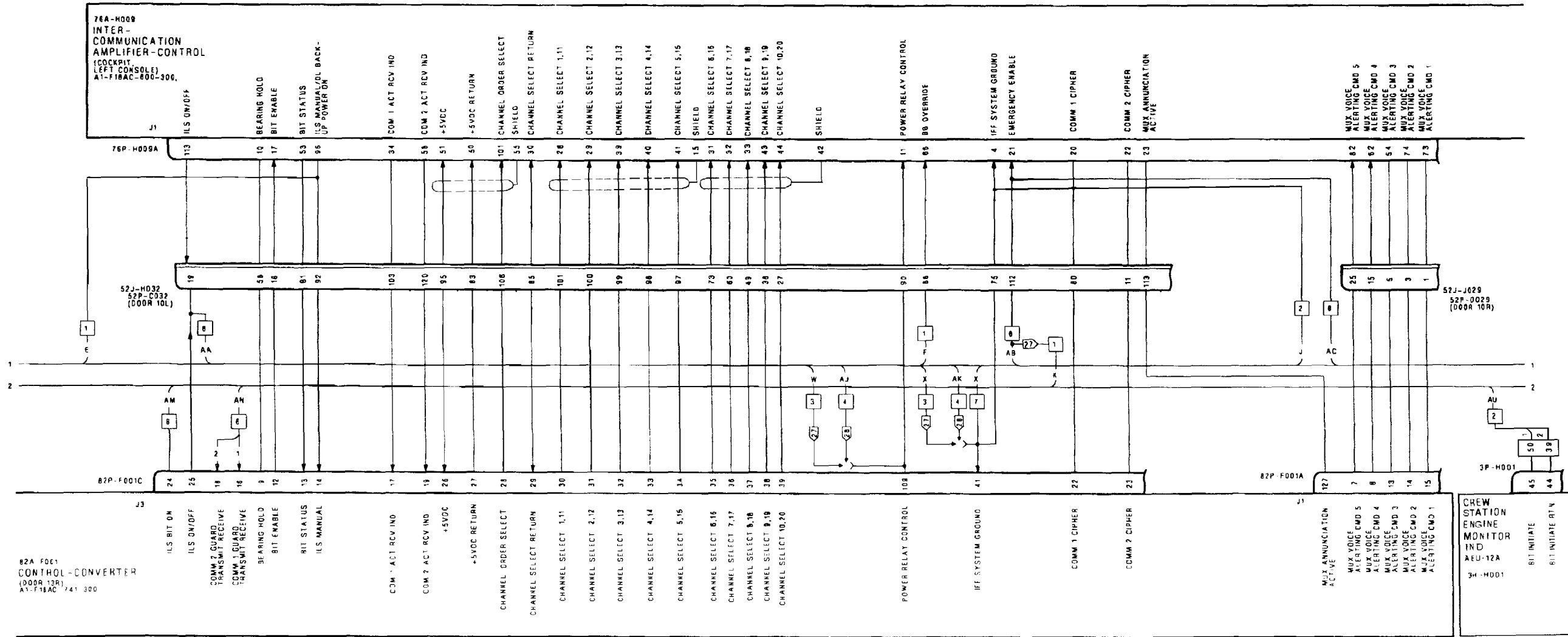


Figure 1.

Figure 1. Control Converter and Electronic Equipment Control Interconnect Schematic (Sheet 5)

Figure 1.

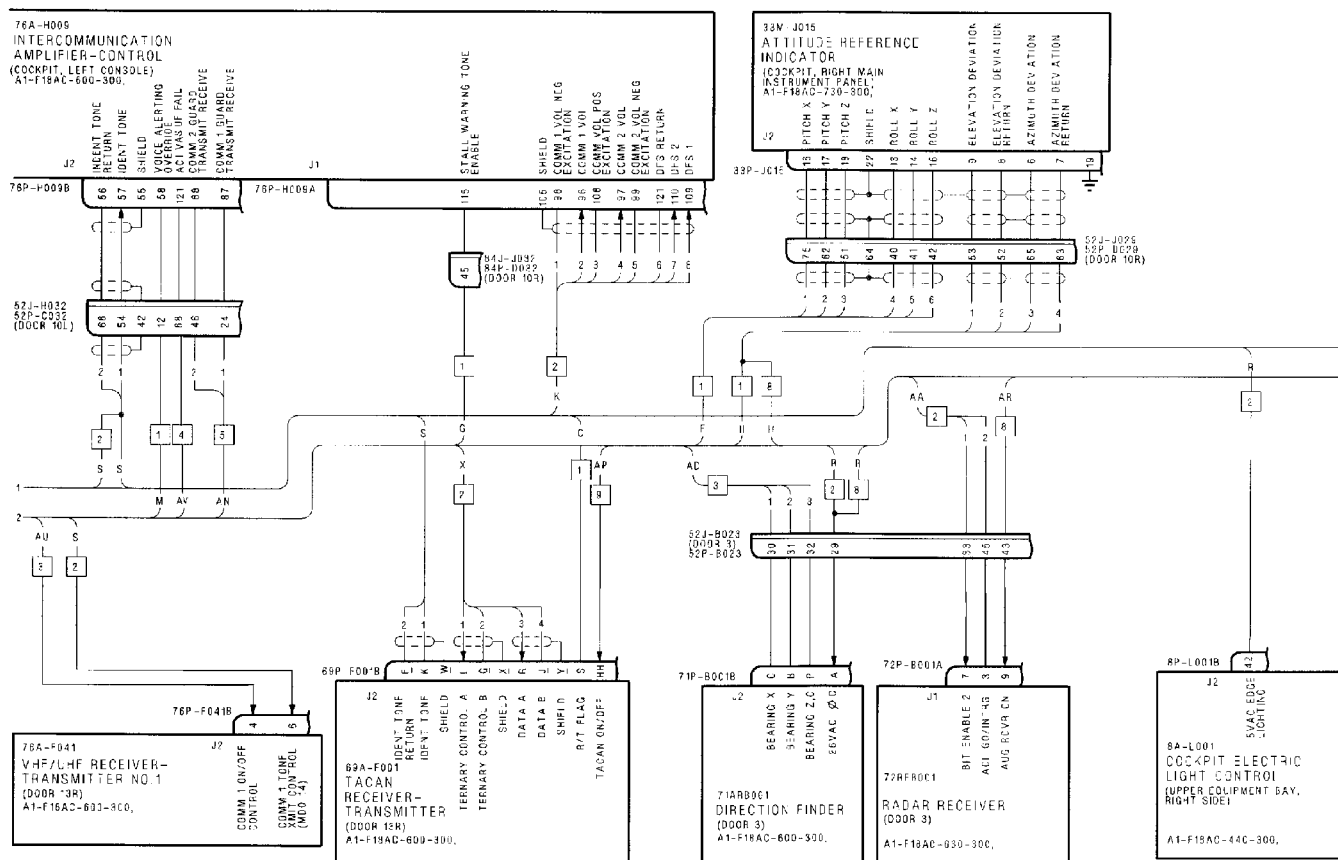


Figure 1.

Figure 1. Control-Converter and Electronic Equipment Control Interconnect Schematic (Sheet 6)

Figure 1.

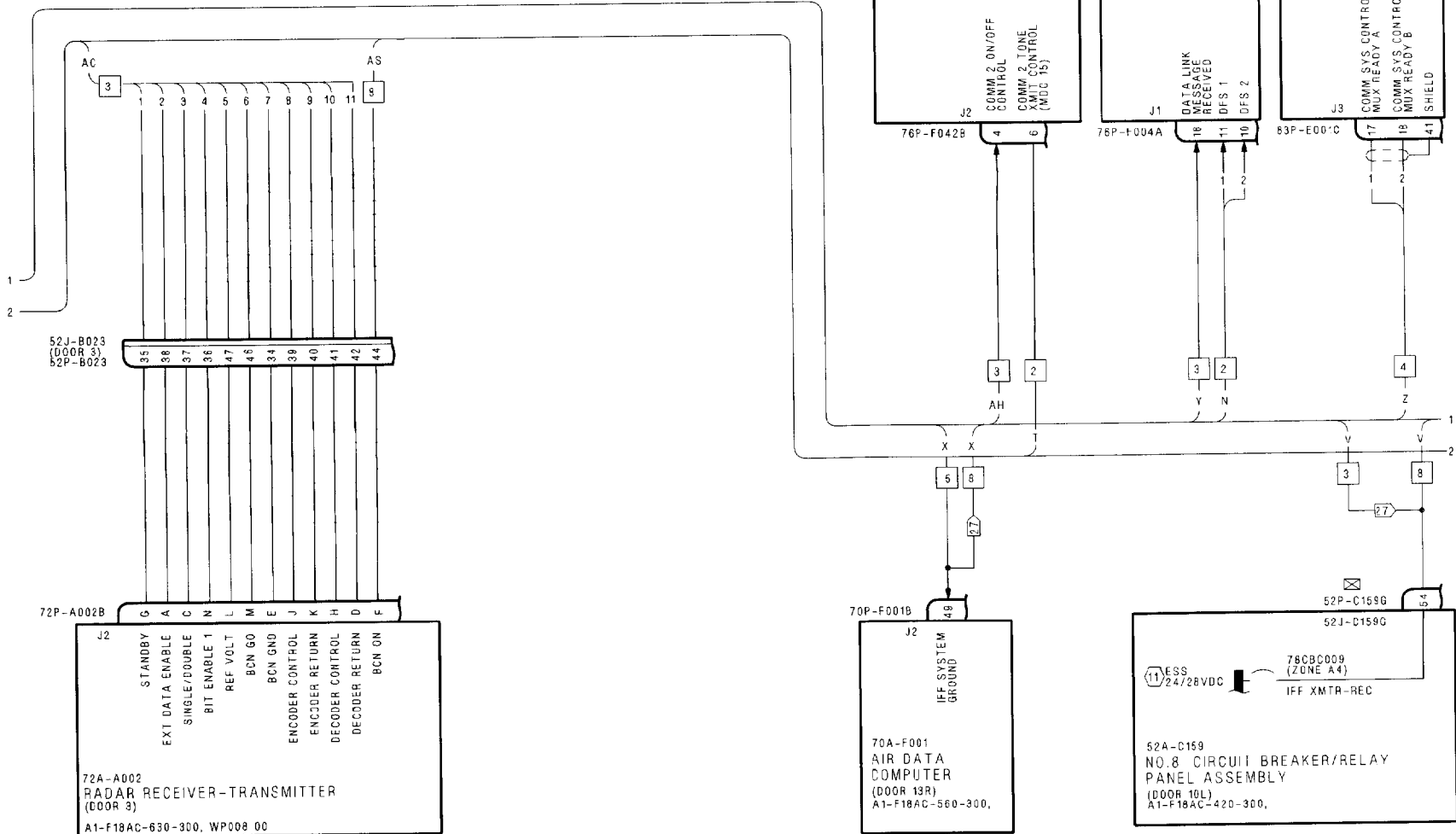


Figure 1. Control-Converter and Electronic Equipment Control Interconnect Schematic (Sheet 7)

Figure 1.

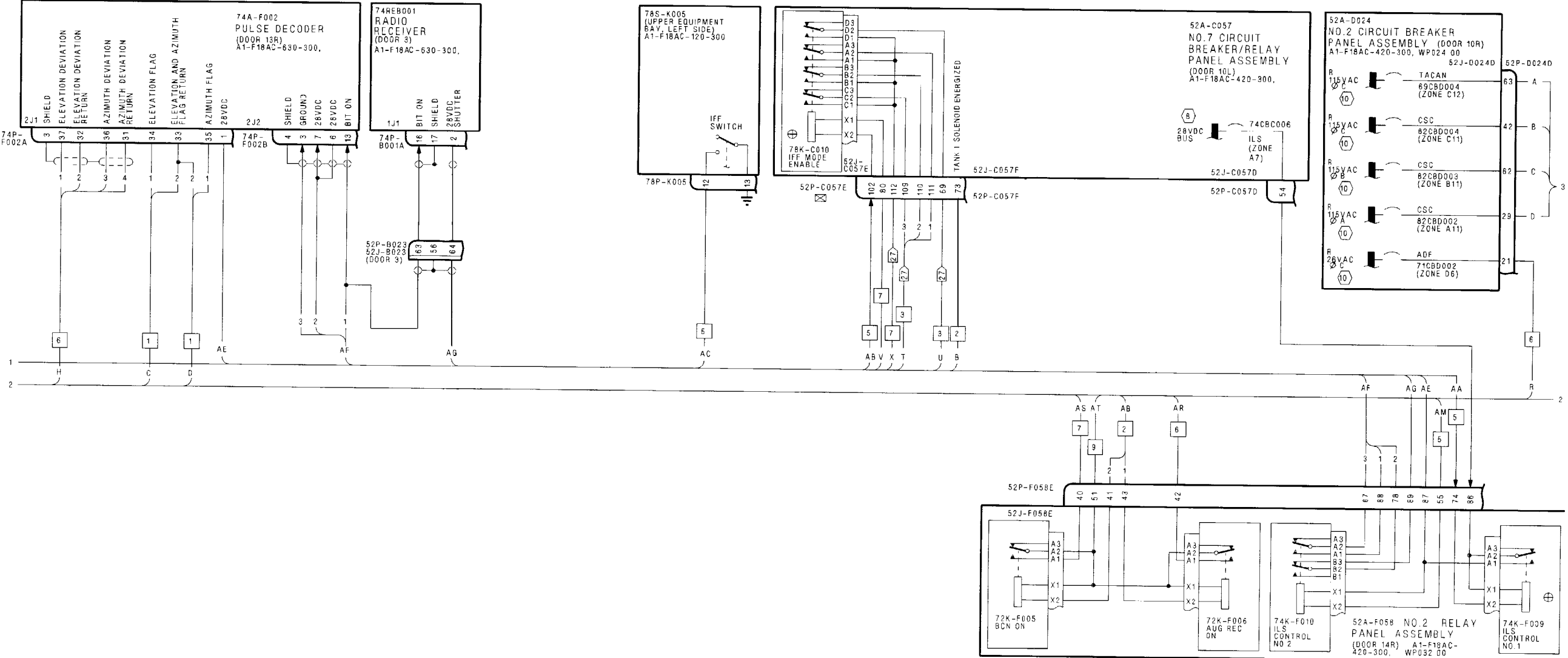
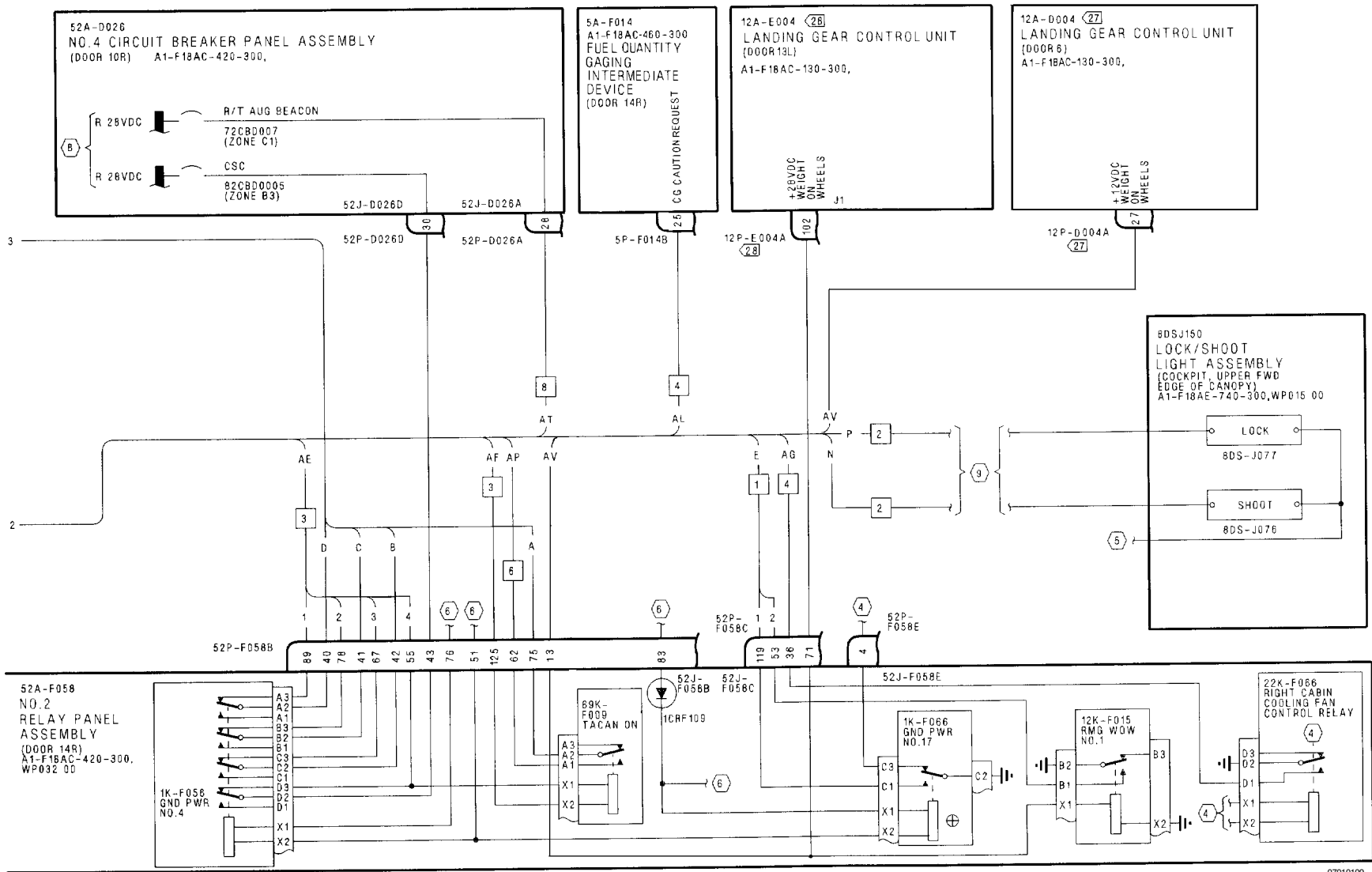


Figure 1.

Figure 1. Control-Converter and Electronic Equipment Control Interconnect Schematic (Sheet 8)

Figure 1.



07010109

Figure 1.

Figure 1. Control-Converter and Electronic Equipment Control Interconnect Schematic (Sheet 9)

Figure 1.

LEGEND

1. CONTINUITY TESTS:

A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.

B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY ⊕) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.

C. WHEN TESTING CONTINUITY, TEST FOR:

(1) SHORTS TO GROUND.

(2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.

(3) SHORTS BETWEEN SHIELD AND CONDUCTORS.

(4) SHIELD CONTINUITY.

D. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY ☒). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

3. LINE UNDER LETTER (S) INDICATES LOWER CASE LETTERS.

④

AVIONIC COOLING SYSTEM SCHEMATIC, A1-F18AC-410-500, WP009 00.

⑤

COCKPIT WARNING/CAUTION/ADVISORY LIGHTING SYSTEM SCHEMATIC, A1-F18AC-440-500, WP006 00.

⑥

GROUND POWER SWITCHING SCHEMATIC, A1-F18AC-420-500, WP005 00.

⑦

AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 01.

⑧

DC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.

⑨

LOCK/SHOOT LIGHT/SHOOT CUE DISPLAY SCHEMATIC, A1-F18AC-740-500, WP038 00.

⑩

AC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP003 00.

⑪

DELETED.

⑫

POWER CONTROL SCHEMATIC, A1-F18AC-580-500, WP005 00.

13

DELETED.

14

DELETED.

15

DELETED.

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DELETED.

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DELETED.

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DELETED.

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DELETED.

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DELETED.

26

DELETED.

27

AFTER F/A-18 AFC 253.

28

AFTER F/A-18 AFC 292.

29

DELETED.

30

DELETED.

Figure 1.

Figure 1. Control-Converter and Electronic Equipment Control Interconnect Schematic (Sheet 10)

Figure 1.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - DIGITAL DATA COMPUTER NO. 1 AND NO. 2 INTERCONNECT****MISSION COMPUTER SYSTEM****This WP supersedes WP 008 00, dated 1 January 2001.**

Title	Work Package
Schematic - Digital Data Computer No. 1 and No. 2 Interconnect (F/A-18A/B)	008 01
Schematic - Digital Data Computer No. 1 and No. 2 Interconnect (AFTER F/A-18 AFC 225 AND F/A-18 AFC 231)	008 02
Schematic - Digital Data Computer No. 1 and No. 2 Interconnect (AFTER F/A-18 AFC 253 OR F/A-18 AFC 292; AND AFTER AFC 231 PART 2 OR PART 3)	008 03

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - DIGITAL DATA COMPUTER NO. 1 AND NO. 2 INTERCONNECT****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A/B**

This WP supersedes WP 008 01, dated 1 January 2001.

Reference Material

None

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Digital Data Computer No. 1 and No. 2 Interconnect Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 39	-	No. 1 Fuel Tank Interconnect Valve Replacement and Fuel Sequencing Modification (ECP MDA-F18-00072C1)	1 Apr 87	-
F/A-18 AFC 41	-	Throttle Sensitivity Improvement (ECP MDA-F18-00054)	1 Apr 8	-
F/A-18 AFC 90	-	GFE Battery Relay Control Unit Incorporation. (ECP MDA-F18-00165R1)	1 Jun 92	-

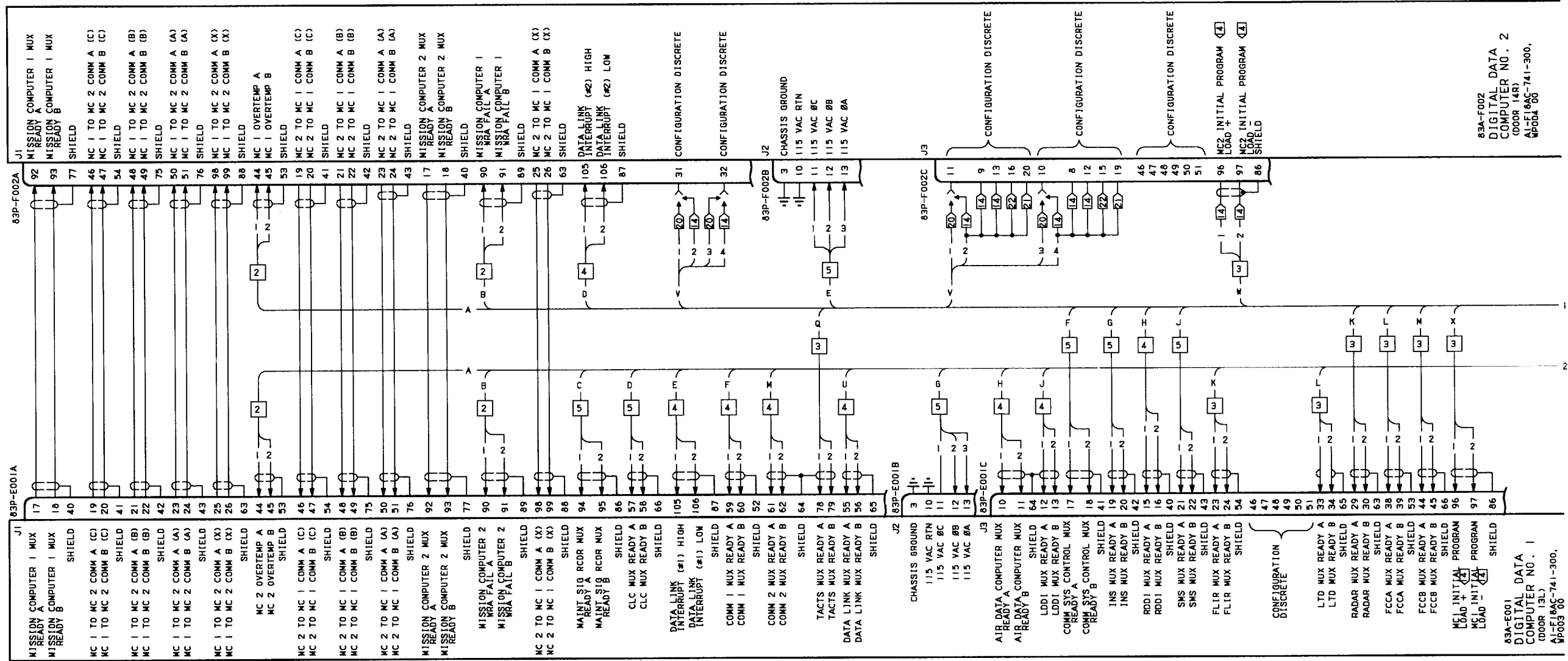


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 1)

Figure 1.

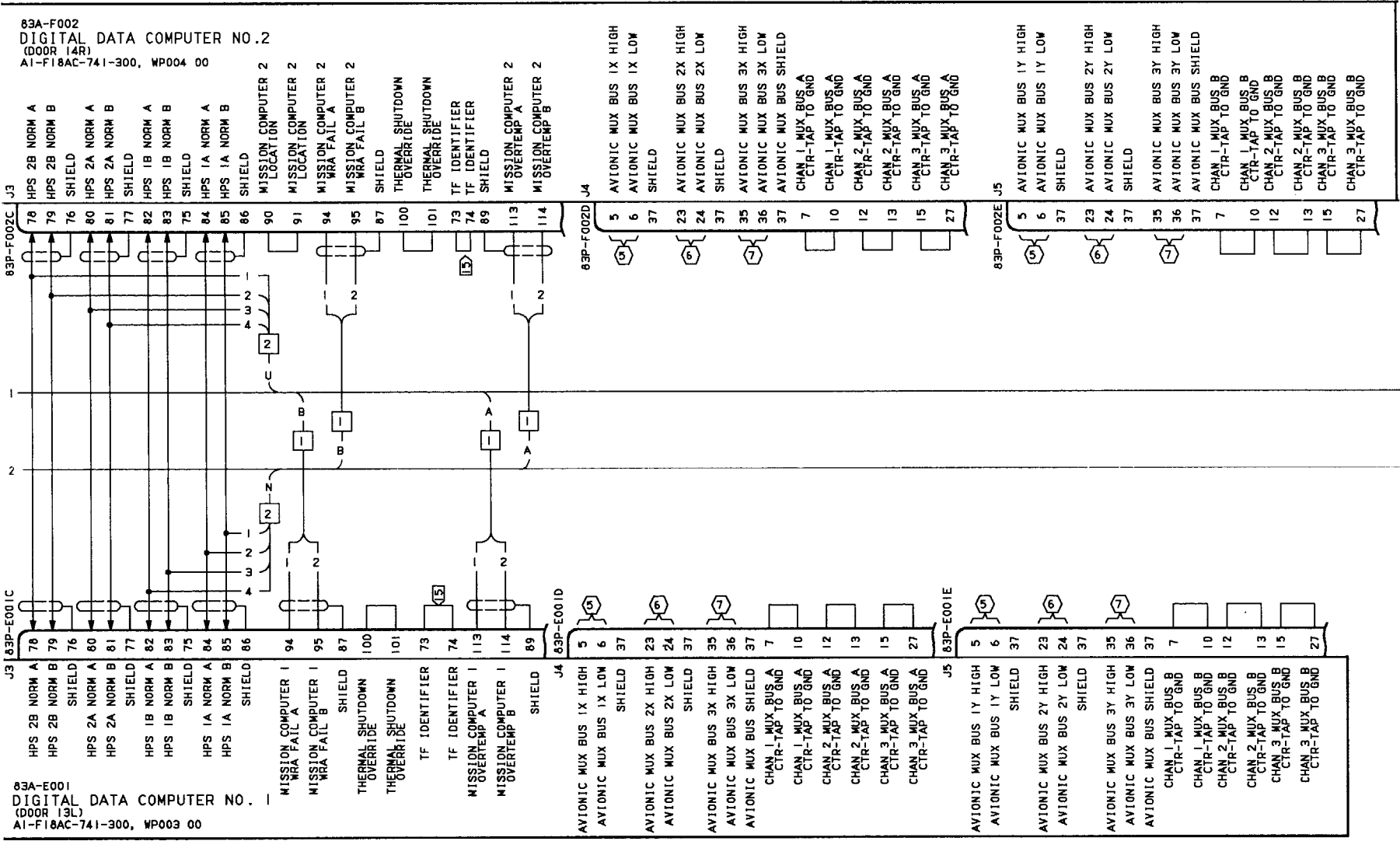


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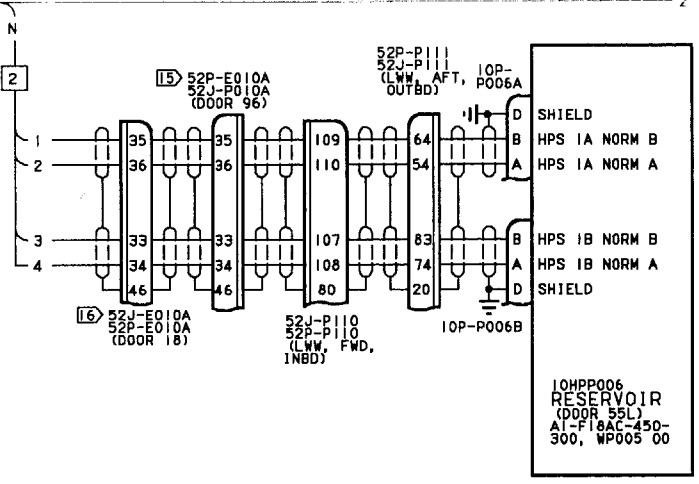
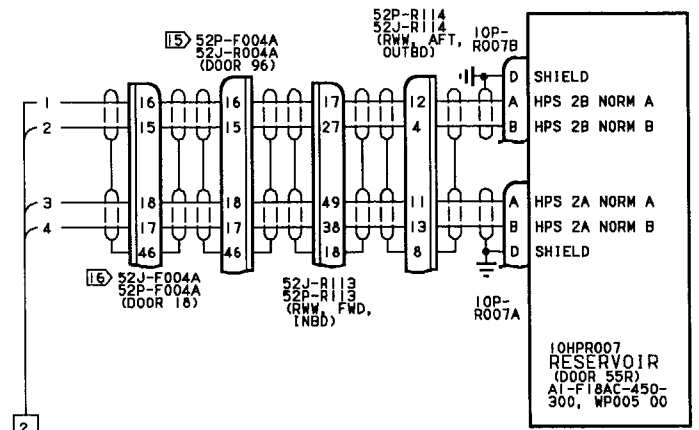


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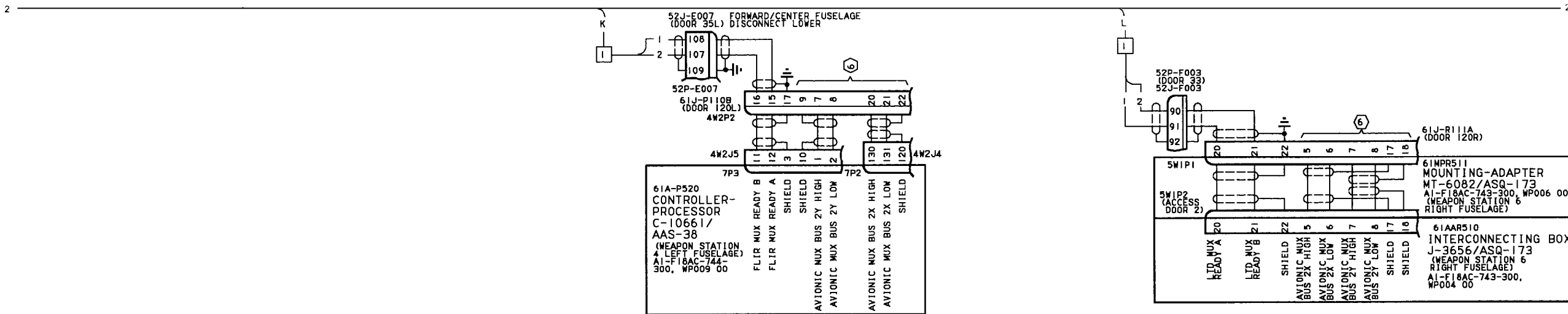
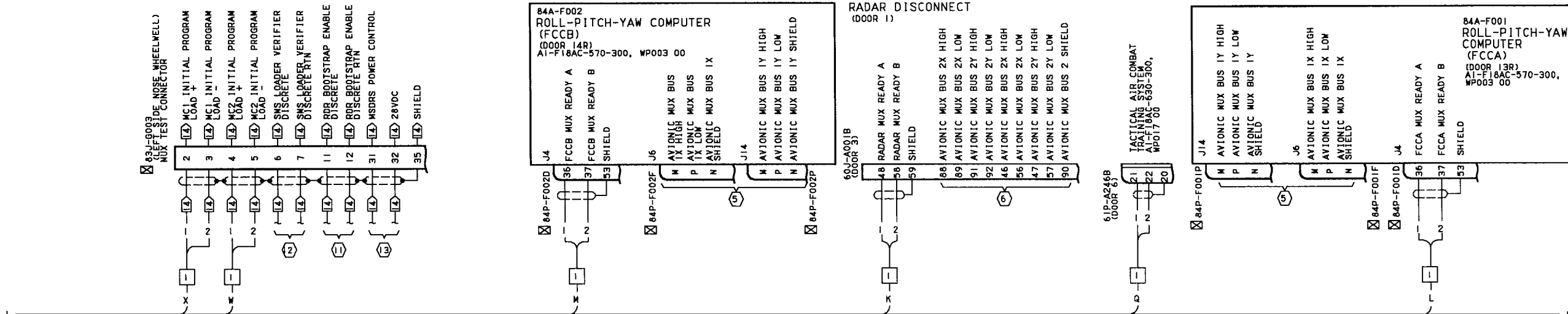


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 3)

Figure 1.

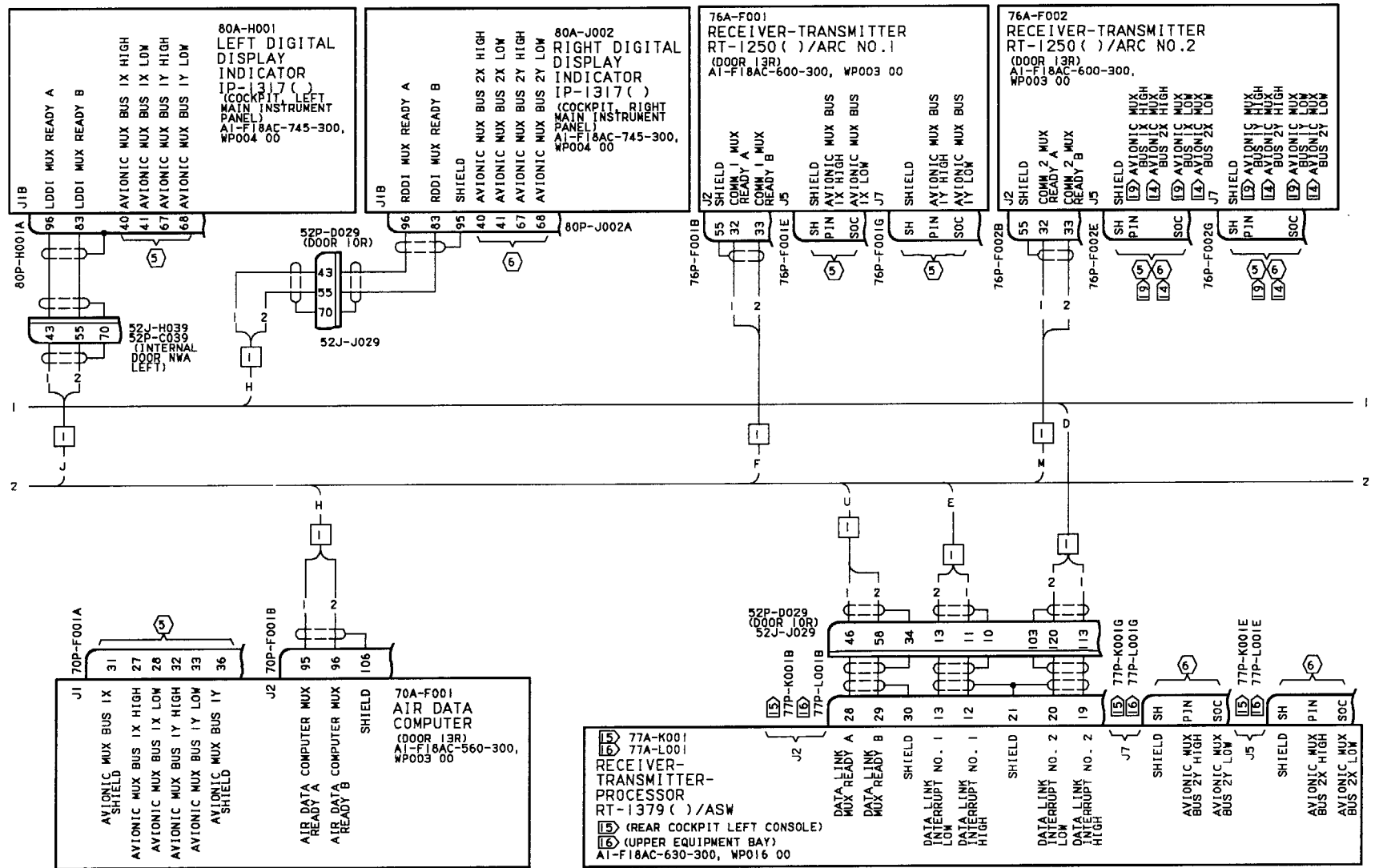


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 4)

Figure 1.

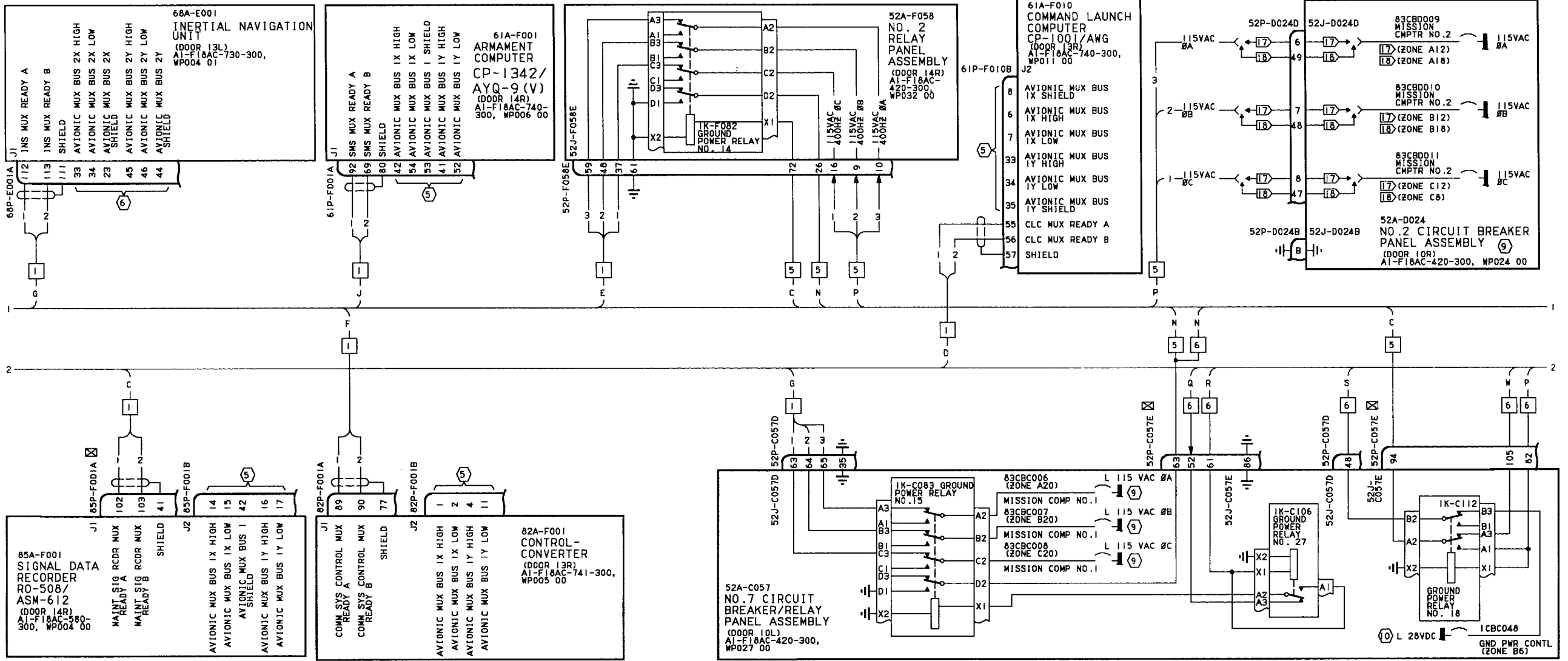


Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 5)

Figure 1.

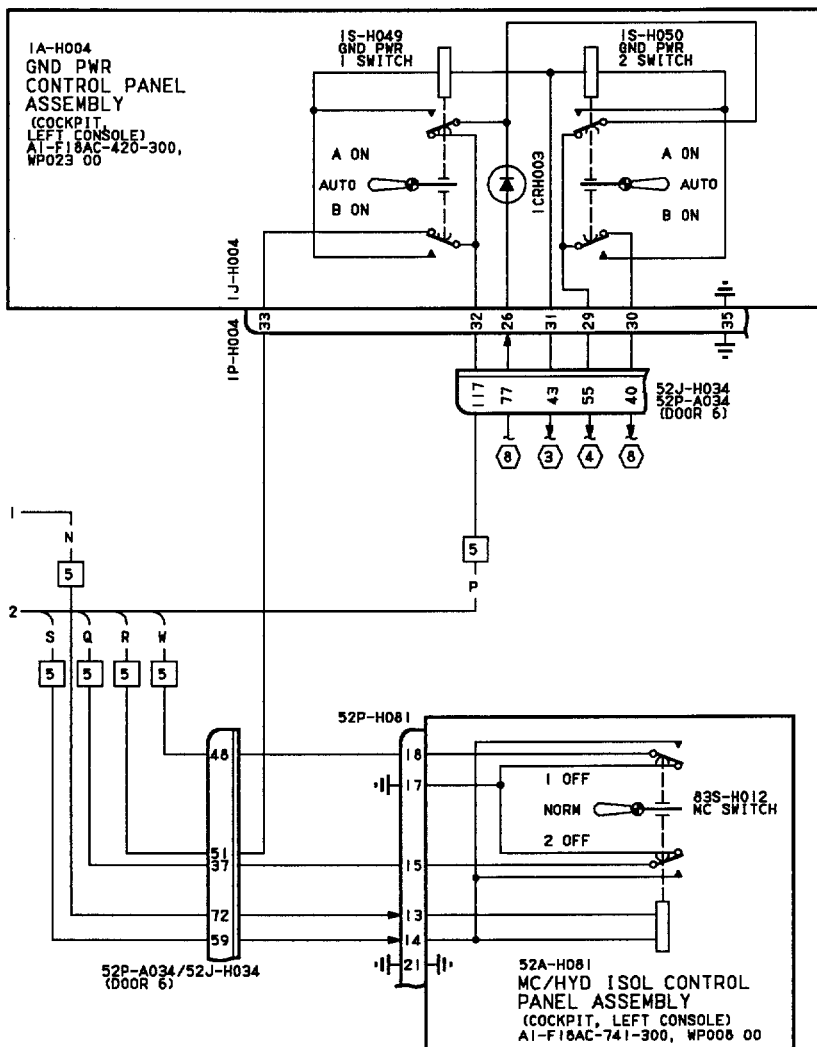


Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 6)

LEGEND























1. CONTINUITY TESTS:
- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY ) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
- (1) SHORTS TO GROUND.
- (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
- (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
- (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY ). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.
-  AVIONIC COOLING SYSTEM SCHEMATIC, A1-F18AC-410-500, WP009 00.
-  AC/DC POWER DISTRIBUTION SCHEMATIC, WP009 00.
-  AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 00.
-  AVIONIC MUX CHANNEL 2 SCHEMATIC, WP005 00.
-  AVIONIC MUX CHANNEL 3 SCHEMATIC, WP006 00.
-  GROUND POWER SWITCHING SCHEMATIC, A1-F18AC-420-500, WP005 00.
-  AC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP003 00.
-  DC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.
-  RADAR SYSTEM INTERCONNECT, A1-F18AC-742-500, WP005 00.
-  BUILT-IN TEST SCHEMATIC, A1-F18AC-740-500, WP021 00.
-  POWER SCHEMATIC, A1-F18AC-580-500, WP005 00.
-  161702 AND UP.
-  F/A-18B.
-  F/A-18A.
-  161353 THRU 161359
-  161360 AND UP.
-  181353 THRU 161528.
-  161520 THRU 161528; ALSO 161353 THRU 161519 AFTER F/A-18 AFC 39.
-  163119 AND UP; ALSO 161353 THRU 163118 AFTER F/A-18 AFC 90.
-  161702 AND UP; ALSO 161353 THRU 161528 AFTER F/A-18 AFC 41.

Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 7)

Figure 1.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - DIGITAL DATA COMPUTER NO. 1 AND NO. 2 INTERCONNECT****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AND F/A-18B AFTER F/A-18 AFC 225 AND F/A-18 AFC 231**

Reference Material

None

Alphabetical Index**Subject****Page No.**

Digital Data Computer No. 1 and No. 2 Interconnect Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 39	-	No. 1 Fuel Tank Interconnect Valve Replacement and Fuel Sequencing Modification (ECP MDA-F18-00072C1)	1 Apr 87	-
F/A-18 AFC 41	-	Throttle Sensitivity Improvement (ECP MDA-F18-00054)	1 Apr 8	-
F/A-18 AFC 90	-	GFE Battery Relay Control Unit Incorporation. (ECP MDA-F18-00165R1)	1 Jun 92	-
F/A-18 AFC 225	-	Five (5) Avionics Multiplex Bus Upgrade, Incor- poration of (ECP MDA-F/A-18 0529)	1 Jun 02	-
F/A-18 AFC 231	-	Embedded Global Positioning System (GPS)/In- ertial Navigation System (INS) (EGI), Incorpora- tion of (ECP MDA-F/A-18 0521)	1 Jun 02	-

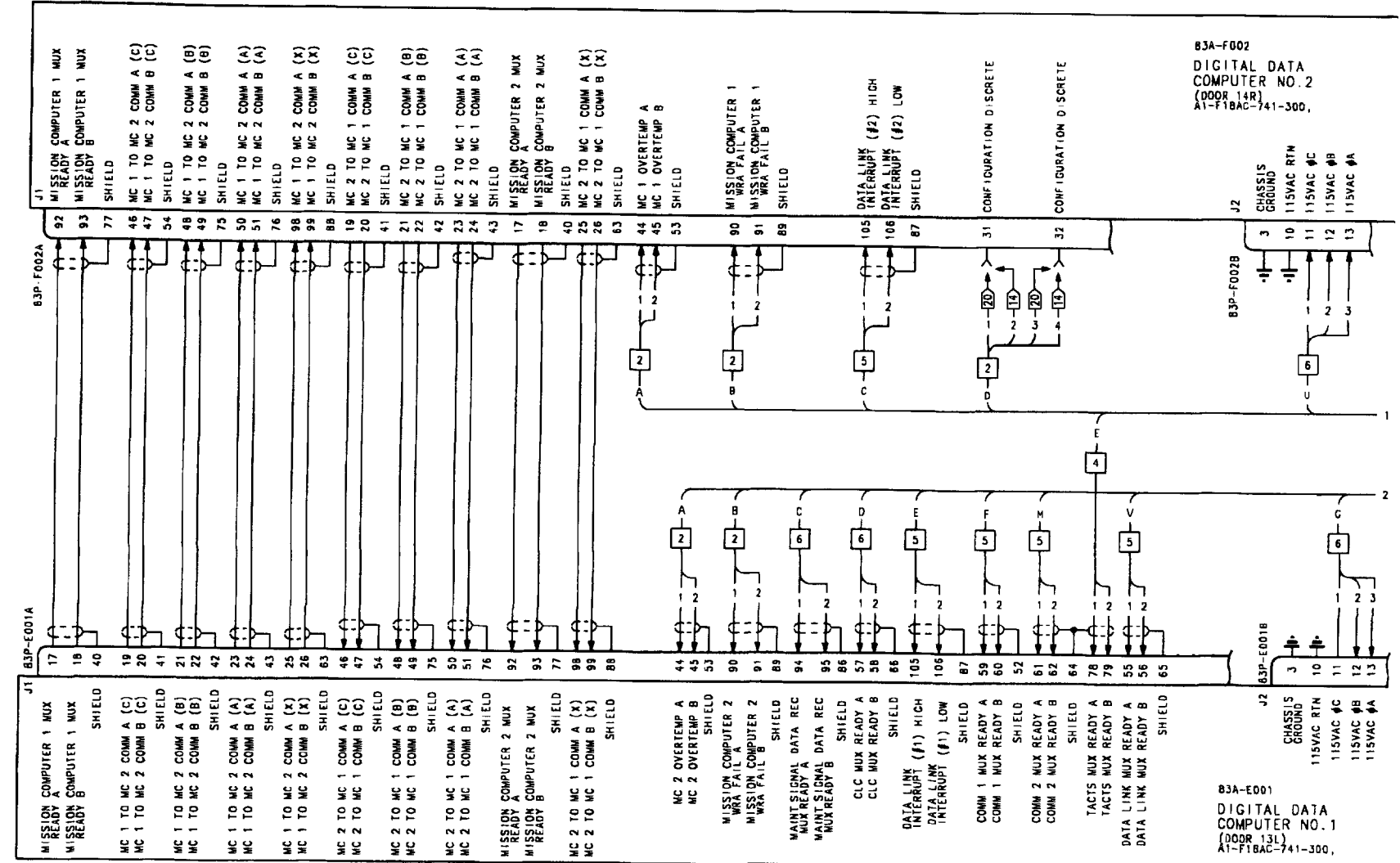


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 1)

Figure 1.



Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 2)

Figure 1.

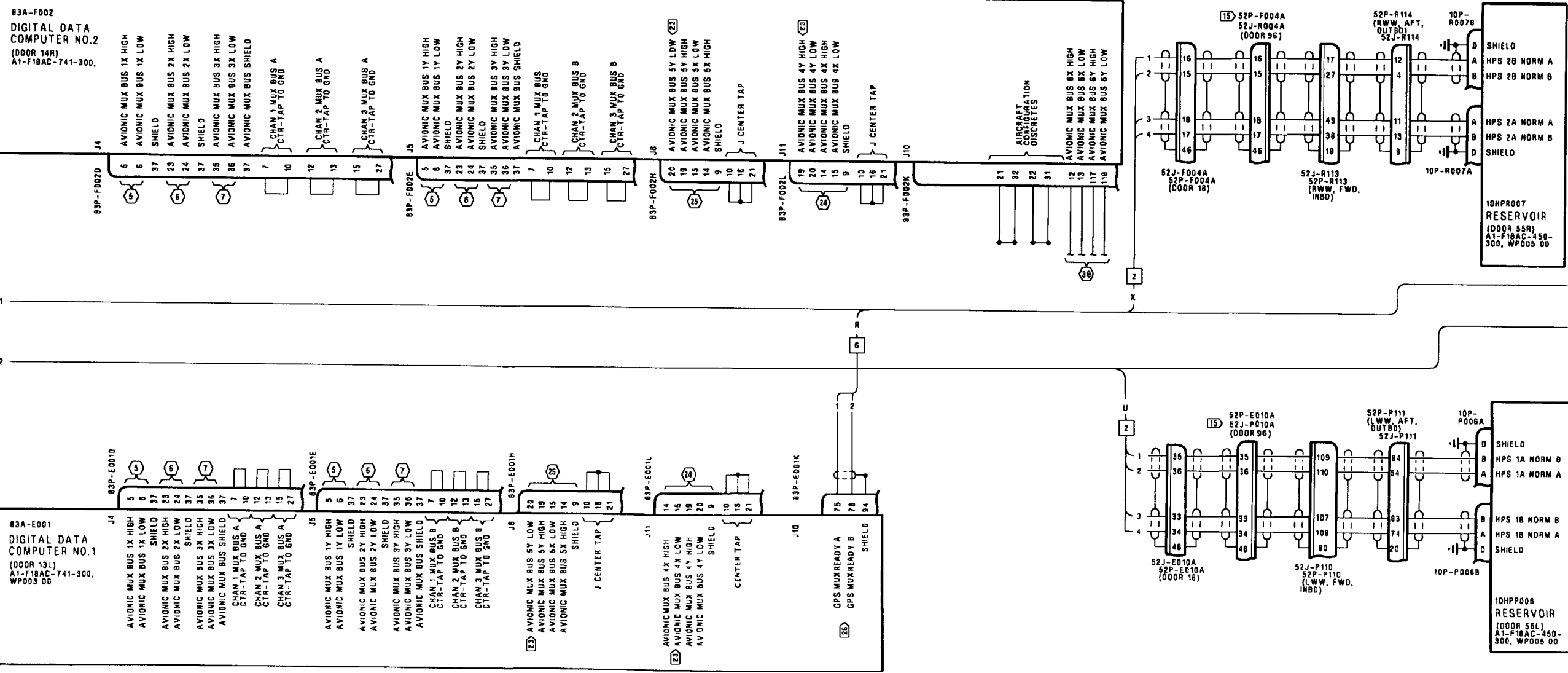


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 3)

Figure 1.



Figure 1.

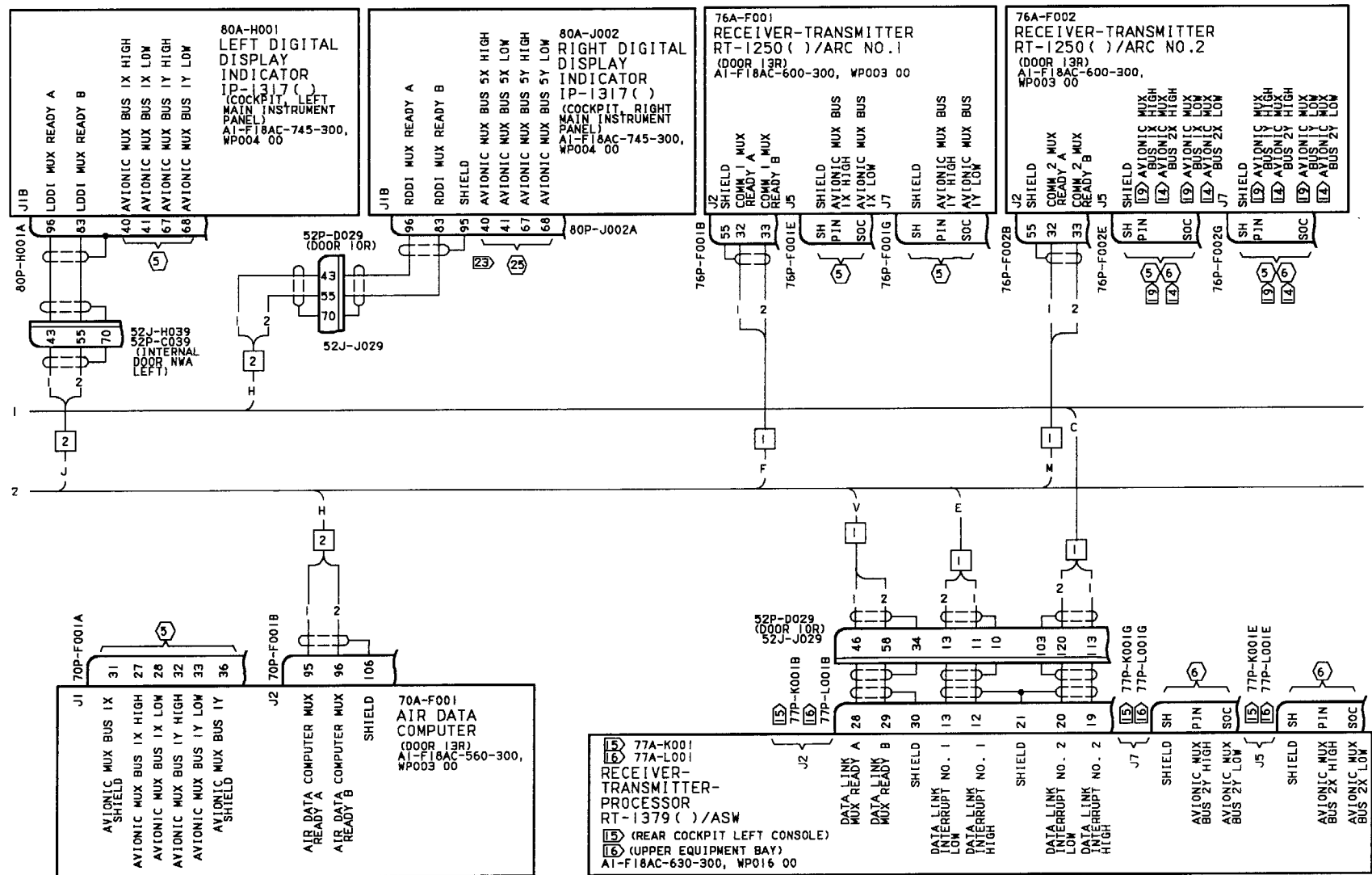


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 5)

Figure 1.

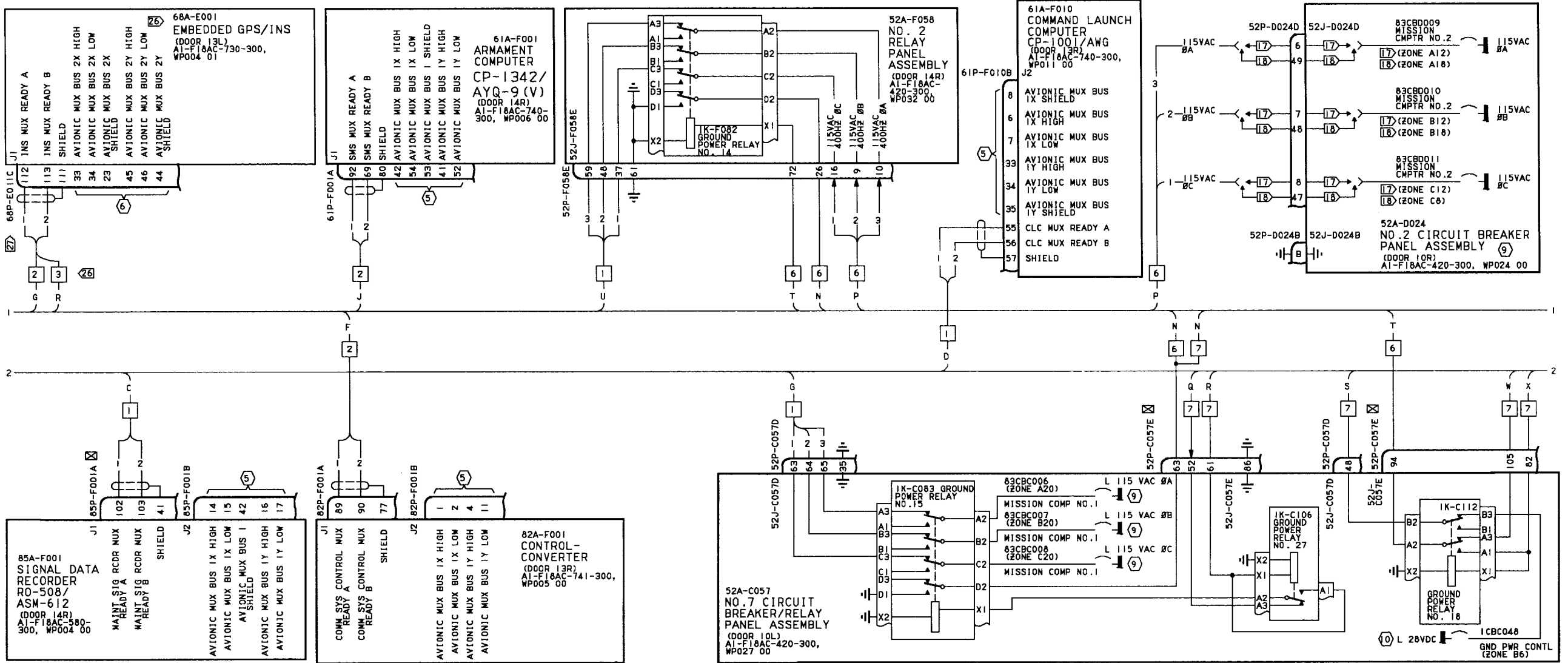


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 6)

Figure 1.

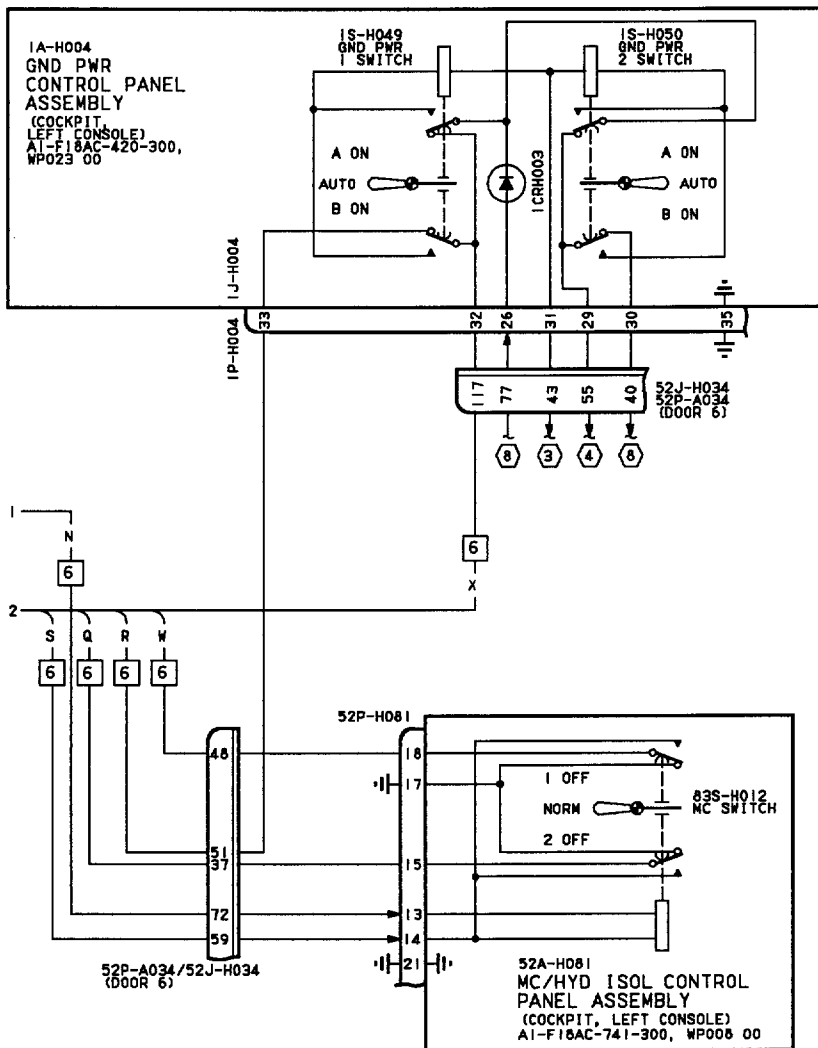


Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 7)

LEGEND

1. CONTINUITY TESTS:
- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY Ⓢ) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
- (1) SHORTS TO GROUND.
- (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
- (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
- (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY Ⓜ). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.
- ③ AVIONIC COOLING SYSTEM SCHEMATIC, A1-F18AC-410-500, WP009 00.
- ④ AC/DC POWER DISTRIBUTION SCHEMATIC, WP009 00.
- ⑤ AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 00.
- ⑥ AVIONIC MUX CHANNEL 2 SCHEMATIC, WP005 00.
- ⑦ AVIONIC MUX CHANNEL 3 SCHEMATIC, WP006 00.
- ⑧ GROUND POWER SWITCHING SCHEMATIC, A1-F18AC-420-500, WP005 00.
- ⑨ AC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP003 00.
- ⑩ DC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.
- ⑪ RADAR SYSTEM INTERCONNECT, A1-F18AC-742-500, WP005 00.
- ⑫ BUILT-IN TEST SCHEMATIC, A1-F18AC-740-500, WP021 00.
- ⑬ POWER SCHEMATIC, A1-F18AC-580-500, WP005 00.
- ⑭ 161702 AND UP.
- ⑮ F/A-18B.
- ⑯ F/A-18A.
- ⑰ 161353 THRU 161359
- ⑱ 161360 AND UP.
- ⑲ 181353 THRU 161528.
- ⑳ 161520 THRU 161528; ALSO 161353 THRU 161519 AFTER F/A-18 AFC 39.
- ㉑ 163119 AND UP; ALSO 161353 THRU 163118 AFTER F/A-18 AFC 90.
- ㉒ 161702 AND UP; ALSO 161353 THRU 161528 AFTER F/A-18 AFC 41.
- ㉓ F/A-18A AND F/A-18B AFTER AFC 225.
- ㉔ AVIONIC MUX CHANNEL 4 SCHEMATIC, WP017 00.
- ㉕ AVIONIC MUX CHANNEL 5 SCHEMATIC, WP018 00.
- ㉖ F/A-18 AFTER AFC 231.

Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 8)

Figure 1.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - DIGITAL DATA COMPUTER NO. 1 AND NO. 2 INTERCONNECT

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292;
AND AFTER F/A-18 AFC 231 PART 2 OR PART 3

Reference Material

None

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Subject

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Digital Data Computer No. 1 and No. 2 Interconnect Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jan 01	-
F/A-18 AFC 253	-	U.S. Naval Reserves A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jun 02	-
F/A-18 AFC 292	-	U.S. Marine Corps A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jun 02	-
F/A-18 AFC 231 Part 2 or Part 3	-	Embedded Global Positioning System (GPS)/Inertial Navigation System (INS) (EGI), Incorporation of (ECP MDA-FA-18 0521)	1 Jun 02	-



Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 1)

Figure 1.

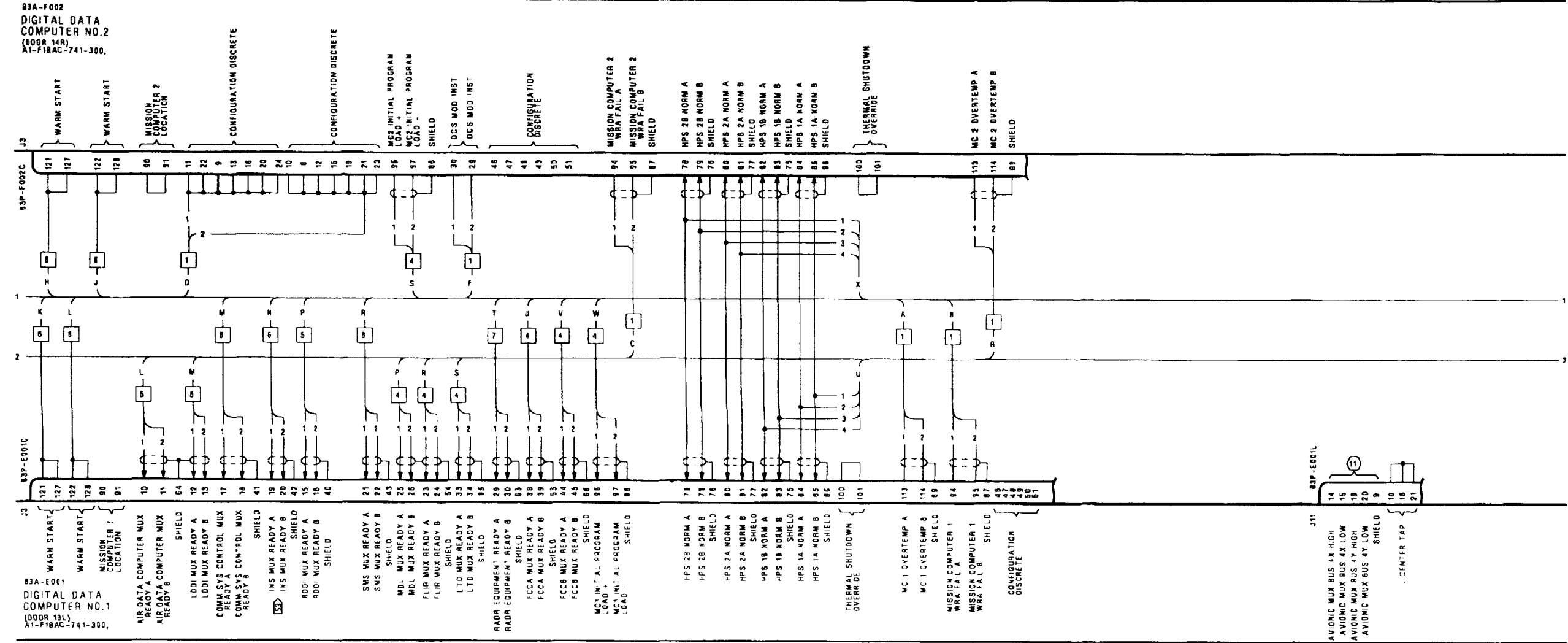


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 2)

Figure 1.



Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 3)

Figure 1.



Figure 1.

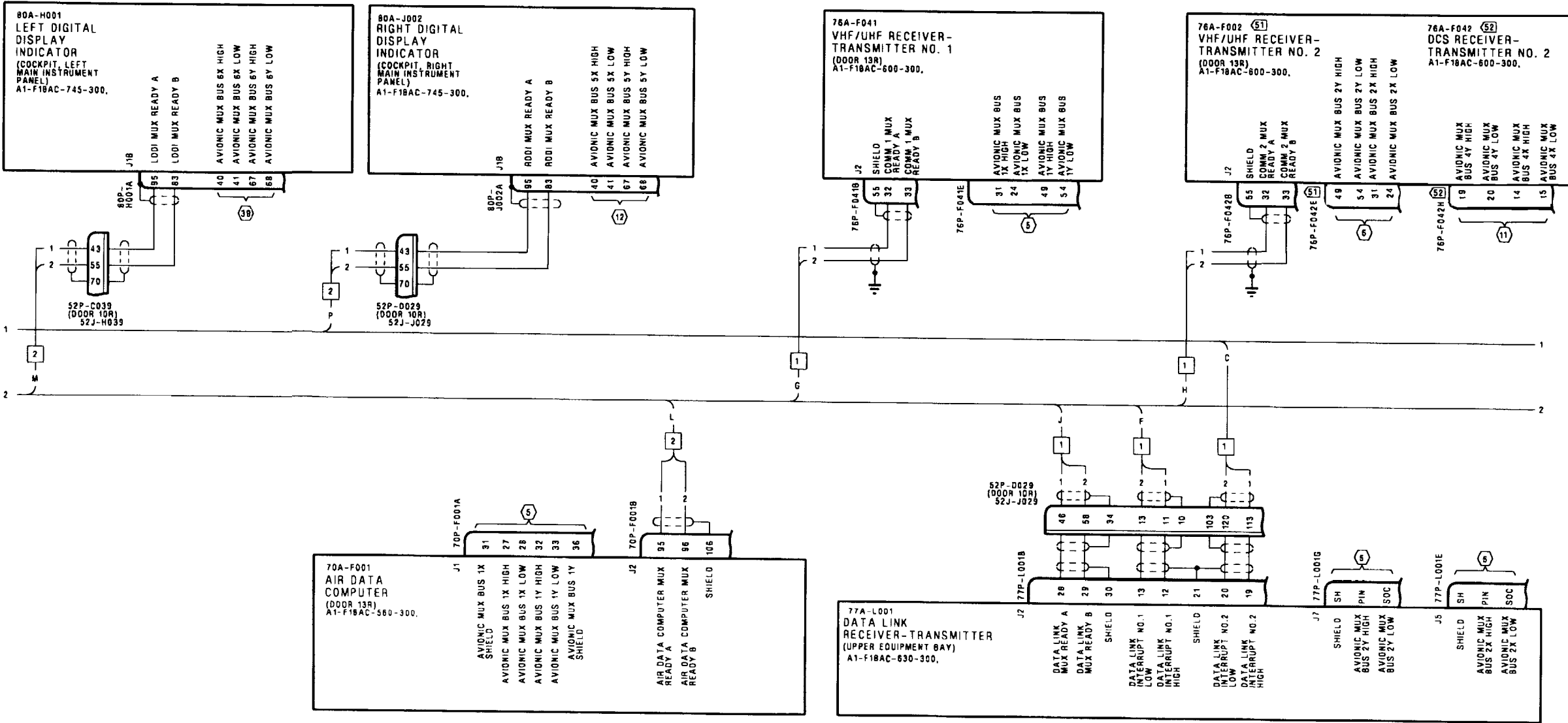


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 5)

Figure 1.

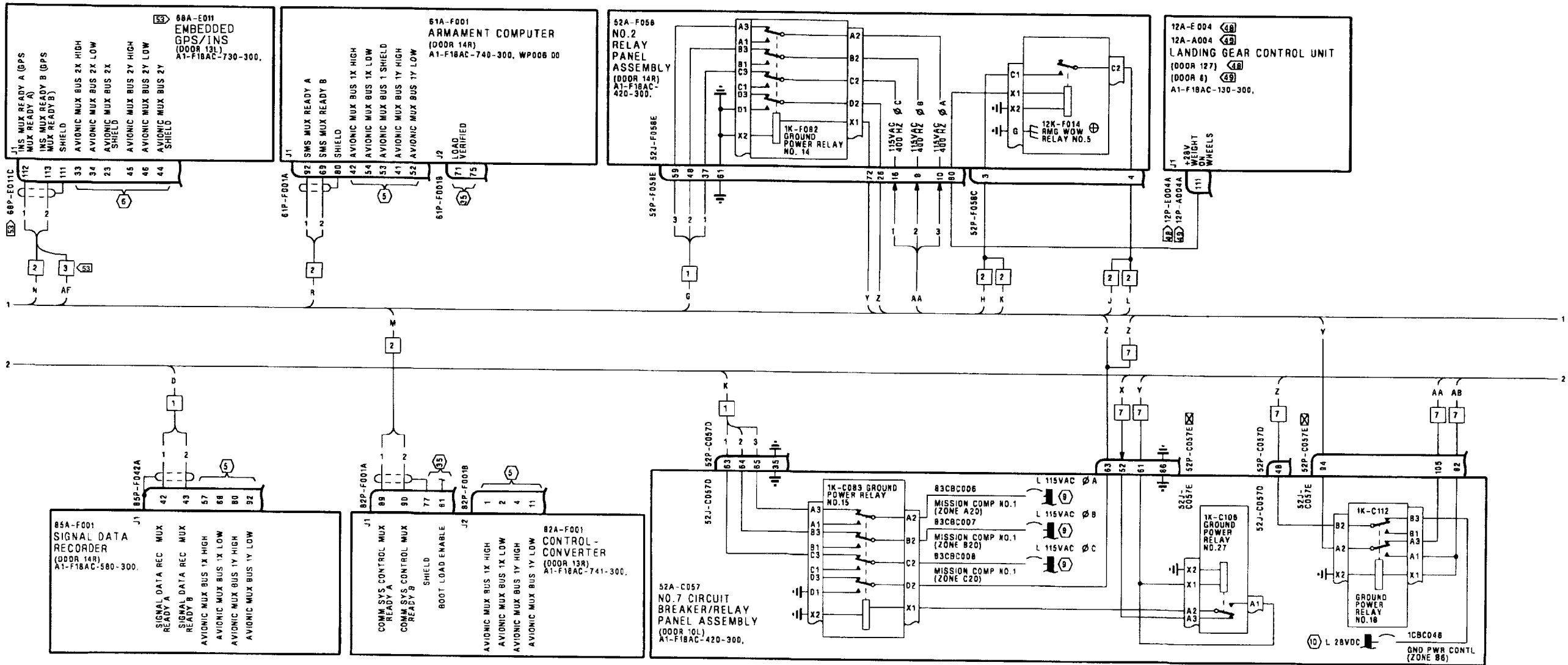


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 6)

Figure 1.

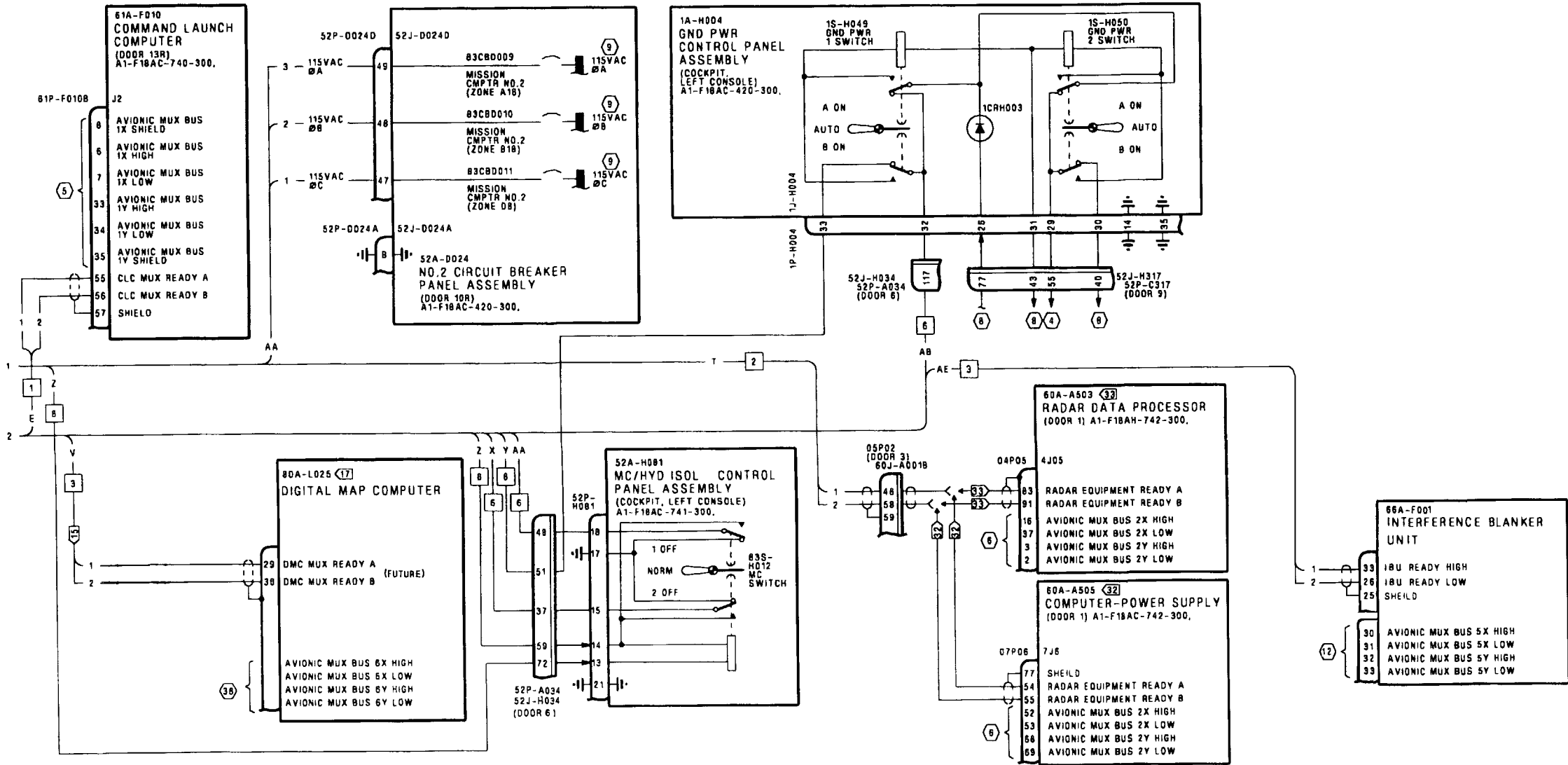


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 7)

Figure 1.

LEGEND	
1. CONTINUITY TESTS:	20 DELETED.
A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000	21 DELETED.
B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY Ⓢ) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.	22 DELETED.
C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.	23 DELETED.
D. WHEN TESTING CONTINUITY, TEST FOR:	24 DELETED.
(1) SHORTS TO GROUND.	25 DELETED.
(2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.	26 DELETED.
(3) SHORTS BETWEEN SHIELD AND CONDUCTORS.	27 DELETED.
(4) SHIELD CONTINUITY.	28 DELETED.
E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY ⓧ). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.	29 DELETED.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.	30 DELETED.
3 DELETED.	31 DELETED.
4 AC/DC POWER DISTRIBUTION SCHEMATIC, WP009 00.	32 AFTER F/A-18 AFC 253.
5 AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 01.	33 AFTER F/A-18 AFC 292.
6 AVIONIC MUX CHANNEL 2 SCHEMATIC, WP005 01.	34 DELETED.
7 AVIONIC MUX CHANNEL 3 SCHEMATIC, WP006 00.	35 MUX TEST SCHEMATIC (83J-G003), A1-F18AC-SCM-000, WP008 00.
8 GROUND POWER SWITCHING SCHEMATIC, A1-F18AC-420-500, WP005 00.	36 DELETED.
9 AC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP003 00.	37 DELETED.
10 DC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.	38 AVIONIC MUX CHANNEL 6 SCHEMATIC, WP019 00.
11 AVIONIC MUX CHANNEL 4 SCHEMATIC, WP017 00.	39 DELETED.
12 AVIONIC MUX CHANNEL 5 SCHEMATIC, WP018 00.	40 DELETED.
13 DELETED.	41 DELETED.
14 DELETED.	42 DELETED.
15 DELETED.	43 DELETED.
16 DELETED.	44 165171 AND UP WITH DIGITAL DATA COMPUTER CONFIG/IDENT 15C AND UP (A1-F18AC-SCM-000).
17 DELETED.	45 DELETED.
18 DELETED.	46 DELETED.
19 DELETED.	47 DELETED.
	48 AFTER F/A-18 AFC 292.
	49 AFTER F/A-18 AFC 253.
	50 DELETED.
	51 CONNECTOR STOWED IF COMM2 DCS INSTALLED. (RT-1824/ARC-210)
	52 CONNECTOR STOWED IF COMM2 VHF/UHF INSTALLED. (RT-1556/ARC-210)
	53 F/A-18 AFTER AFC 231 PART 2 OR PART 3.

Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Interconnect Schematic (Sheet 8)

Figure 1.

ORGANIZATIONAL MAINTENANCE
SYSTEM SCHEMATICS
SCHEMATIC - AC/DC POWER
MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292 AND F/A-18B

This WP supersedes WP 009 00, dated 1 June 1992.

Reference Material

None

Alphabetical Index

Subject	Page No.
AC/DC Power Schematic, Figure 1	2

Record of Applicable Technical Directives

None

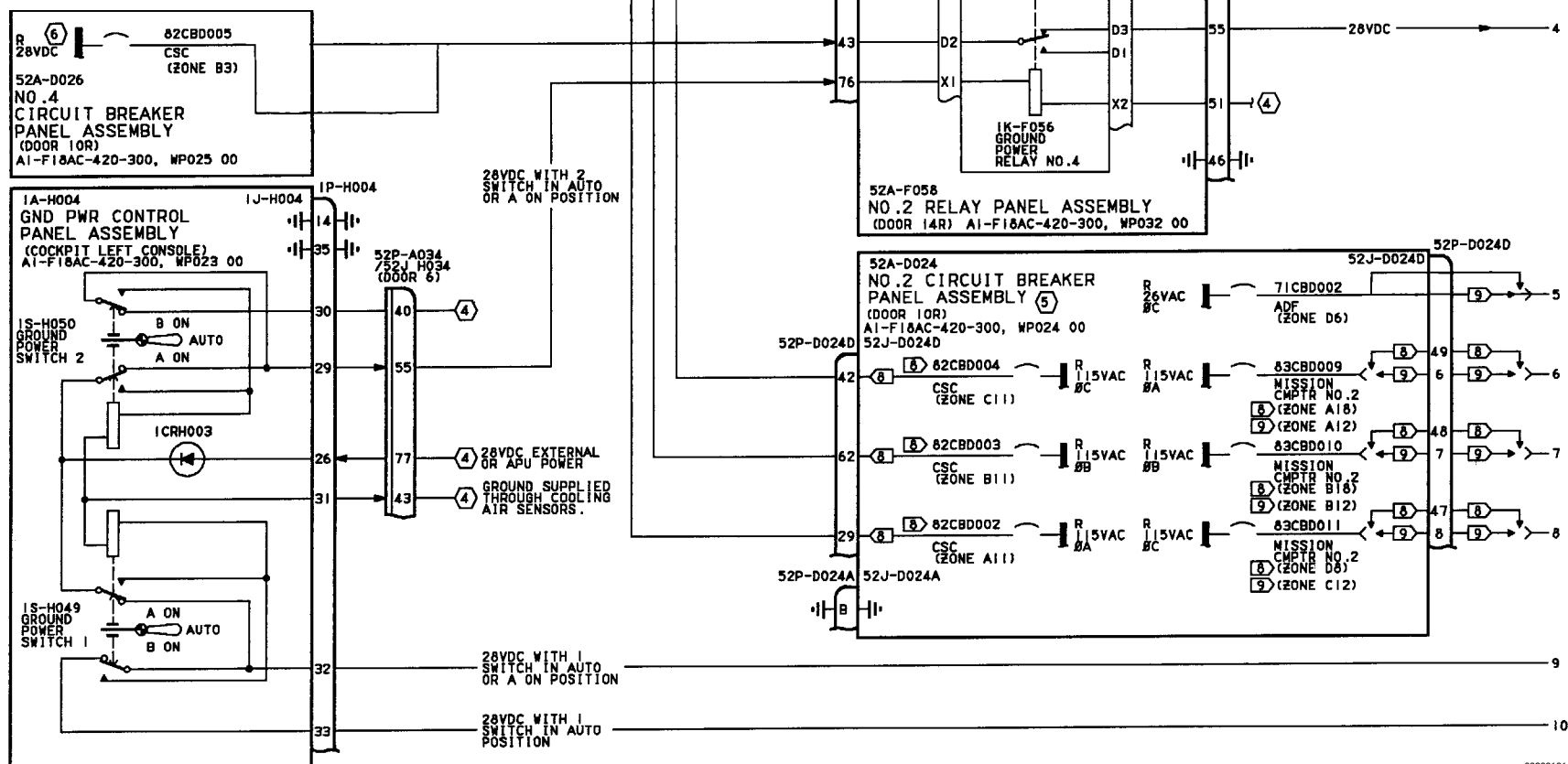


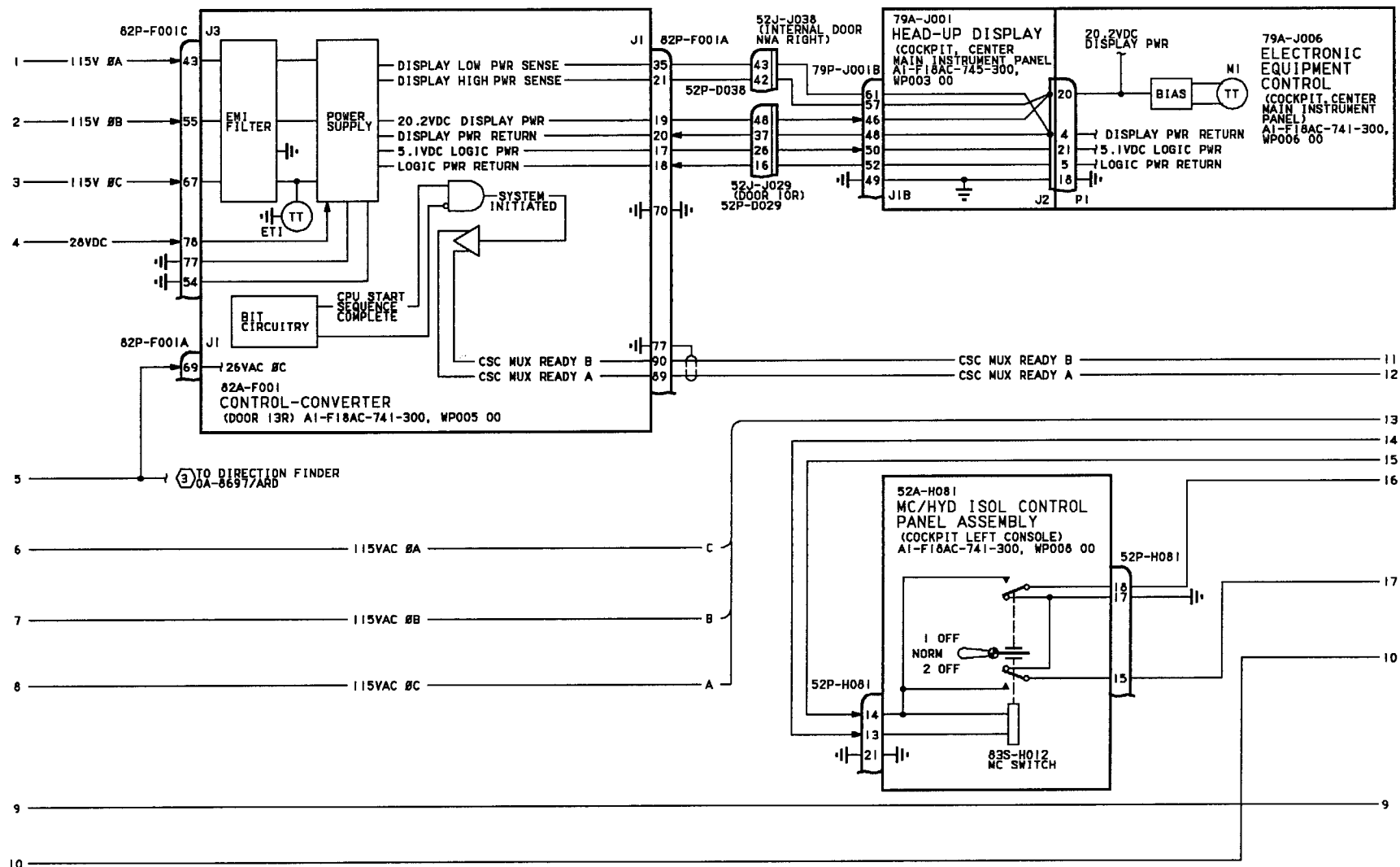
Figure 1.

Figure 1. AC/DC Power Schematic (Sheet 1)

Figure 1.

Change 2

Page 3



00900102

Figure 1.

Figure 1. AC/DC Power Schematic (Sheet 2)

Figure 1.

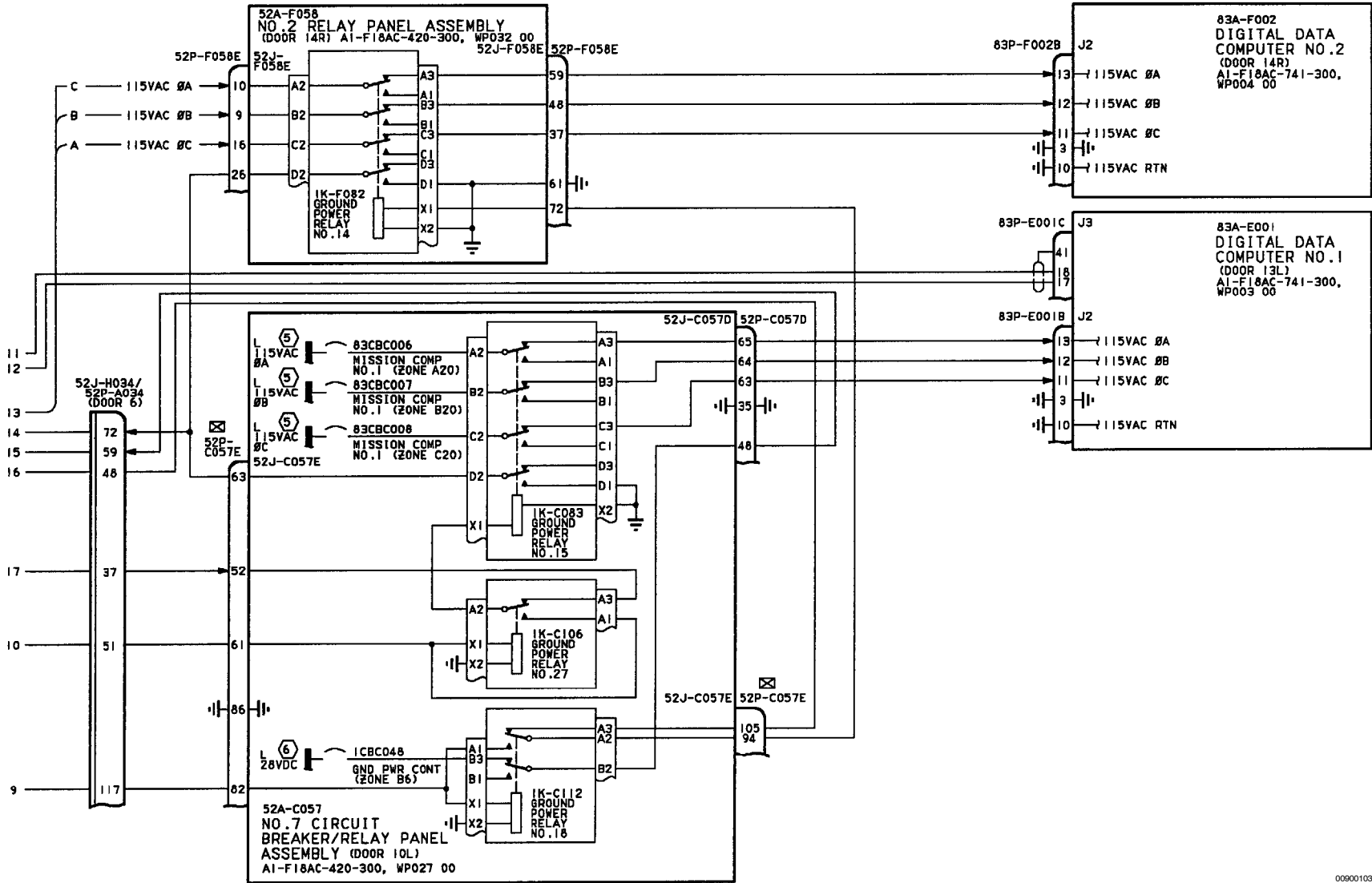


Figure 1.

Figure 1. AC/DC Power Schematic (Sheet 3)

Figure 1.

LEGEND

1. CONTINUITY TESTS:

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

③ ADF SYSTEM INTERCONNECT SCHEMATIC, A1-F18AC-600-500, WP004 00.

④ GROUND POWER SWITCHING SCHEMATIC, A1-F18AC-420-500, WP005 00.

⑤ AC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP003 00.

⑥ DC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.

7 F/A-18B.

8 161360 AND UP.

9 161353 THRU 161359.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - AC/DC POWER

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

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Subject	Page No.
AC/DC Power Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jan 01	-

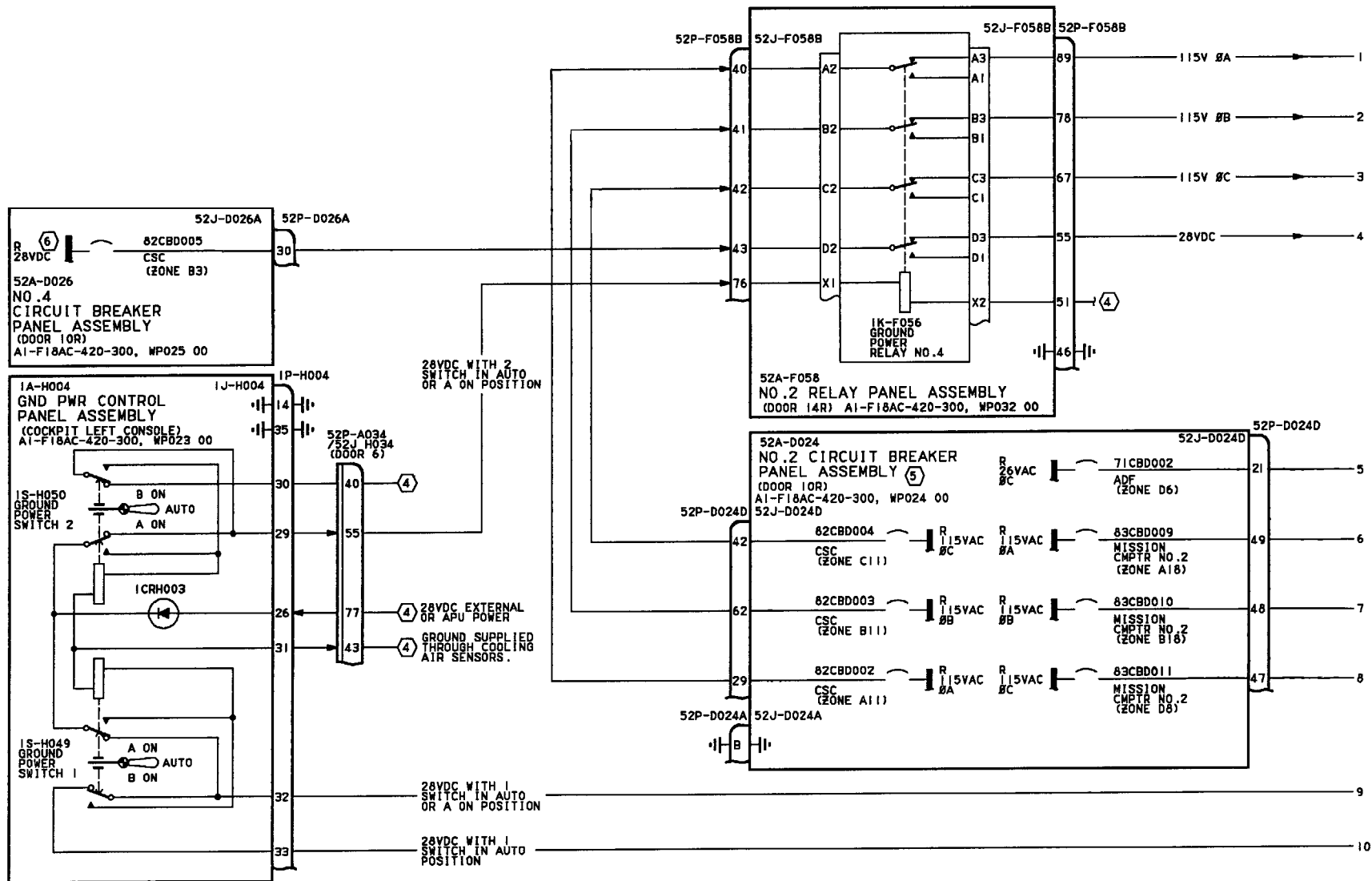


Figure 1.

Figure 1. AC/DC Power Schematic (Sheet 1)

Figure 1.

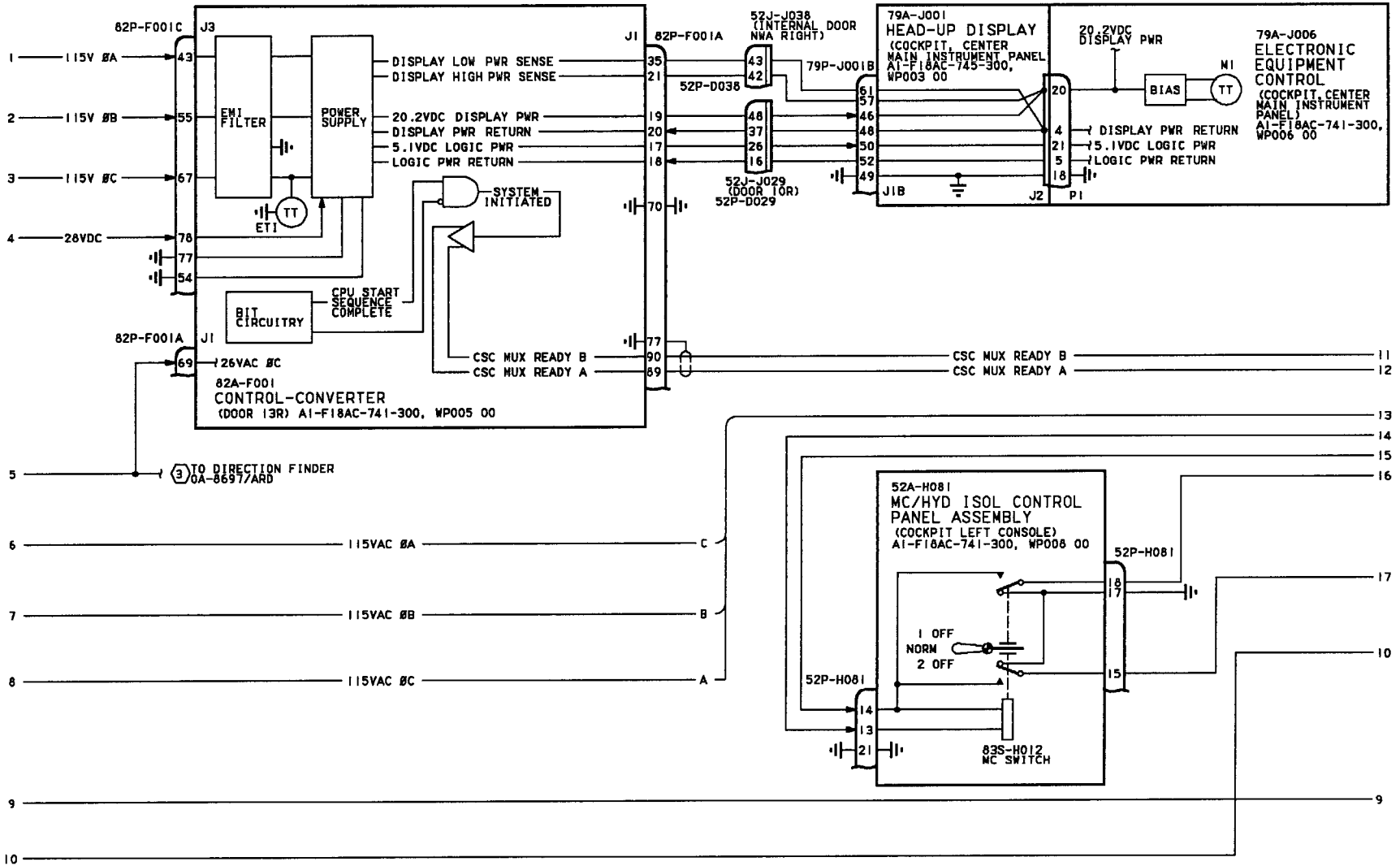


Figure 1.

Figure 1. AC/DC Power Schematic (Sheet 2)

Figure 1.

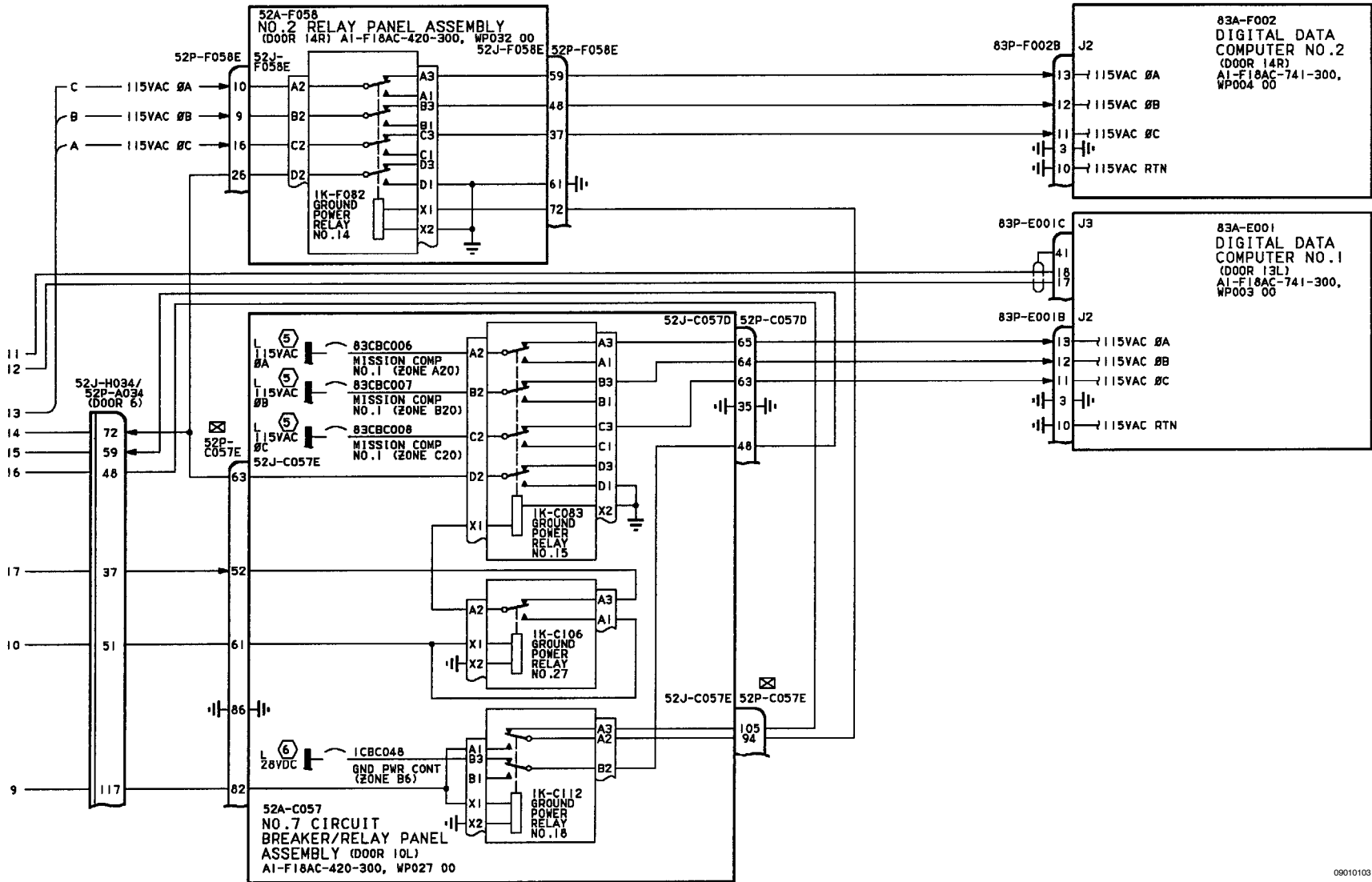


Figure 1.

Figure 1. AC/DC Power Schematic (Sheet 3)

Figure 1.

LEGEND**1. CONTINUITY TESTS:**

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
 - B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
 - C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
 - E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
- 2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.**
- (3)** ADF SYSTEM INTERCONNECT SCHEMATIC, A1-F18AC-600-500, WP004 00.
 - (4)** GROUND POWER SWITCHING SCHEMATIC, A1-F18AC-420-500, WP005 00.
 - (5)** AC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP003 00.
 - (6)** DC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - CONTROL-CONVERTER C-10382/A BUILT-IN TEST****MISSION COMPUTER SYSTEM**

This WP supersedes WP 010 00, dated 1 January 2001.

Title	Work Package
Schematic - Control-Converter C-10382/A Built-In Test (F/A-18A/B)	010 01
Schematic - Control-Converter C-10382/A Built-In Test (AFTER F/A-18 AFC 225)	010 02
Schematic - Control-Converter C-10382/A Built-In Test (AFTER F/A-18 AFC 253 OR F/A-18 AFC 292)	010 03

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - CONTROL-CONVERTER C-10382/A BUILT-IN TEST

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A/B

This WP supersedes WP 010 01, dated 1 January 2001.

Reference Material

None

Alphabetical Index

Subject	Page No.
Control-Converter C-10382/A Built-In Test Schematic, Figure 1	2

Record of Applicable Technical Directives

None

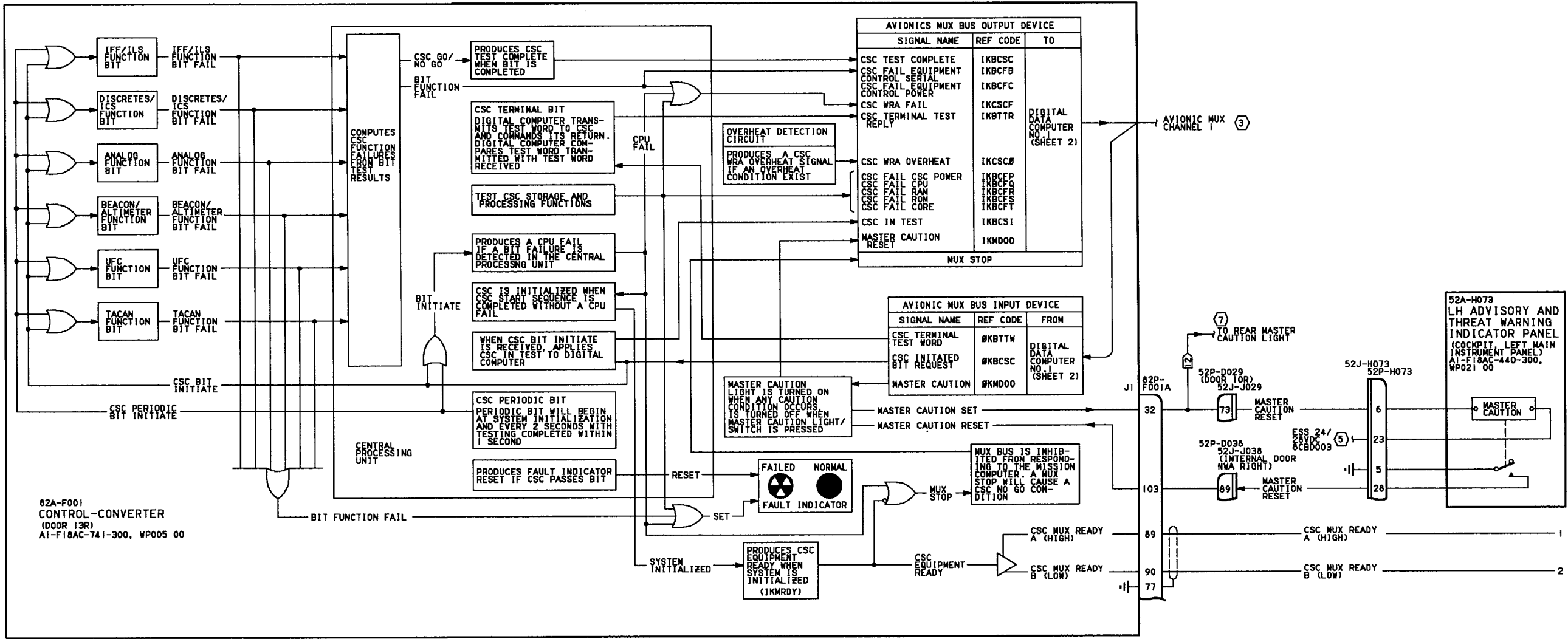


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 1)

Figure 1.

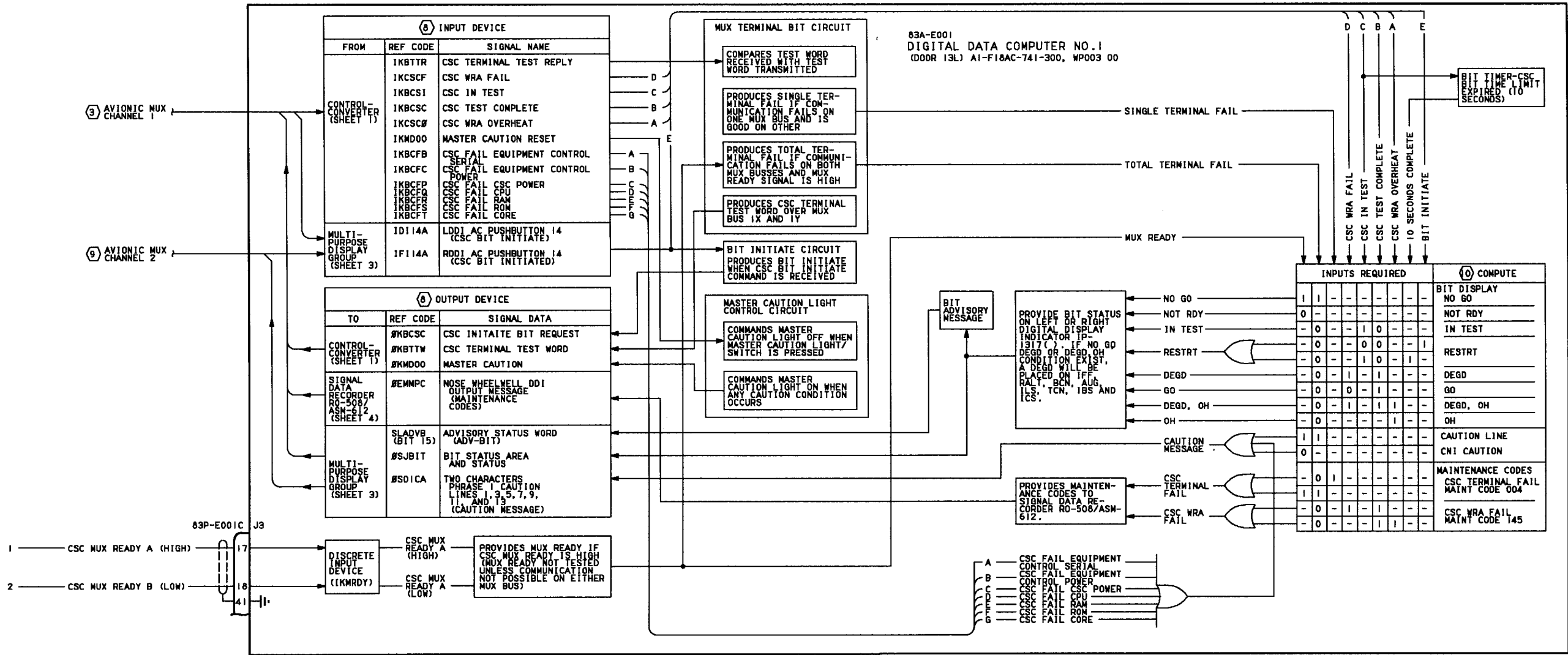


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 2)

Figure 1.

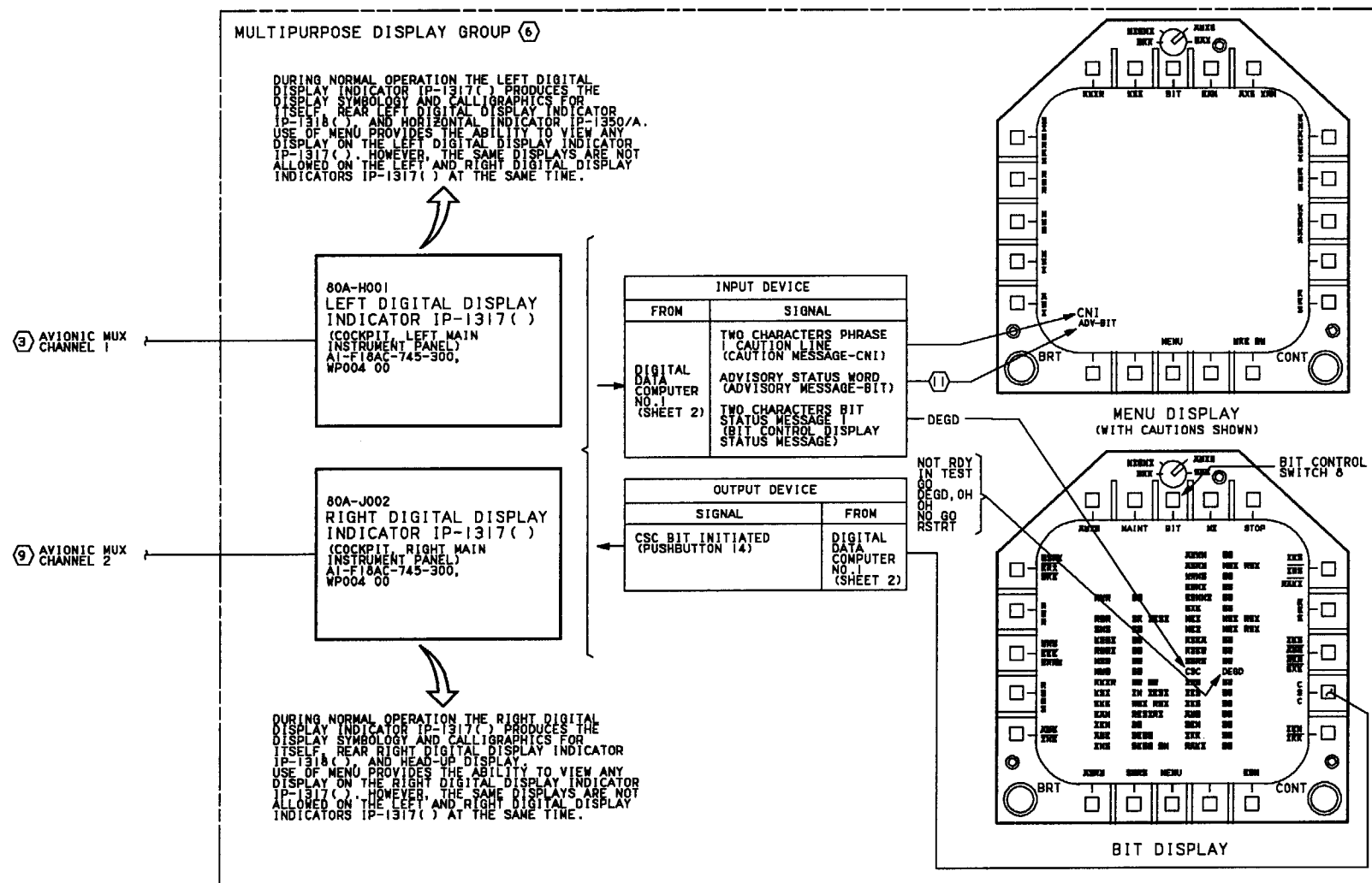


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 3)

Figure 1.

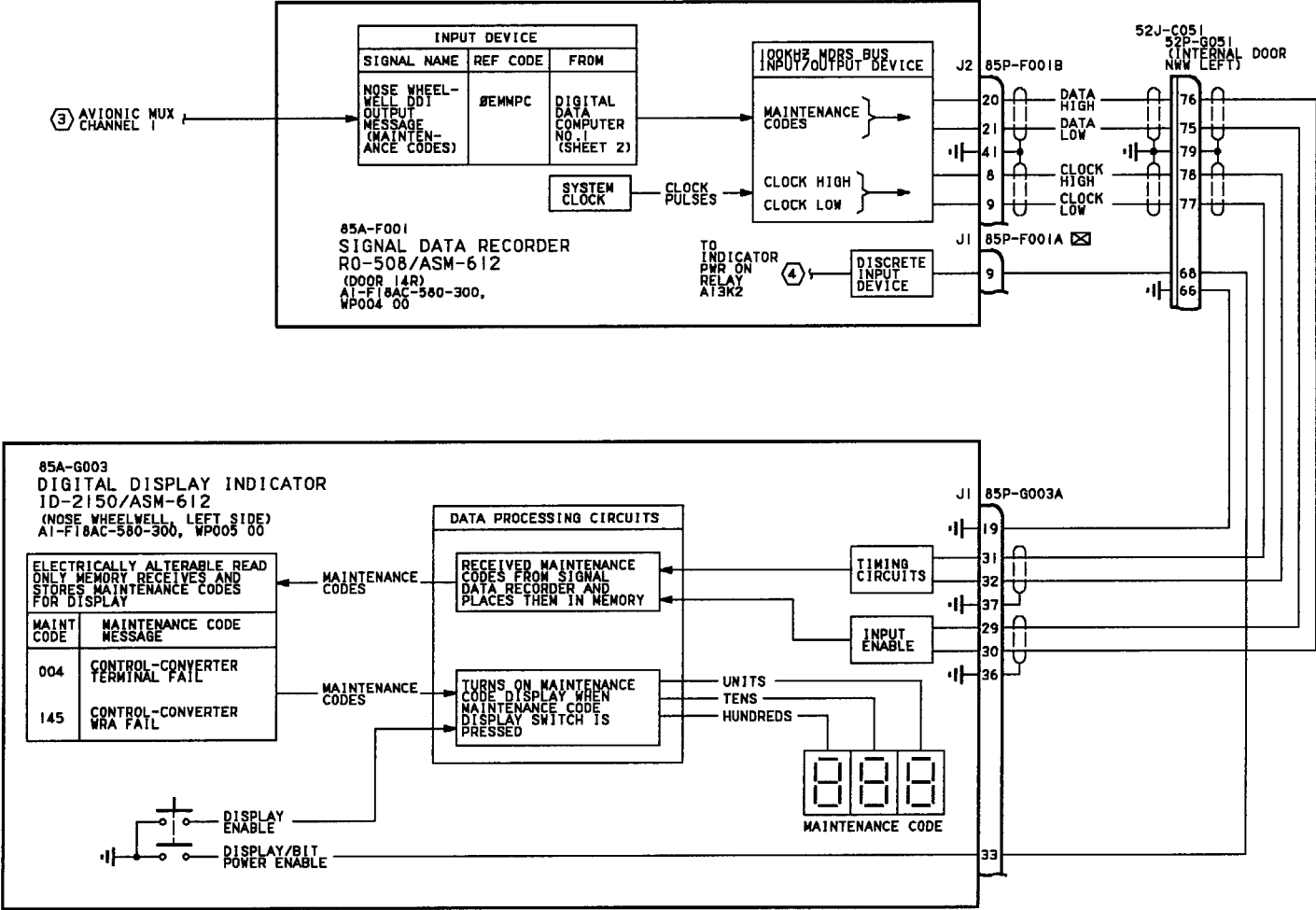


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 4)

Figure 1.

LEGEND

1. CONTINUITY TESTS:

A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.

B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY ⊕) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.

C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.

D. WHEN TESTING CONTINUITY, TEST FOR:
(1) SHORTS TO GROUND.
(2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
(3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
(4) SHIELD CONTINUITY.

E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY ☒). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

③ AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 00.

④ MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM POWER AND START SEQUENCE SCHEMATIC, A1-F18AC-580-600, WP005 00.

⑤ WARNING/CAUTION/ADVISORY LIGHTING SYSTEM SCHEMATIC, A1-F18AC-440-500, WP006 00.
- ⑥ THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD UP DISPLAY, HORIZONTAL INDICATOR IP-1360/A, ALSO ON F/A-18B; REAR LEFT DIGITAL DISPLAY INDICATOR IP-1318(), REAR RIGHT DIGITAL DISPLAY INDICATOR IP-1918(), AND REAR CENTER DIGITAL DISPLAY INDICATOR IP-1317(). FOR MULTIPURPOSE DISPLAY GROUP SCHEMATIC, REFER TO A1-F18AC-745-500, WP001 00.

⑦ REAR WARNING/CAUTION/ADVISORY LIGHTING SYSTEM SCHEMATIC, A1-F18AC-440-500, WP007 00.

⑧ FOR MEMORY INSPECT ADDRESS LOCATION RELATING TO REF CODE, REFER TO A1-F18AC-FIM-100.

⑨ AVIONIC MUX CHANNEL 2 SCHEMATIC, WP005 00.

⑩ DESCRIPTION OF MATRIX:

A. COMPUTE: COLUMN LISTS THE SIGNAL OUTPUT.

B. INPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUT.

C. THE SIGNAL OUTPUT IS READ HORIZONTALLY. EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL.

D. INTERPRET MATRIX TABLE AS INDICATED.
(1) ONE (1) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.
(2) ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE TO GET THE OUTPUT.
(3) DASH (-) INDICATES THE OUTPUT DOES NOT DEPEND ON THIS INPUT.

⑪ WHEN CSC FAIL CONDITION EXISTS, THE MISSION COMPUTER SYSTEM COMMANDS ADV-BIT TO BE DISPLAYED. TO DISPLAY THE STATUS MESSAGE, BIT CONTROL DISPLAY (SWITCH 8) IS PRESSED.

12 F/A-18B.

Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 5)

Figure 1.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - CONTROL-CONVERTER C-10382/A BUILT-IN TEST****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AND F/A-18B AFTER F/A-18 AFC 225**

Reference Material

None

Alphabetical Index**Subject****Page No.**

Control-Converter C-10382/A Built-In Test Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 225	-	Five (5) Avionics Multiplex Bus Upgrade, Incorporation of (ECP MDA-F/A-18 0529)	1 Jun 02	-

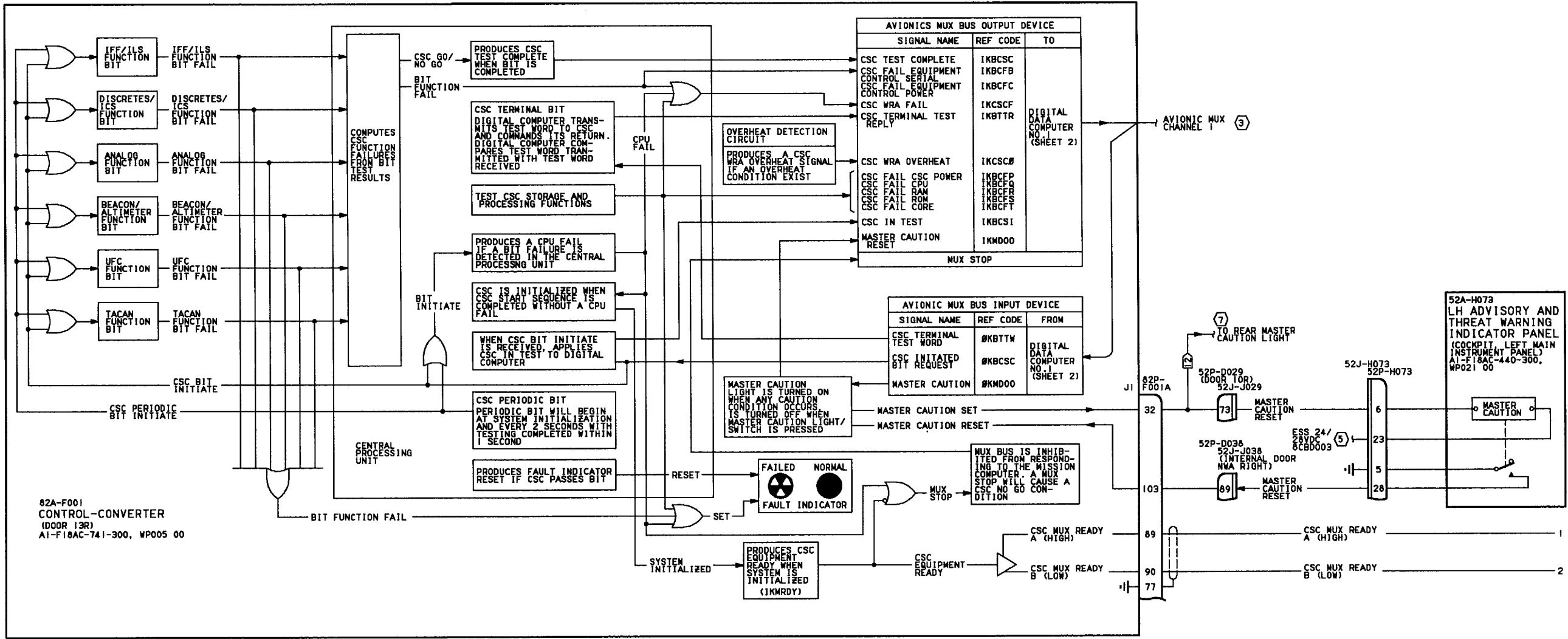


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 1)

Figure 1.

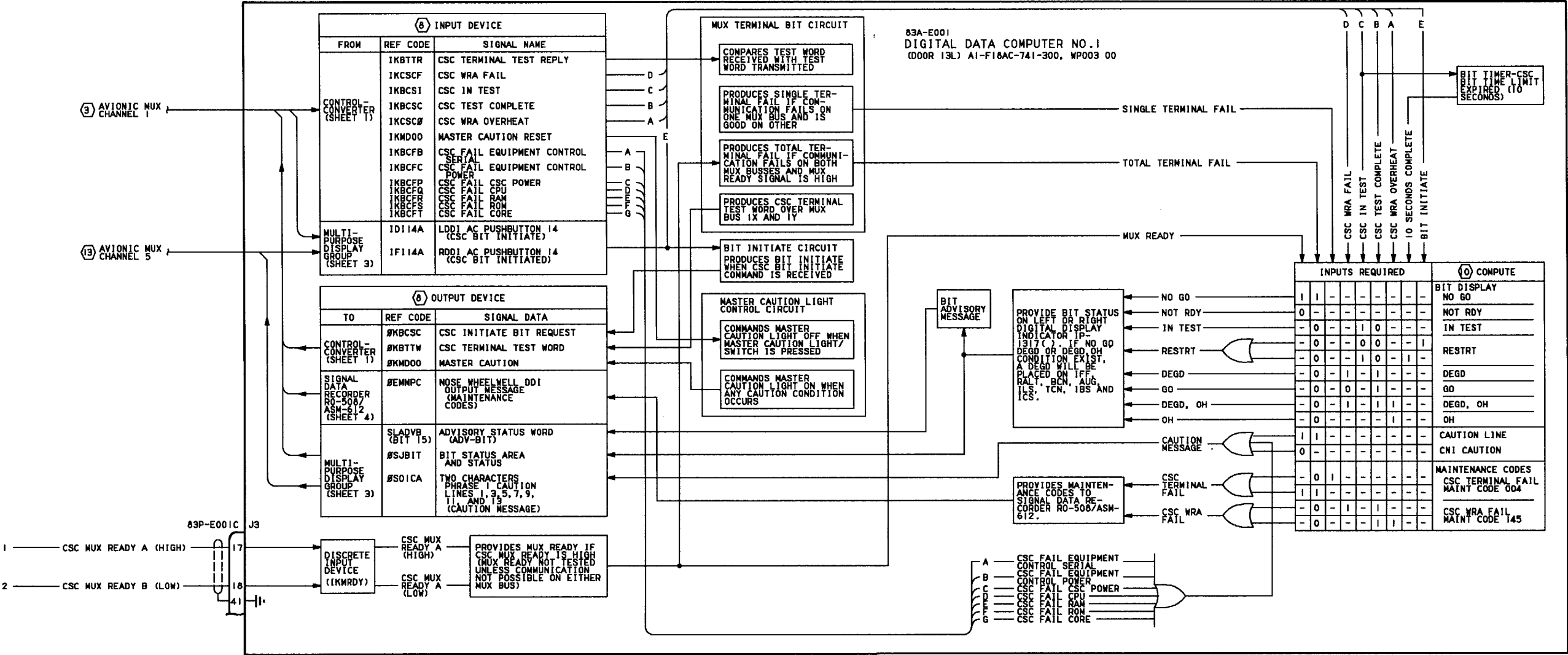


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 2)

Figure 1.

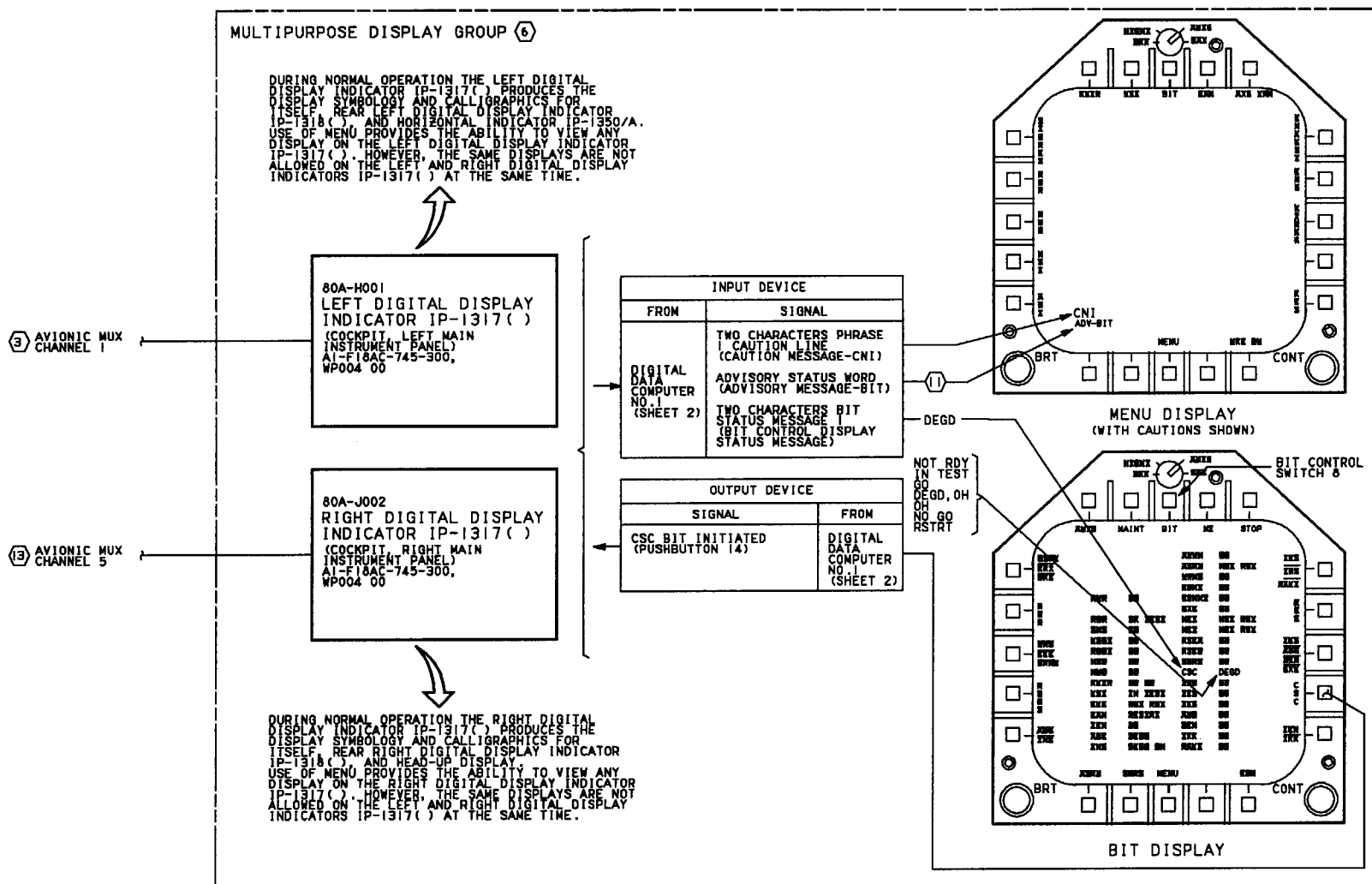


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 3)

Figure 1.

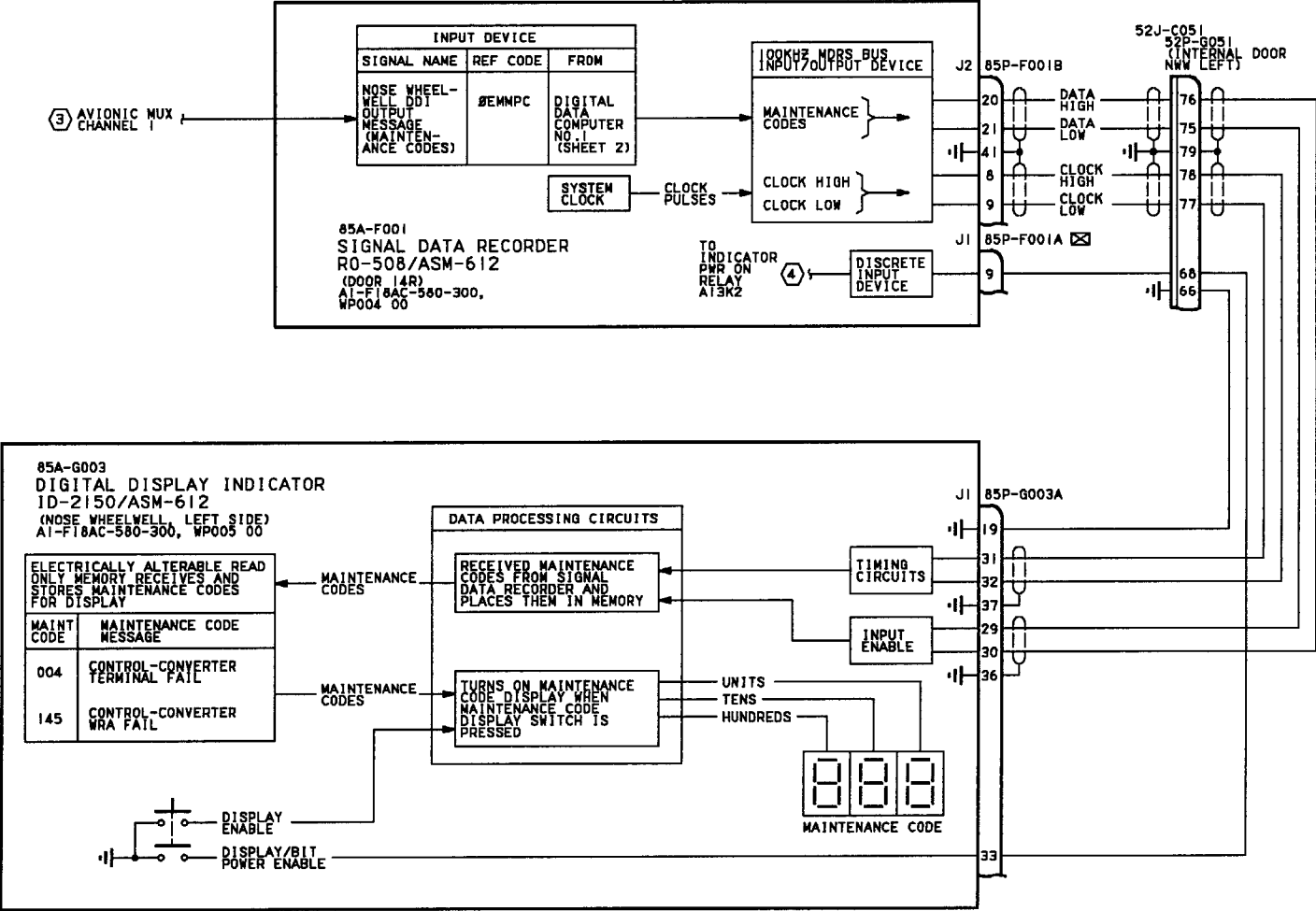


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 4)

Figure 1.

LEGEND

1. CONTINUITY TESTS:

A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.

B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.

C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.

D. WHEN TESTING CONTINUITY, TEST FOR:

(1) SHORTS TO GROUND.

(2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.

(3) SHORTS BETWEEN SHIELD AND CONDUCTORS.

(4) SHIELD CONTINUITY.

E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

3. AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 00.

4. MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM POWER AND START SEQUENCE SCHEMATIC, A1-F18AC-580-600, WP005 00.

5. WARNING/CAUTION/ADVISORY LIGHTING SYSTEM SCHEMATIC, A1-F18AC-440-500, WP006 00.
6. THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD UP DISPLAY, HORIZONTAL INDICATOR IP-1360/A, ALSO ON F/A-18B; REAR LEFT DIGITAL DISPLAY INDICATOR IP-1318(), REAR RIGHT DIGITAL DISPLAY INDICATOR IP-1918(), AND REAR CENTER DIGITAL DISPLAY INDICATOR IP-1317(). FOR MULTIPURPOSE DISPLAY GROUP SCHEMATIC, REFER TO A1-F18AC-745-500, WP001 00.

7. REAR WARNING/CAUTION/ADVISORY LIGHTING SYSTEM SCHEMATIC, A1-F18AC-440-500, WP007 00.

8. FOR MEMORY INSPECT ADDRESS LOCATION RELATING TO REF CODE, REFER TO A1-F18AC-FIM-100.

9. AVIONIC MUX CHANNEL 2 SCHEMATIC, WP005 00.

10. DESCRIPTION OF MATRIX:

A. COMPUTE: COLUMN LISTS THE SIGNAL OUTPUT.

B. INPUTS REQUIRED ARE USED TO DEVELOP THE SIGNAL OUTPUT.

C. THE SIGNAL OUTPUT IS READ HORIZONTALLY. EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL.

D. INTERPRET MATRIX TABLE AS INDICATED.

(1) ONE (1) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.

(2) ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE TO GET THE OUTPUT.

(3) DASH (-) INDICATES THE OUTPUT DOES NOT DEPEND ON THIS INPUT.

11. WHEN CSC FAIL CONDITION EXISTS, THE MISSION COMPUTER SYSTEM COMMANDS ADV-BIT TO BE DISPLAYED. TO DISPLAY THE STATUS MESSAGE, BIT CONTROL DISPLAY (SWITCH 8) IS PRESSED.

12. F/A-18B.

13. AVIONIC MUX CHANNEL 5 SCHEMATIC, WP018 01.

Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 5)

Figure 1.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - CONTROL-CONVERTER C-10382/A BUILT-IN TEST****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AFTER AFC 253 OR AFC 292**

Reference Material

None

Alphabetical Index**Subject****Page No.**

Control-Converter C-10382/A Built-In Test Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	U.S. Naval Reserves A+ Avionics Upgrade, Incorporation of (ECP MDA-F/A-18 0560R1)	-	-
F/A-18 AFC 292	-	U.S. Marine Corps Reserves A+ Avionics Upgrade, Incorporation of (ECP MDA-F/A-18 0583)	-	-

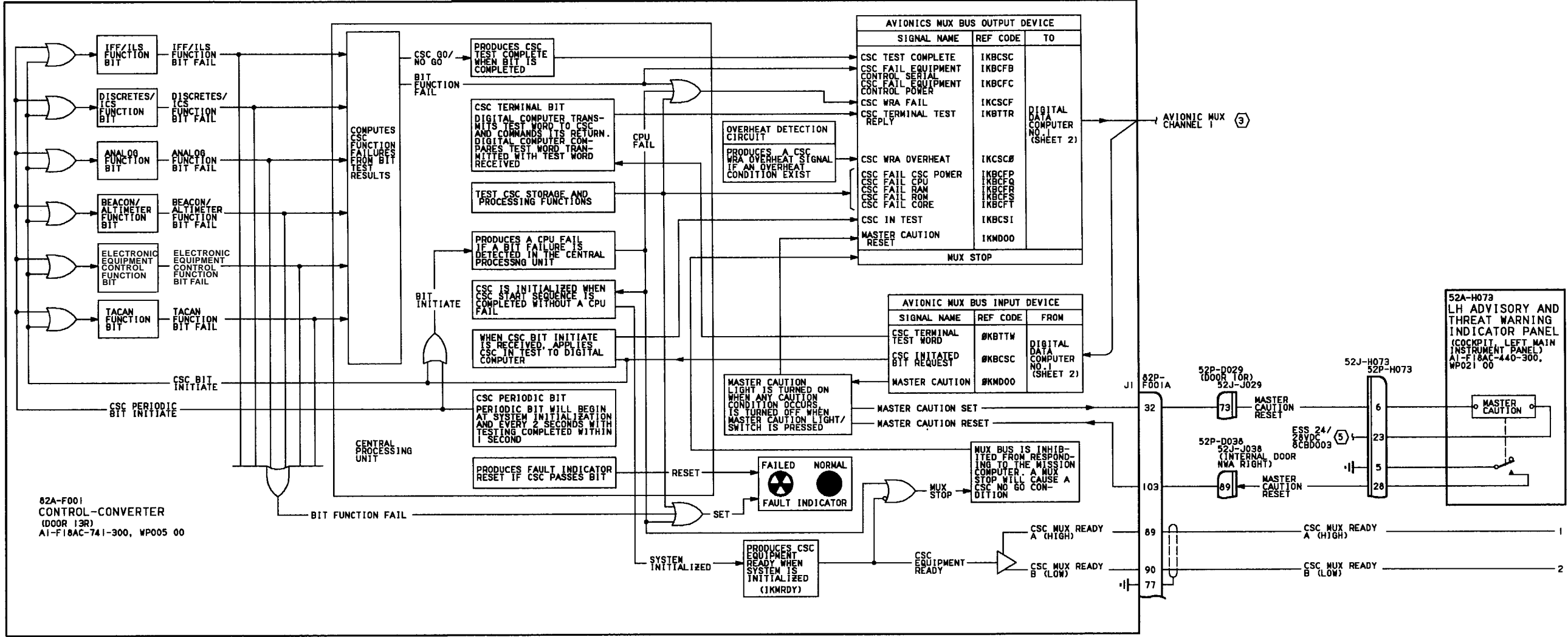


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 1)

Figure 1.

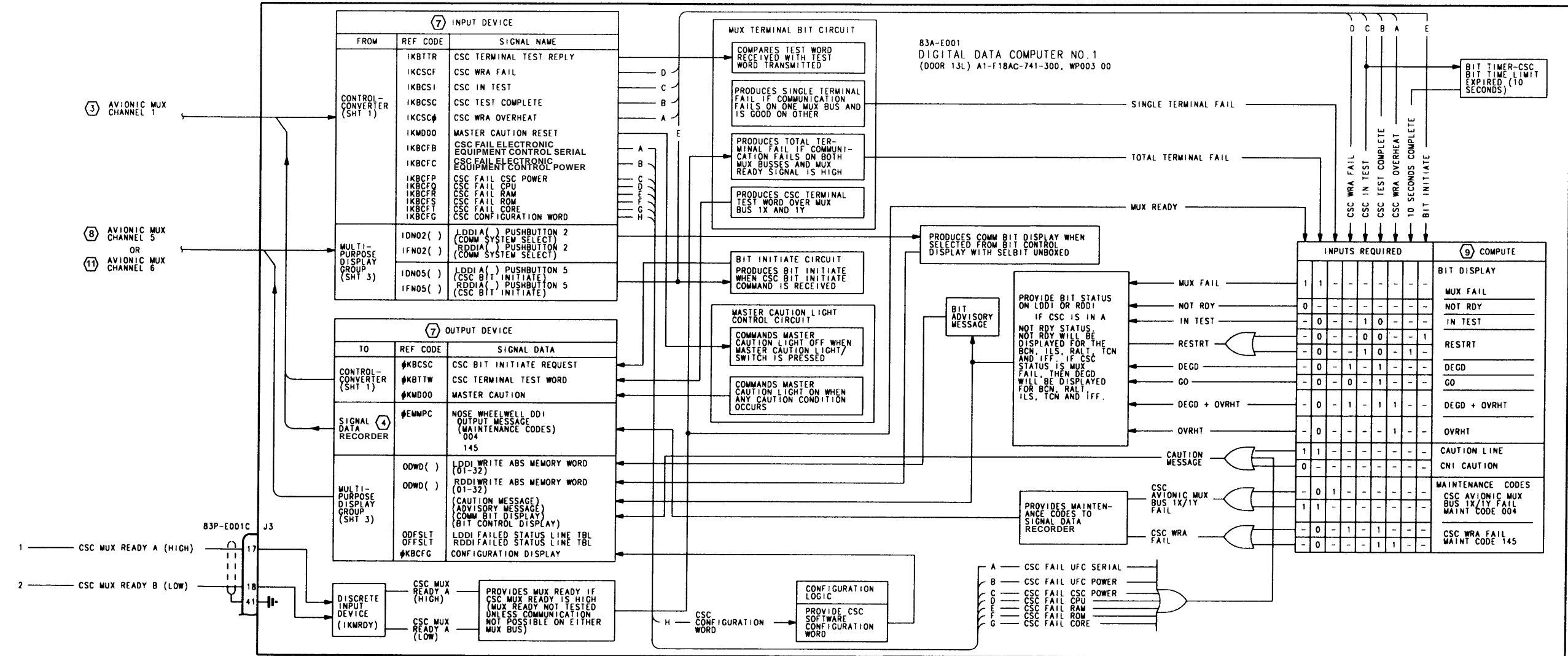


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 2)

Figure 1.

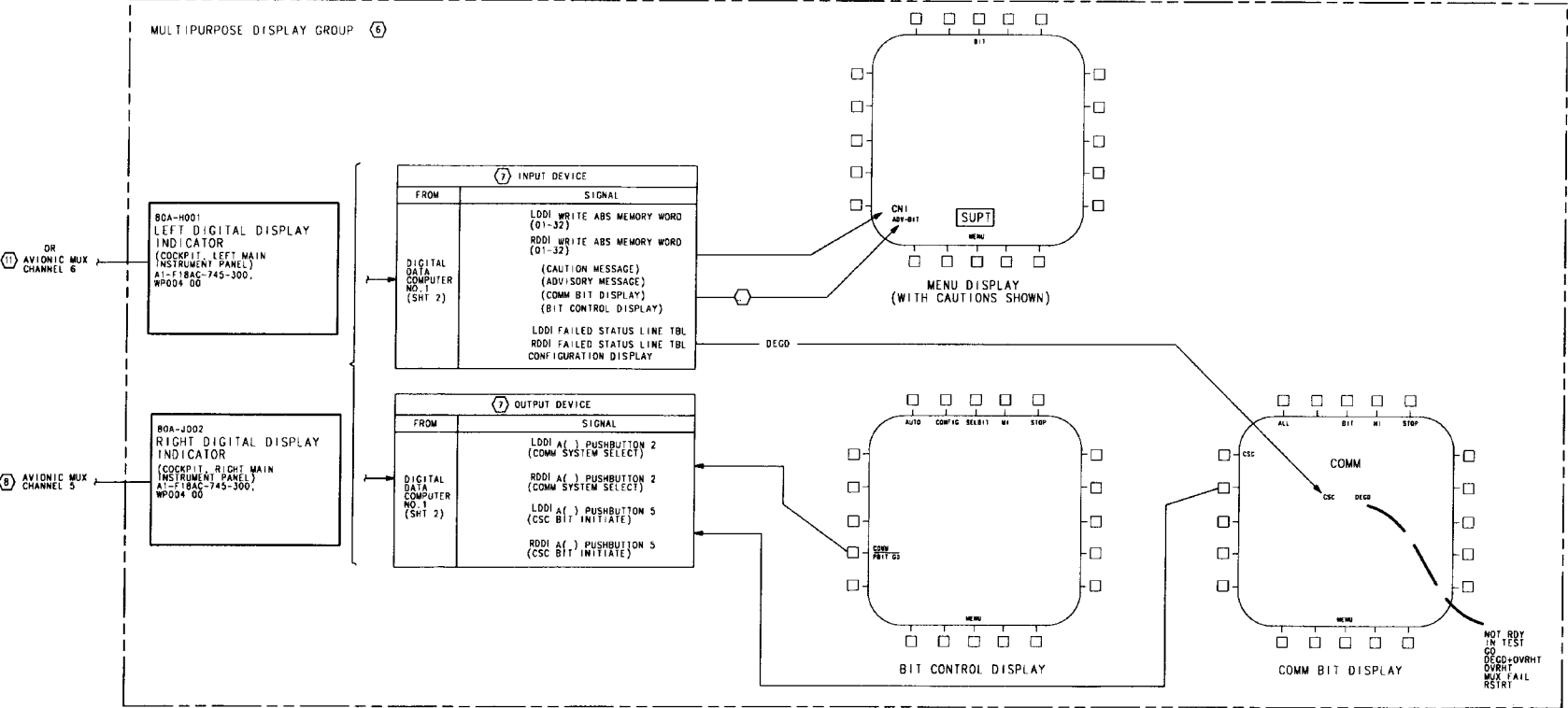


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 3)

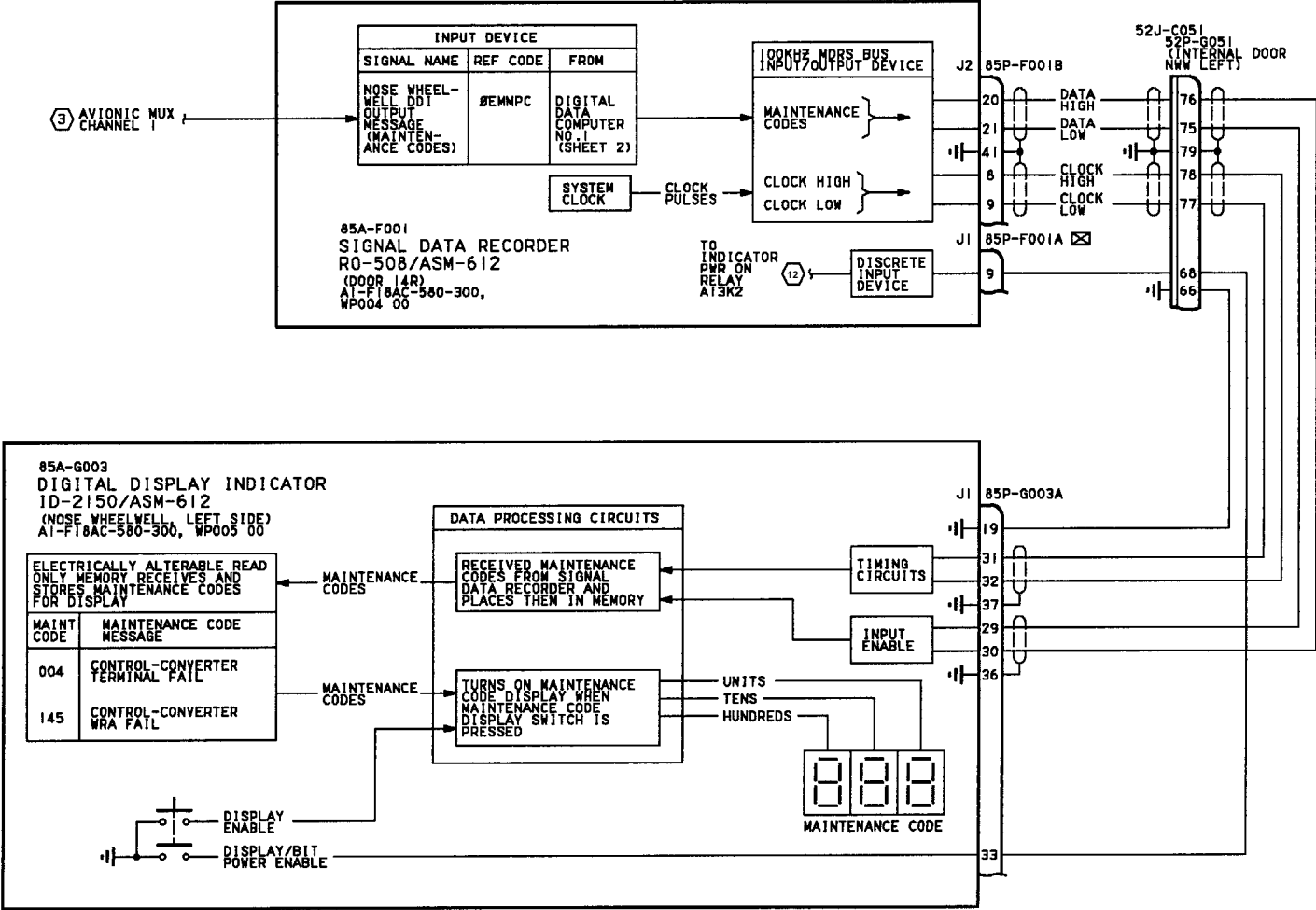


Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 4)

Figure 1.

LEGEND

1. CONTINUITY TESTS:

A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.

B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.

C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.

D. WHEN TESTING CONTINUITY, TEST FOR:

(1) SHORTS TO GROUND.

(2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.

(3) SHORTS BETWEEN SHIELD AND CONDUCTORS.

(4) SHIELD CONTINUITY.

E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

3. AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 01.

4. FAULT REPORTING INTEGRATION SCHEMATIC, WP016 01.

5. WARNING/CAUTION/ADVISORY LIGHTING SYSTEM SCHEMATIC, A1-F18AC-440-500, WP006 00.

6. THE MULTIPURPOSE DISPLAY GROUP INTERCONNECT SCHEMATIC, A1-F18AC-745-500, WP004 00.

7. FOR MEMORY INSPECT ACCESS LOCATION RELATING TO REF CODE, REFER TO A1-F18AC-FIM-100.

8. AVIONIC MUX CHANNEL 5 SCHEMATIC, WP018 00.

9. DESCRIPTION OF MATRIX:

A. COMPUTE COLUMN LISTS THE SIGNAL OUTPUT.

B. INPUTS REQUIRED ARE USED TO DEVELOP THE OUTPUT SIGNAL.

C. THE SIGNAL IS READ HORIZONTALLY. EACH HORIZONTAL LINE IS AN INDEPENDENT SIGNAL OUTPUT.

D. INTERPRET MATRIX TABLE AS INDICATED:

(1) ONE (1) INDICATES THIS INPUT AS NAMED MUST BE THERE TO GET THE OUTPUT.

(2) ZERO (0) INDICATES THIS INPUT AS NAMED MUST NOT BE THERE TO GET THE OUTPUT.

(3) DASH (-) INDICATES THE OUTPUT DOES NOT DEPEND ON THIS INPUT.

10. WHEN CSC FAIL CONDITION EXISTS, THE MISSION COMPUTER SYSTEM COMMANDS ADV-BIT TO BE DISPLAYED. TO DISPLAY THE STATUS MESSAGE, BIT CONTROL DISPLAY (SWITCH 8) IS PRESSED.

11. AVIONICS MUX CHANNEL 6 SCHEMATIC, WP019 00.

12. MAINTENANCE STATUS DISPLAY AND RECORDING SYSTEM POWER AND START SEQUENCE SCHEMATIC, A1-F18AC-580-600, WP005 00.

Figure 1.

Figure 1. Control-Converter C-10382/A Built-In Test Schematic (Sheet 5)

Figure 1.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - ELECTRONIC EQUIPMENT CONTROL C-10380/ASQ BUILT-IN TEST****MISSION COMPUTER SYSTEM**

This WP supersedes WP 011 00, dated 1 January 2001.

Title	Work Package
Schematic - Electronic Equipment Control C-10380/ASQ Built-In Test (BEFORE F/A-18 AFC 225, F/A-18 AFC 253, AND F/A-18 AFC 292)	011 01
Schematic - Electronic Equipment Control C-10380/ASQ Built-In Test (AFTER F/A-18 AFC 225)	011 02
Schematic - Electronic Equipment Control C-10380/ASQ Built-In Test (AFTER F/A-18 AFC 253 OR F/A-18 AFC 292)	011 03

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - ELECTRONIC EQUIPMENT CONTROL C-10380/ASQ BUILT-IN TEST

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A/B

This WP supersedes WP 011 01, dated 1 January 2001.

Reference Material

None

Alphabetical Index

Subject	Page No.
Electronic Equipment Control C-10380/ASQ Built-In Test Schematic, Figure 1	2

Record of Applicable Technical Directives

None

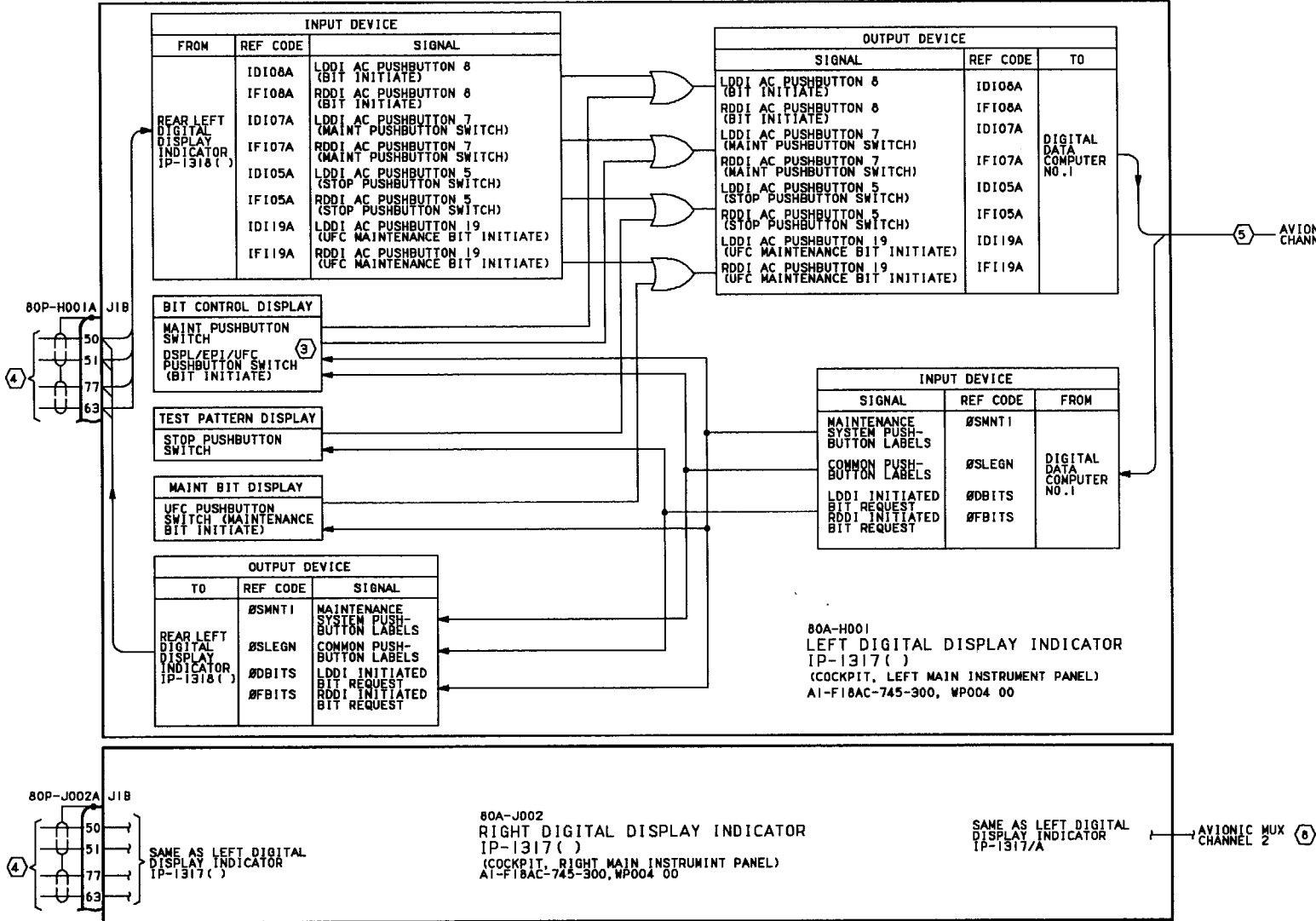


Figure 1.

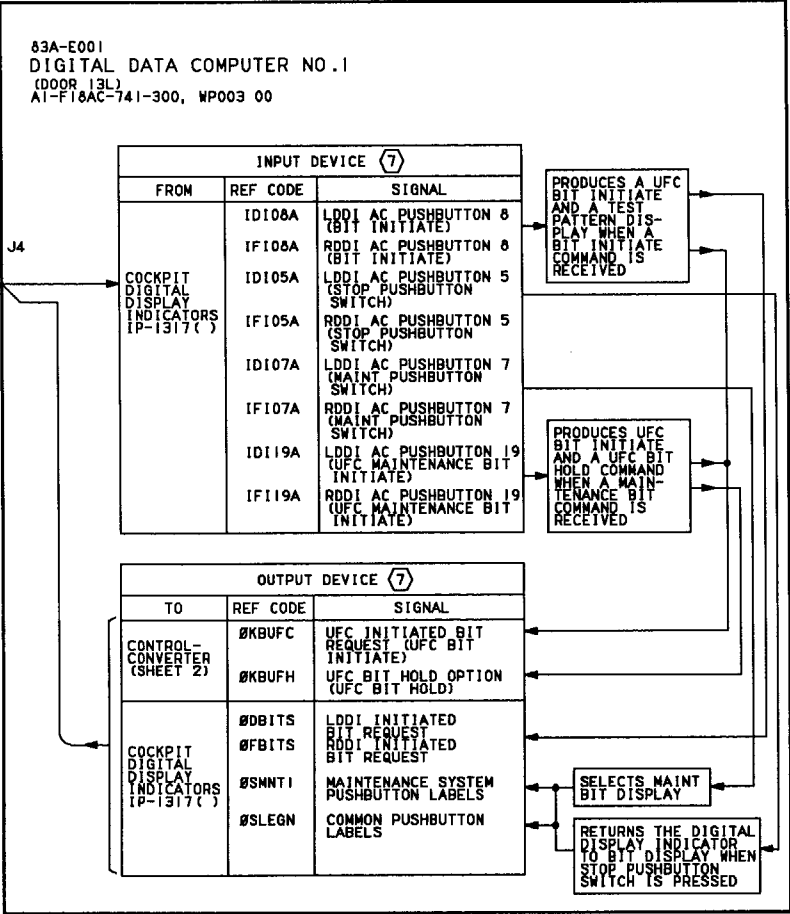


Figure 1. Electronic Equipment Control C-10380/ASQ Built-In Test Schematic (Sheet 1)

Figure 1.

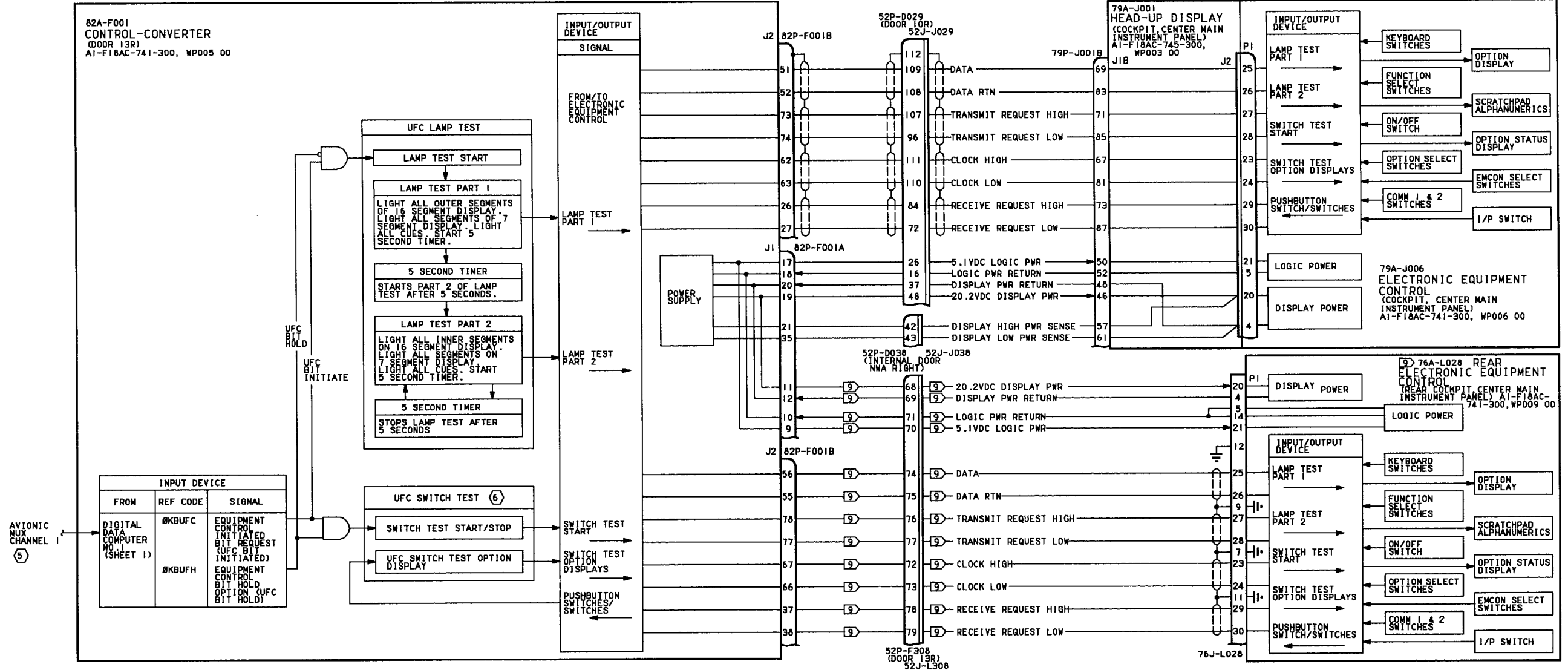


Figure 1.

Figure 1. Electronic Equipment Control C-10380/ASQ Built-In Test Schematic (Sheet 2)

Figure 1.

LEGEND

1. CONTINUITY TESTS:
- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000,
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
- (1) SHORTS TO GROUND.
- (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
- (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
- (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.
- ③ WHEN THE DSPL/EPI/UFC PUSHBUTTON SWITCH IS PRESSED, BIT TESTS ARE RUN SIMULTANEOUSLY ON THE MULTIPURPOSE DISPLAY GROUP, ENGINE MONITOR DISPLAY AND ELECTRONIC EQUIPMENT CONTROL.
- ④ MULTIPURPOSE DISPLAY GROUP INTERCONNECT SCHEMATIC, A1-F18AC-745-500, WP004 00.
- ⑤ AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 00.
- ⑥ SWITCH TEST OPTION 1 DISPLAY

PUSHBUTTON SWITCH/SWITCH	CHAR 1	CHAR 2	CHAR 3	CHAR 4
A/P FUNCTION SELECT	A	/	P	
IFF FUNCTION SELECT	I	F	F	
TCN FUNCTION SELECT	T	C	N	
ILS FUNCTION SELECT	I	L	S	
D/L FUNCTION SELECT	D	/	L	
BCN FUNCTION SELECT	B	C	N	

Figure 1.

PUSHBUTTON SWITCH/SWITCH	CHAR 1	CHAR 2	CHAR 3	CHAR 4
ON/OFF	O	N		
EMCON SELECT	E	M	C	N
OPTION 1 SELECT	O	P	T	1
OPTION 2 SELECT	O	P	T	2
OPTION 3 SELECT	O	P	T	3
OPTION 4 SELECT	O	P	T	4
OPTION 5 SELECT	O	P	T	5
COMM 1 CHANNEL SELECTS (PULL)	C	O	M	1
COMM 2 CHANNEL SELECTS (PULL)	C	O	M	2
I/P	I	/	P	
KEYBOARD 0	0			
KEYBOARD 1	1			
KEYBOARD 2	2			
KEYBOARD 3	3			
KEYBOARD 4	4			
KEYBOARD 5	5			
KEYBOARD 6	6			
KEYBOARD 7	7			
KEYBOARD 8	8			
KEYBOARD 9	9			
KEYBOARD CLR	C	L	R	
KEYBOARD ENT	E	N	T	

- ⑦ FOR MEMORY INSPECT ADDRESS LOCATION RELATING TO REF CODE, REFER TO A1-F18AC-FIM-100.
- ⑧ AVIONIC MUX CHANNEL 2 SCHEMATIC, WP005 00.
- 9 F/A-18B.

Figure 1. Electronic Equipment Control C-10380/ASQ Built-In Test Schematic (Sheet 3)

Figure 1.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - ELECTRONIC EQUIPMENT CONTROL C-10380/ASQ BUILT-IN TEST****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AND F/A-18B AFTER F/A-18 AFC 225**

Reference Material

None

Alphabetical Index**Subject****Page No.**

Electronic Equipment Control C-10380/ASQ Built-In Test Schematic, Figure 1

2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 225	-	Five (5) Avionics Multiplex Bus Upgrade, Incorporation of (ECP MDA-F/A-18 0529)	1 Jun 02	-

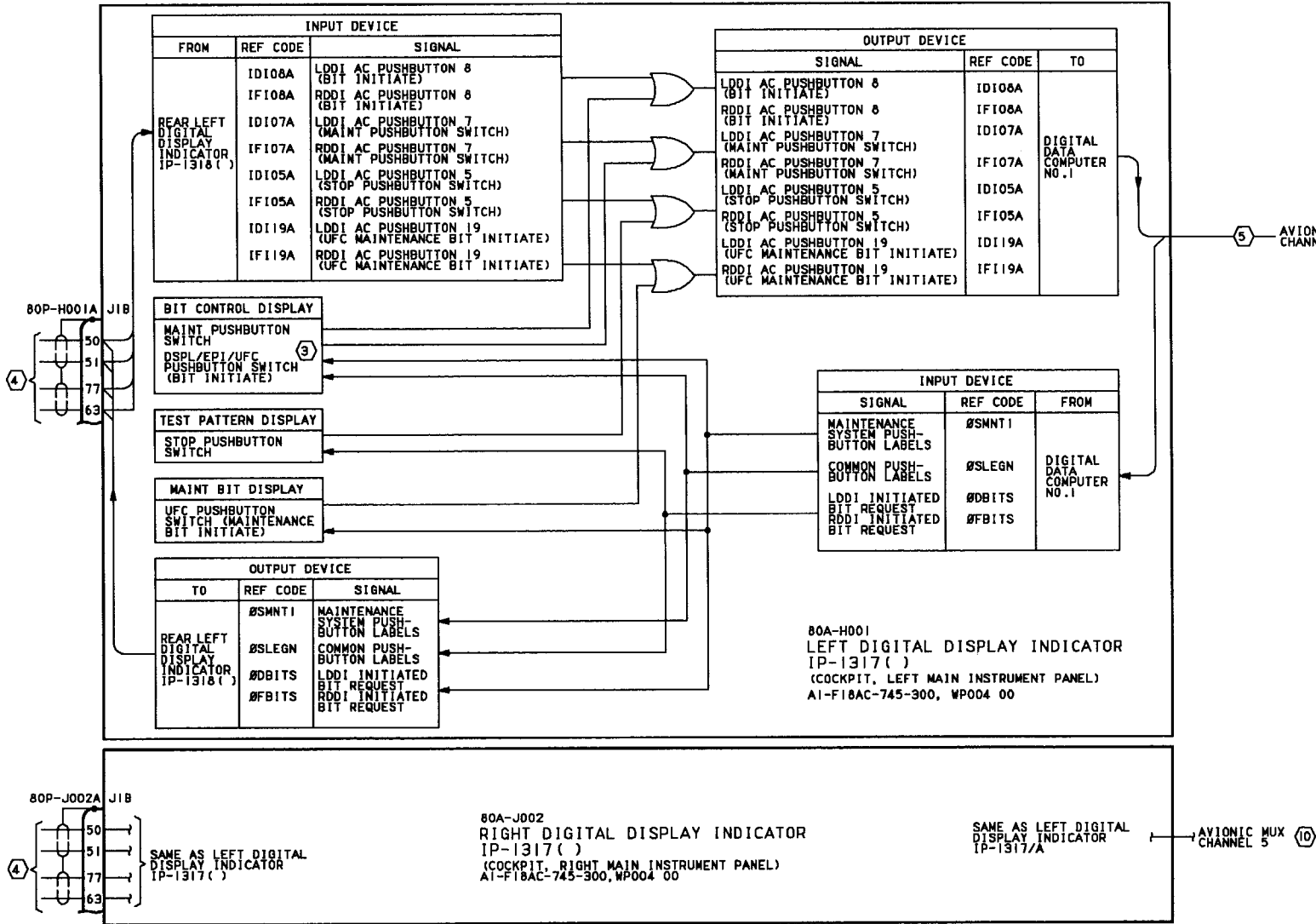


Figure 1.

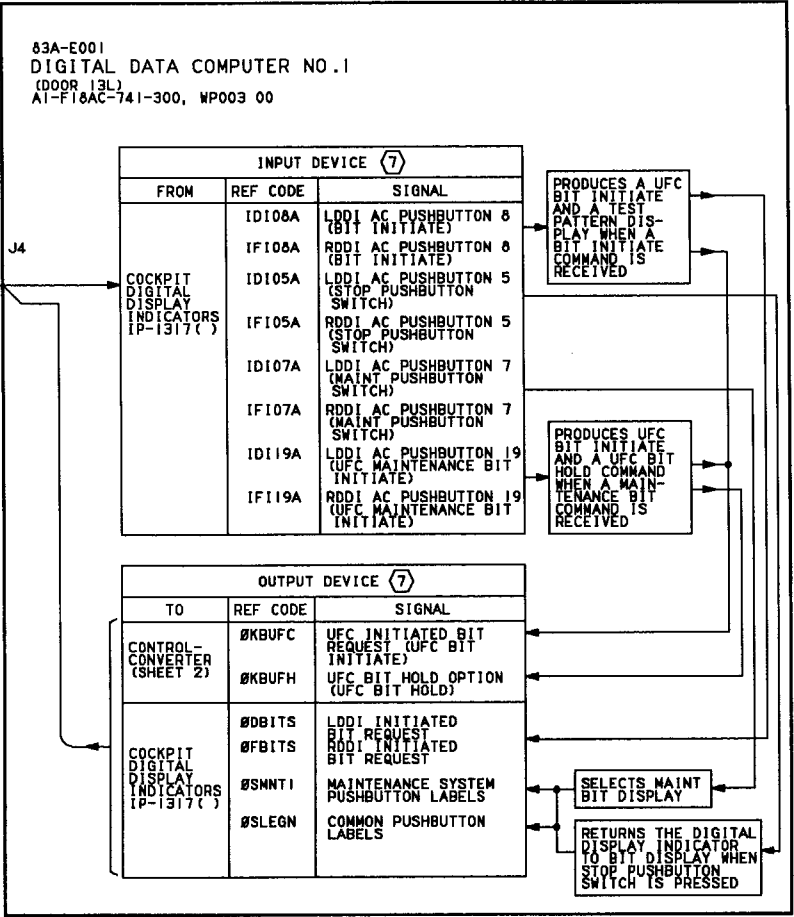


Figure 1. Electronic Equipment Control C-10380/ASQ Built-In Test Schematic (Sheet 1)

Figure 1.

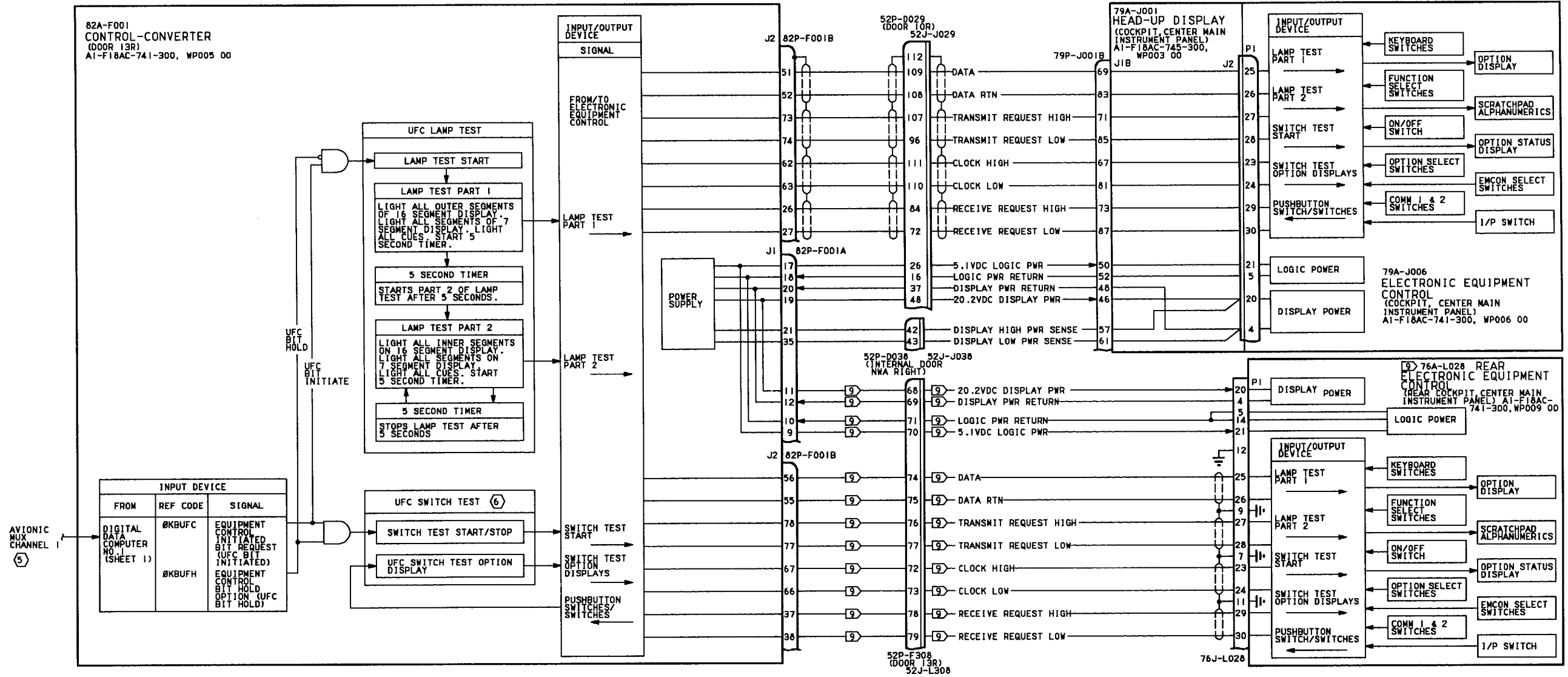


Figure 1.

Figure 1. Electronic Equipment Control C-10380/ASQ Built-In Test Schematic (Sheet 2)

Figure 1.

LEGEND

1. CONTINUITY TESTS:
- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
- (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.
- $\textcircled{3}$ WHEN THE DSPL/EPI/UFC PUSHBUTTON SWITCH IS PRESSED, BIT TESTS ARE RUN SIMULTANEOUSLY ON THE MULTIPURPOSE DISPLAY GROUP, ENGINE MONITOR DISPLAY AND ELECTRONIC EQUIPMENT CONTROL.
- $\textcircled{4}$ MULTIPURPOSE DISPLAY GROUP INTERCONNECT SCHEMATIC, A1-F18AC-745-500, WP004 00.
- $\textcircled{5}$ AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 00.
- $\textcircled{6}$ SWITCH TEST OPTION 1 DISPLAY

PUSHBUTTON SWITCH/SWITCH	CHAR 1	CHAR 2	CHAR 3	CHAR 4
A/P FUNCTION SELECT	A	/	P	
IFF FUNCTION SELECT	I	F	F	
TCN FUNCTION SELECT	T	C	N	
ILS FUNCTION SELECT	I	L	S	
D/L FUNCTION SELECT	D	/	L	
BCN FUNCTION SELECT	B	C	N	

Figure 1.

PUSHBUTTON SWITCH/SWITCH	CHAR 1	CHAR 2	CHAR 3	CHAR 4
ON/OFF	O	N		
EMCON SELECT	E	M	C	N
OPTION 1 SELECT	O	P	T	1
OPTION 2 SELECT	O	P	T	2
OPTION 3 SELECT	O	P	T	3
OPTION 4 SELECT	O	P	T	4
OPTION 5 SELECT	O	P	T	5
COMM 1 CHANNEL SELECTS (PULL)	C	O	M	1
COMM 2 CHANNEL SELECTS (PULL)	C	O	M	2
I/P	I	/	P	
KEYBOARD 0	0			
KEYBOARD 1	1			
KEYBOARD 2	2			
KEYBOARD 3	3			
KEYBOARD 4	4			
KEYBOARD 5	5			
KEYBOARD 6	6			
KEYBOARD 7	7			
KEYBOARD 8	8			
KEYBOARD 9	9			
KEYBOARD CLR	C	L	R	
KEYBOARD ENT	E	N	T	

- $\textcircled{7}$ FOR MEMORY INSPECT ADDRESS LOCATION RELATING TO REF CODE, REFER TO A1-F18AC-FIM-100.
- $\textcircled{8}$ AVIONIC MUX CHANNEL 2 SCHEMATIC, WP005 00.
- $\textcircled{9}$ F/A-18B.
- $\textcircled{10}$ AVIONIC MUX CHANNEL 5 SCHEMATIC, WP018 01.

Figure 1. Electronic Equipment Control C-10380/ASQ Built-In Test Schematic (Sheet 3)

Figure 1.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - ELECTRONIC EQUIPMENT CONTROL C-10380/ASQ BUILT-IN TEST****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292**

Reference Material

None

Alphabetical Index**Subject****Page No.**

Electronic Equipment Control C-10380/ASQ Built-In Test Schematic, Figure 1

2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jan 01	-

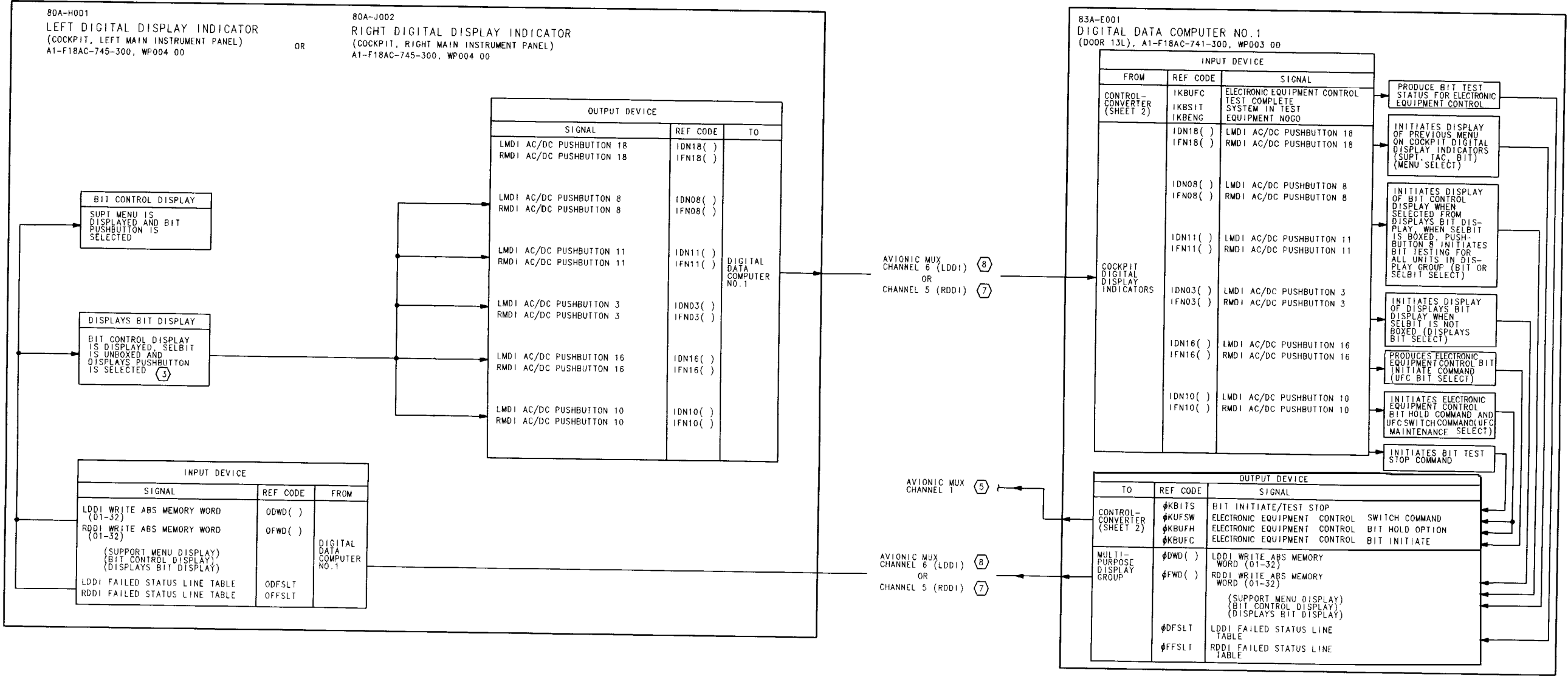


Figure 1. Electronic Equipment Control C-10380/ASQ Built-In Test Schematic (Sheet 1)

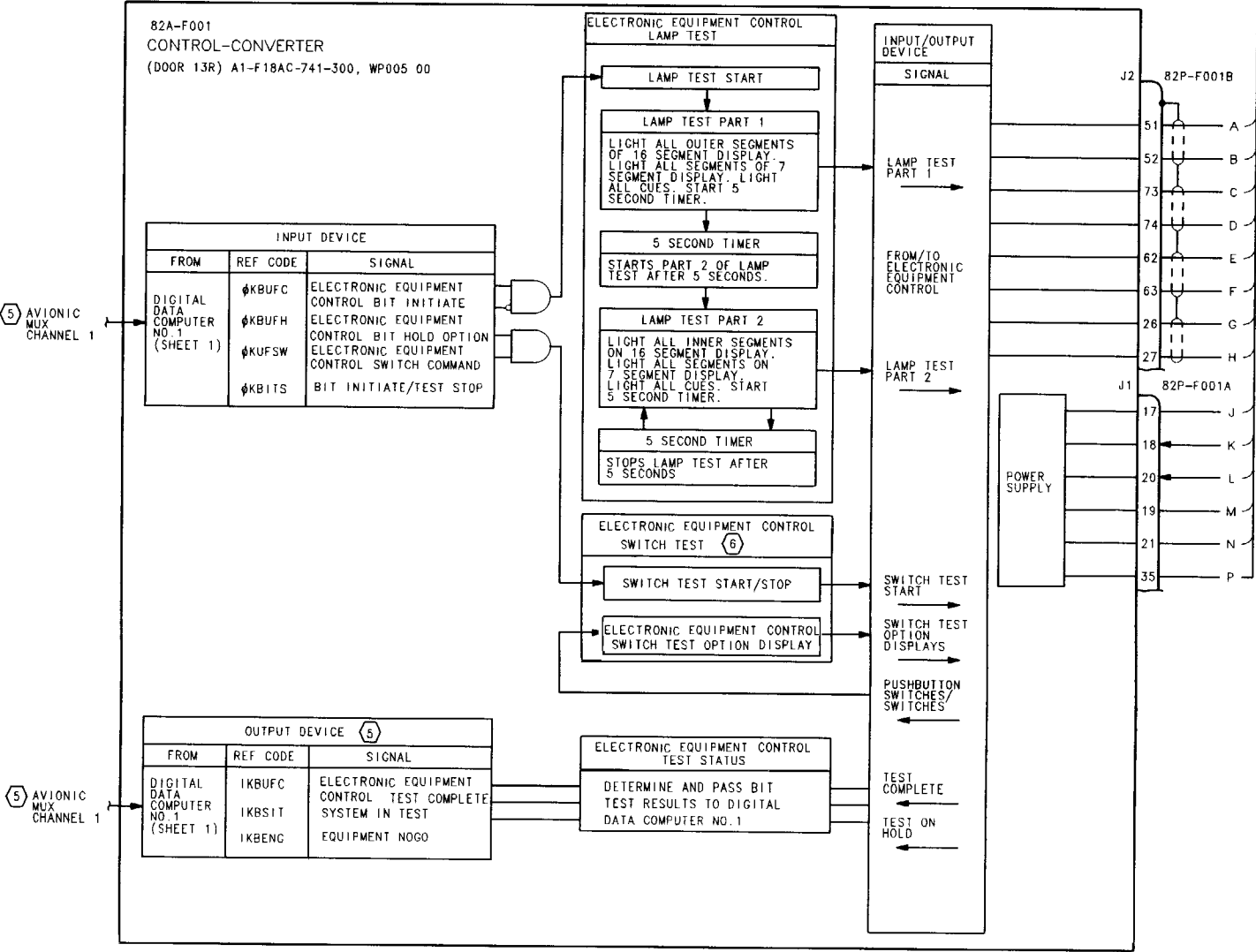


Figure 1.

Figure 1. Electronic Equipment Control C-10380/ASQ Built-In Test Schematic (Sheet 2)

Figure 1.

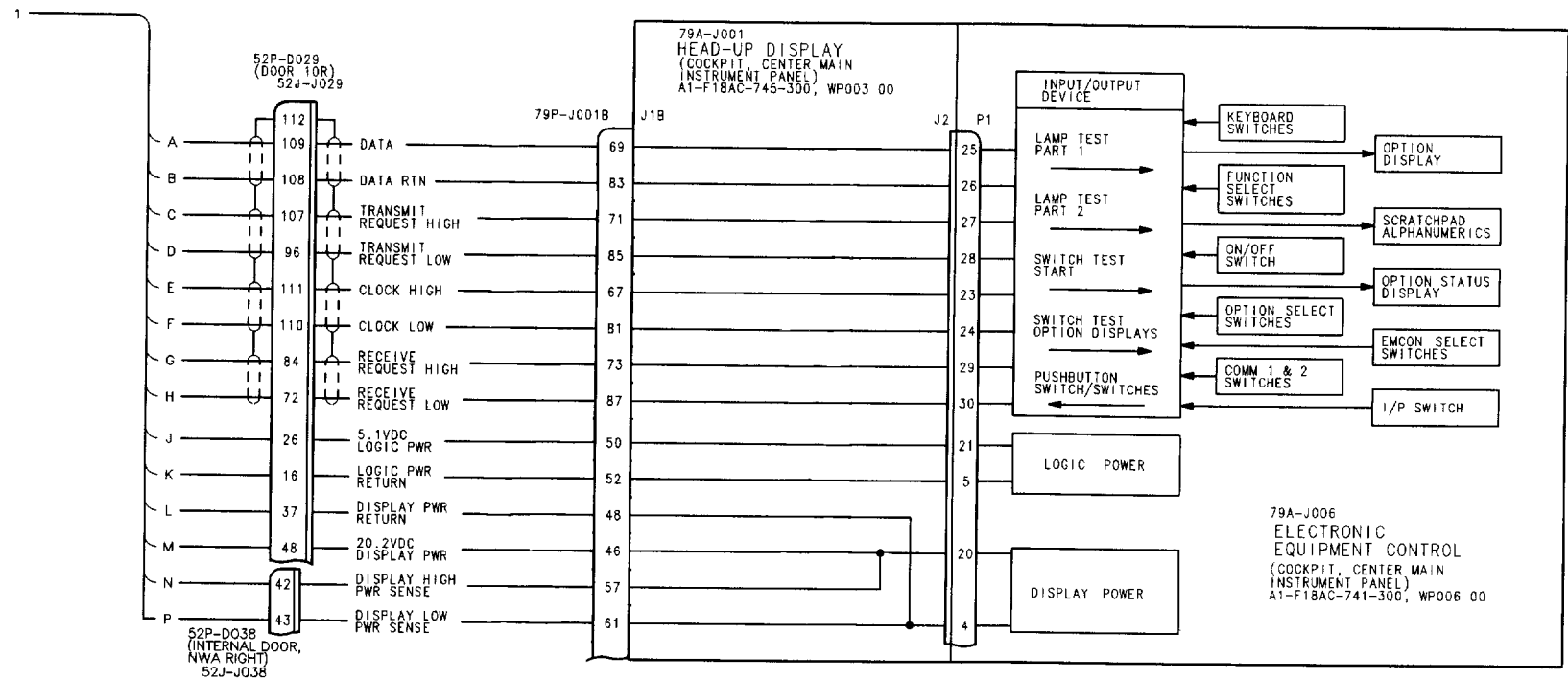


Figure 1.

Figure 1. Electronic Equipment Control C-10380/ASQ Built-In Test Schematic (Sheet 3)

Figure 1.

LEGEND

1. CONTINUITY TESTS:
- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
- (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.
- ③ WHEN THE DSPL/EPI/UFC PUSHBUTTON SWITCH IS PRESSED, BIT TESTS ARE RUN SIMULTANEOUSLY ON THE MULTIPURPOSE DISPLAY GROUP, ENGINE MONITOR DISPLAY AND ELECTRONIC EQUIPMENT CONTROL.
- ④ MULTIPURPOSE DISPLAY GROUP INTERCONNECT SCHEMATIC, A1-F18AC-745-500, WP004 00.
- ⑤ AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 00.
- ⑥ SWITCH TEST OPTION 1 DISPLAY

PUSHBUTTON SWITCH/SWITCH	CHAR 1	CHAR 2	CHAR 3	CHAR 4
A/P FUNCTION SELECT	A	/	P	
IFF FUNCTION SELECT	I	F	F	
TCN FUNCTION SELECT	T	C	N	
ILS FUNCTION SELECT	I	L	S	
D/L FUNCTION SELECT	D	/	L	
BCN FUNCTION SELECT	B	C	N	

Figure 1.

Figure 1. Electronic Equipment Control C-10380/ASQ Built-In Test Schematic (Sheet 4)

PUSHBUTTON SWITCH/SWITCH	CHAR 1	CHAR 2	CHAR 3	CHAR 4
ON/OFF	O	N		
EMCON SELECT	E	M	C	N
OPTION 1 SELECT	O	P	T	1
OPTION 2 SELECT	O	P	T	2
OPTION 3 SELECT	O	P	T	3
OPTION 4 SELECT	O	P	T	4
OPTION 5 SELECT	O	P	T	5
COMM 1 CHANNEL SELECTS (PULL)	C	O	M	1
COMM 2 CHANNEL SELECTS (PULL)	C	O	M	2
I/P	I	/	P	
KEYBOARD 0	0			
KEYBOARD 1	1			
KEYBOARD 2	2			
KEYBOARD 3	3			
KEYBOARD 4	4			
KEYBOARD 5	5			
KEYBOARD 6	6			
KEYBOARD 7	7			
KEYBOARD 8	8			
KEYBOARD 9	9			
KEYBOARD CLR	C	L	R	
KEYBOARD ENT	E	N	T	

- ⑦ AVIONIC MUX CHANNEL 5 SCHEMATIC, WP018 00.
- ⑧ AVIONIC MUX CHANNEL 6 SCHEMATIC, WP019 00.

Figure 1.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - EMCON

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A BEFORE F/A-18 AFC 253 OR F/A-18 AFC 292 AND F/A-18B

This WP supersedes WP 012 00, dated 1 June 1992.

Reference Material

None

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Record of Applicable Technical Directives

None

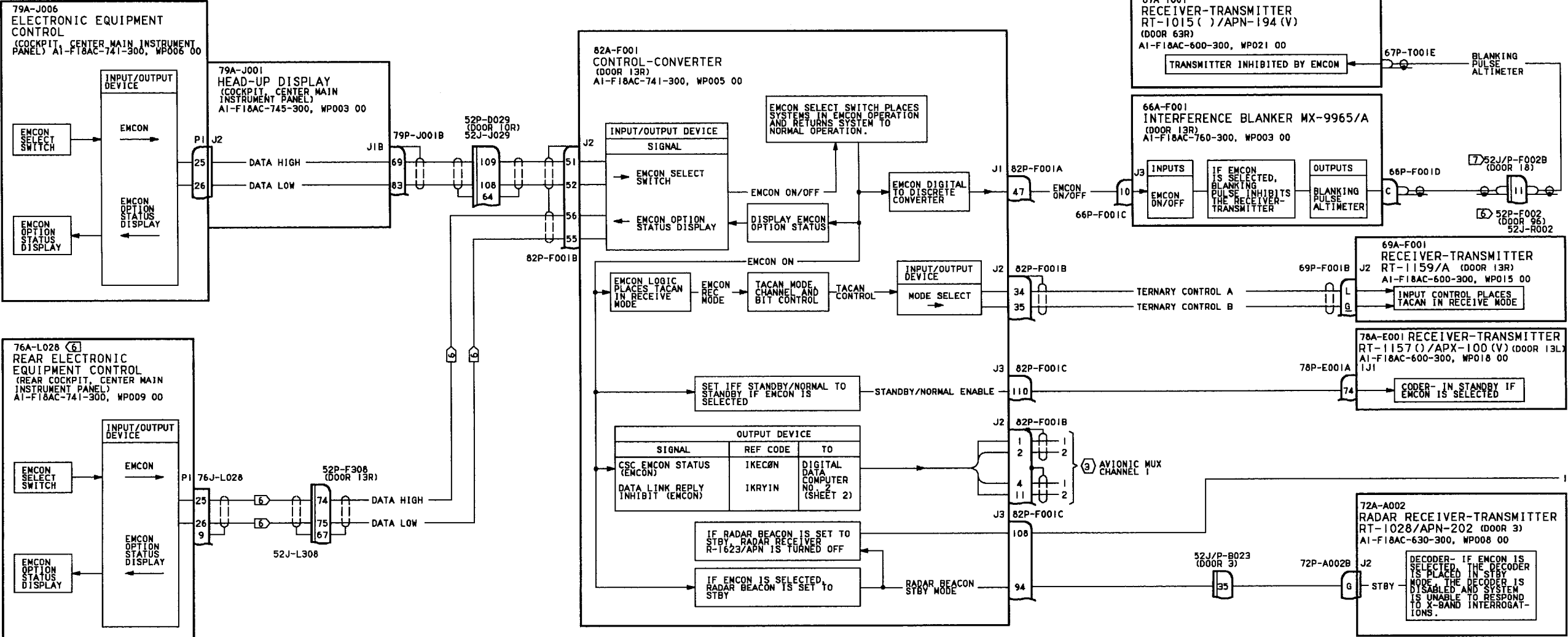


Figure 1.

Figure 1. EMCON Schematic (Sheet 1)

Figure 1.

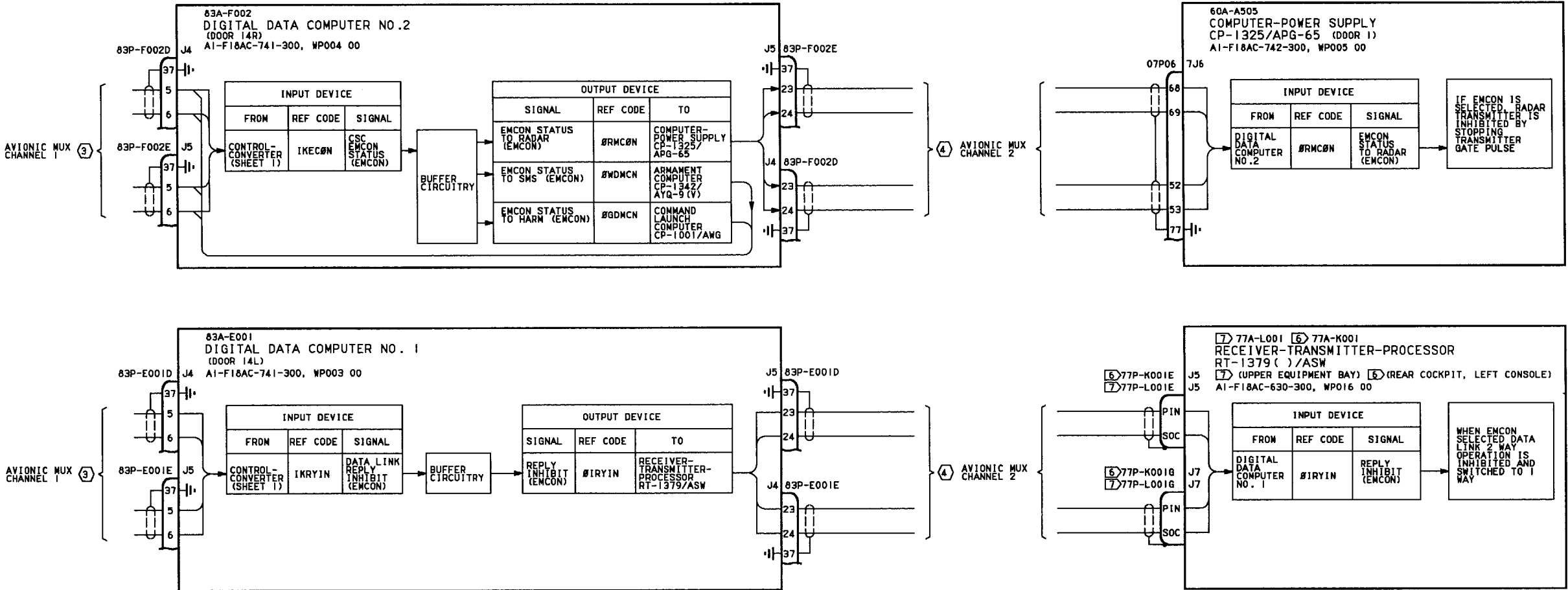


Figure 1.

Figure 1. EMCON Schematic (Sheet 2)

Figure 1.

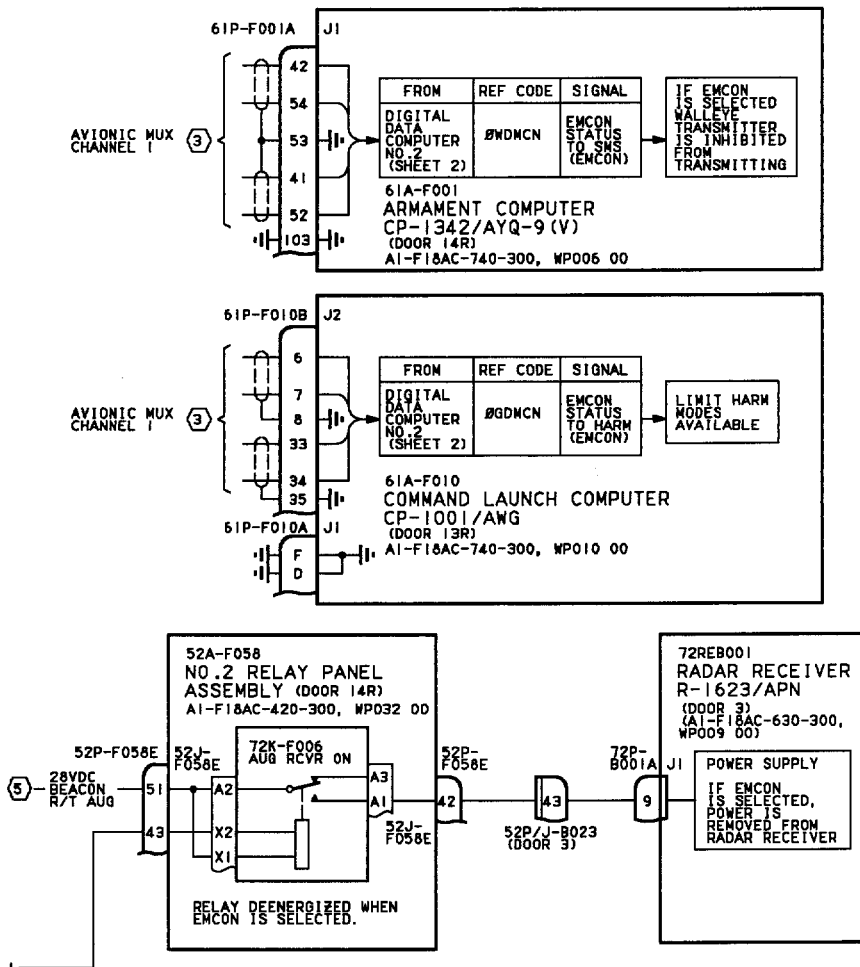


Figure 1. EMCON Schematic (Sheet 3)

LEGEND

1. CONTINUITY TESTS:

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

- ③ AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 00.
- ④ AVIONIC MUX CHANNEL 2 SCHEMATIC, WP005 00.
- ⑤ RADAR BEACON SYSTEM SCHEMATIC, A1-F18AC-630-500, WP006 00.
- 6 F/A-18B.
- 7 F/A-18A.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - EMCON

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

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Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jan 01	-

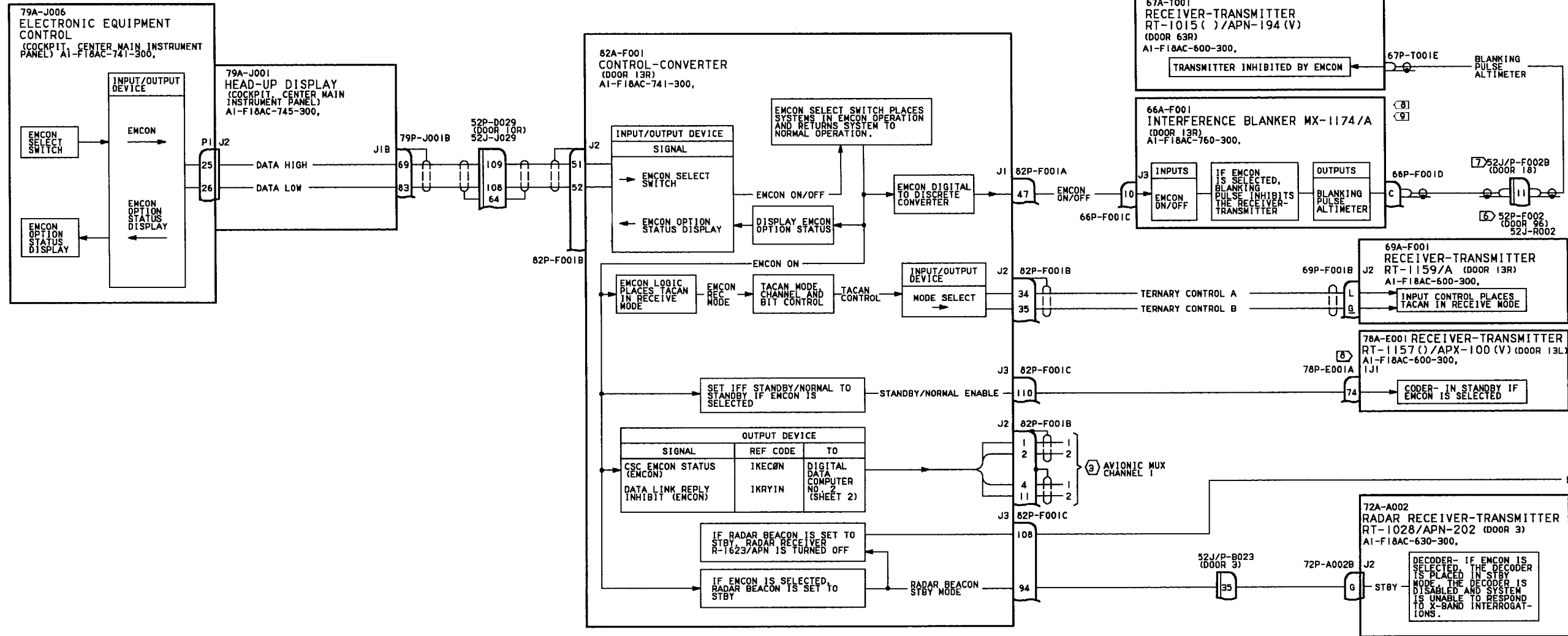


Figure 1. EMCON Schematic (Sheet 1)

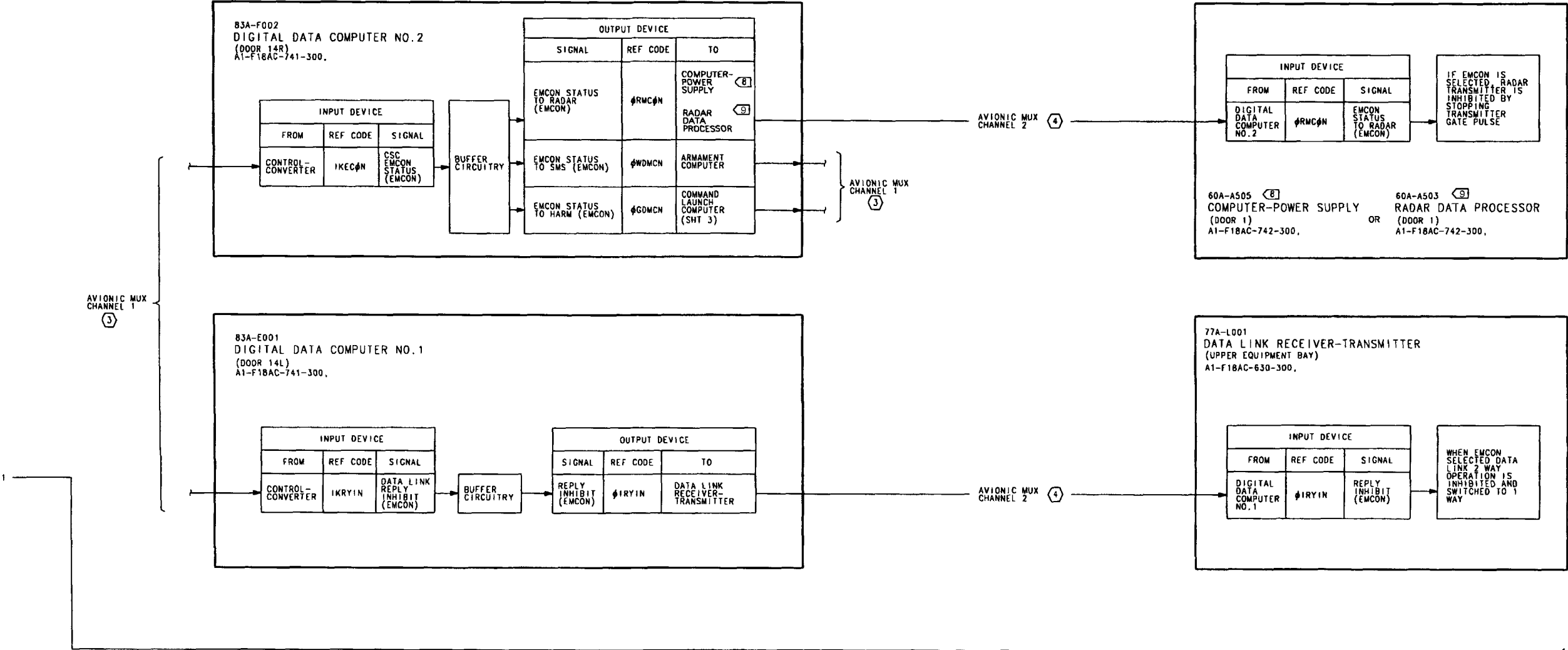


Figure 1. EMCON Schematic (Sheet 2)

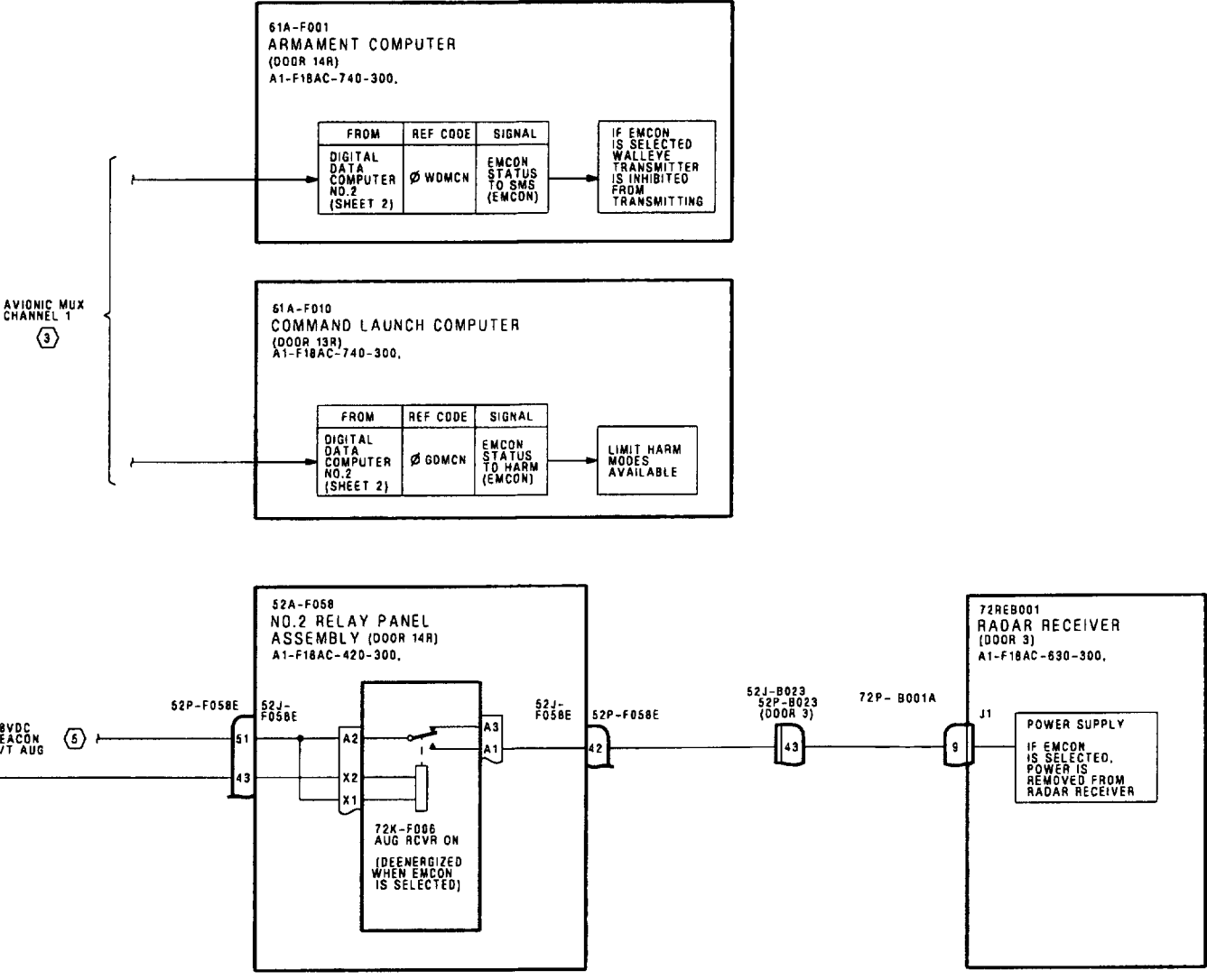


Figure 1.

Figure 1. EMCON Schematic (Sheet 3)

Figure 1.

LEGEND**1. CONTINUITY TESTS:**

A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.

B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.

C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.

D. WHEN TESTING CONTINUITY, TEST FOR:

- (1) SHORTS TO GROUND.
- (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
- (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
- (4) SHIELD CONTINUITY.

E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.


 AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 01.


 AVIONIC MUX CHANNEL 2 SCHEMATIC, WP005 01.

 RADAR BEACON SYSTEM SCHEMATIC, A1-F18AC-630-500, WP006 00.

 DELETED.

 DELETED.

 AFTER F/A-18A AFC 253.

 AFTER F/A-18A AFC 292.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - DIGITAL DATA COMPUTER NO. 1 AND NO. 2
CAUTIONS, ADVISORY AND MAINTENANCE CODES****MISSION COMPUTER SYSTEM****This WP supersedes WP013 00, dated 1 January 2001.**

Title	Work Package
Schematic - Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes (F/A-18A/B)	013 01
Schematic - Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes (AFTER F/A-18 AFC 225 AND F/A-18 AFC 231)	013 02
Schematic - Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes (AFTER F/A-18 AFC 253 OR F/A-18 AFC 292; AND AFTER AFC 231 PART 2 OR PART 3)	013 03

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - DIGITAL DATA COMPUTER NO. 1 AND NO. 2
CAUTIONS, ADVISORY AND MAINTENANCE CODES****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A/B****This WP supersedes WP013 01, dated 1 January 2001.****Reference Material**

None

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Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance

Codes Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 20	-	Add Provisions for Electronic Boresight Radar (ECP MDA-F18-00050C1)	1 Jun 92	-
F/A-18 AFC 39	-	No. 1 Fuel Tank Interconnect Valve Replacement and Fuel Sequencing Modification (ECP MDA-F18-00072C1)	1 Jun 92	-
F/A-18 AFC 41	-	Throttle Sensitivity Improvement (ECP MDA-F/A-18-00054)	1 Jun 92	-
F/A-18 AFC 90	-	GFE Battery Relay Control Unit Incorporation (ECP MDA-F/A-18-00165R1)	1 Jun 92	-

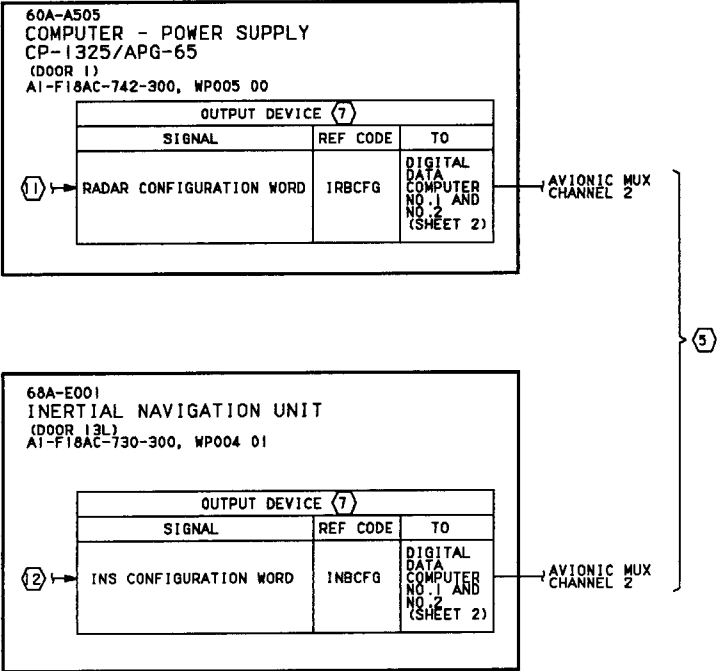


Figure 1.

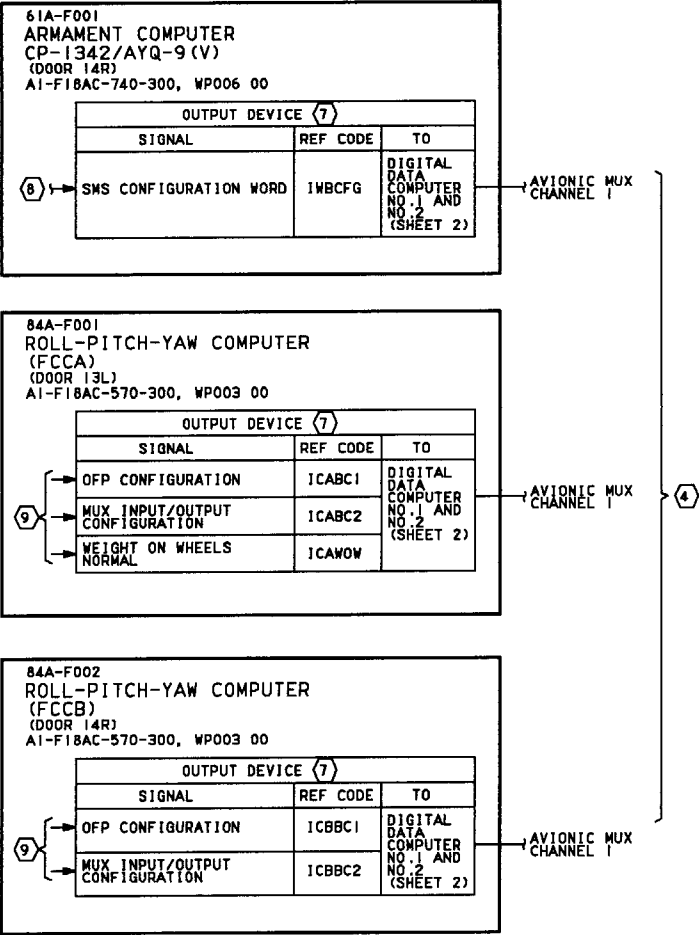


Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 1)

Figure 1.

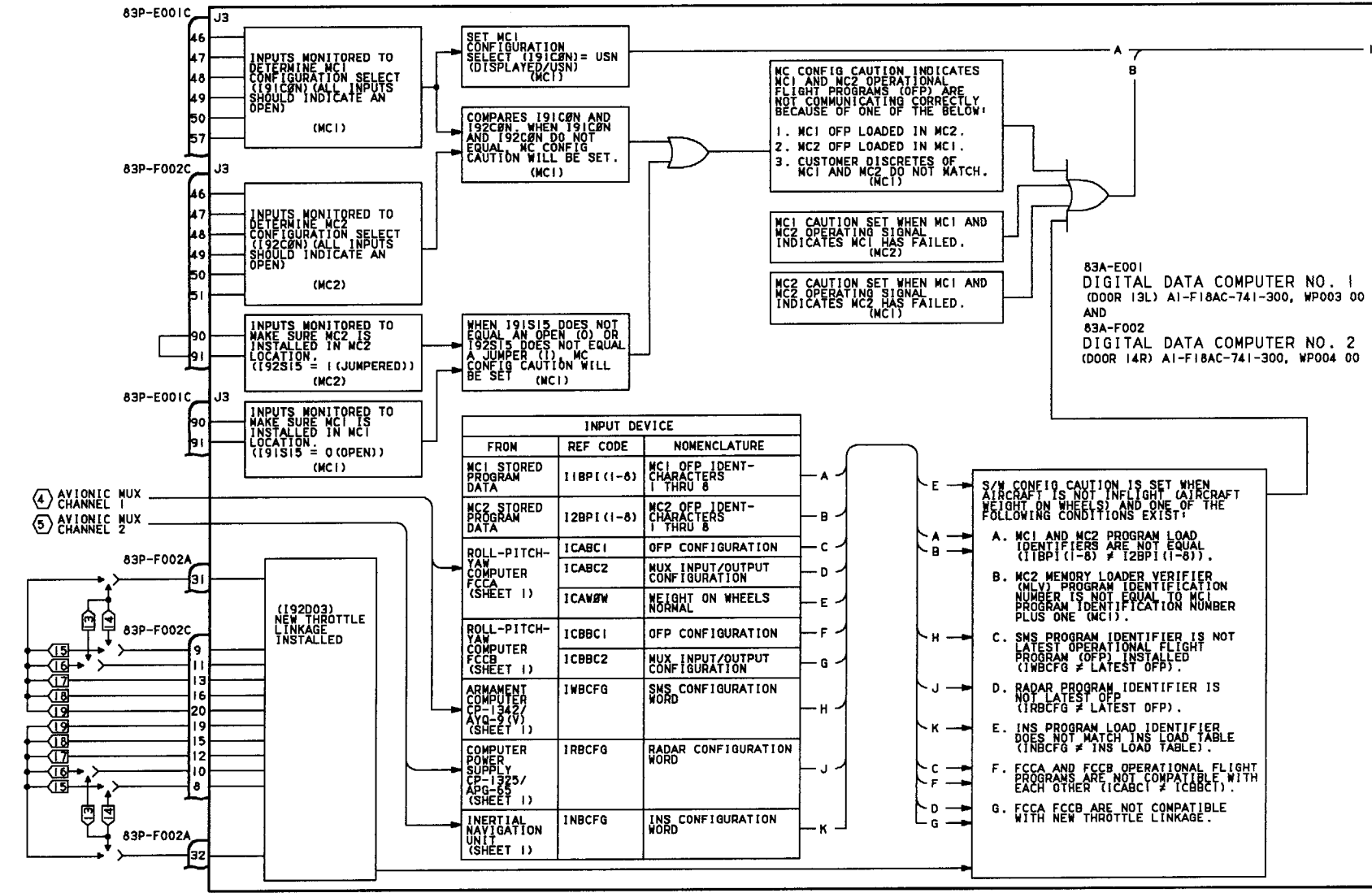


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 2)

Figure 1.

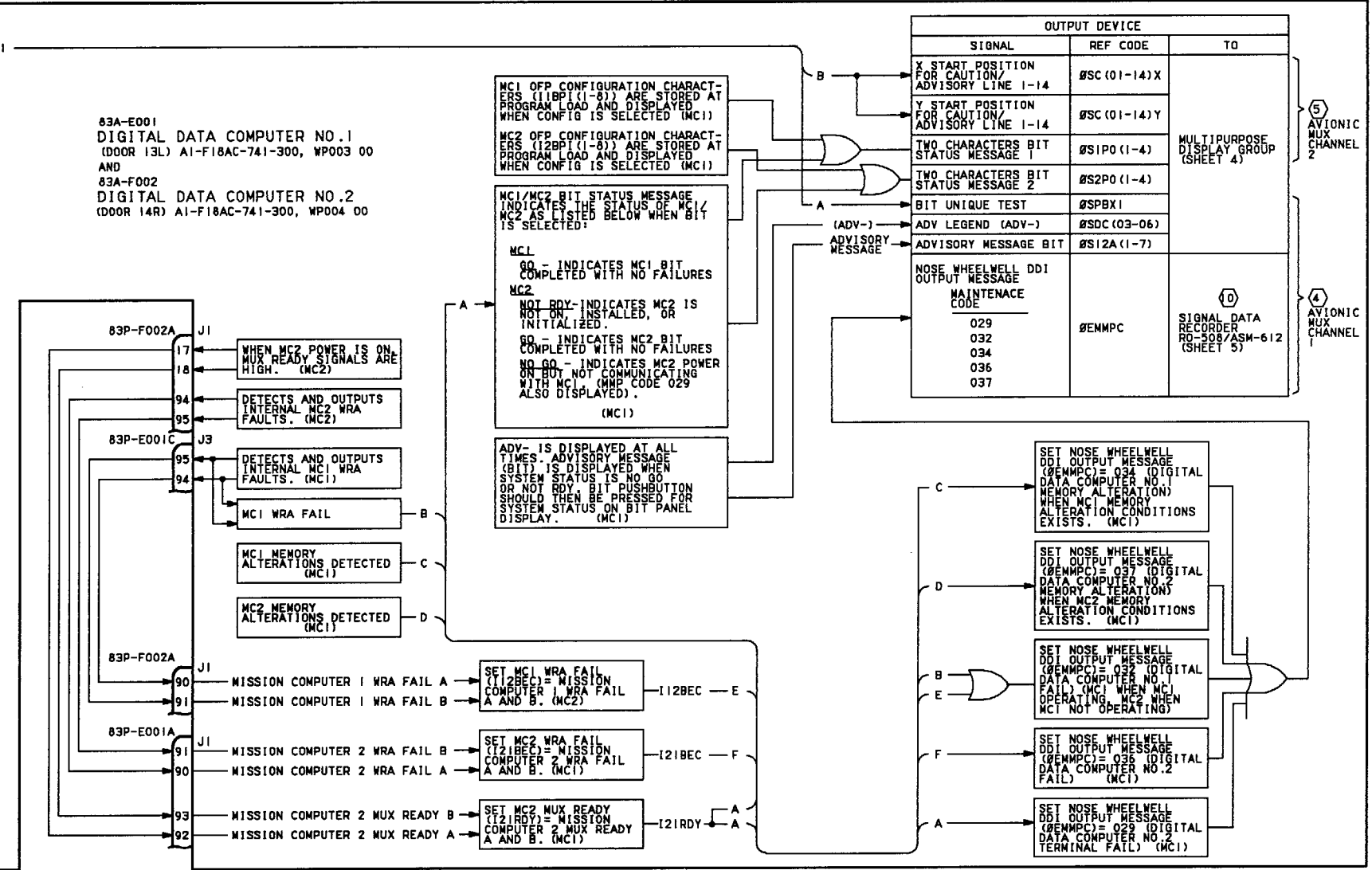


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 3)

Figure 1.

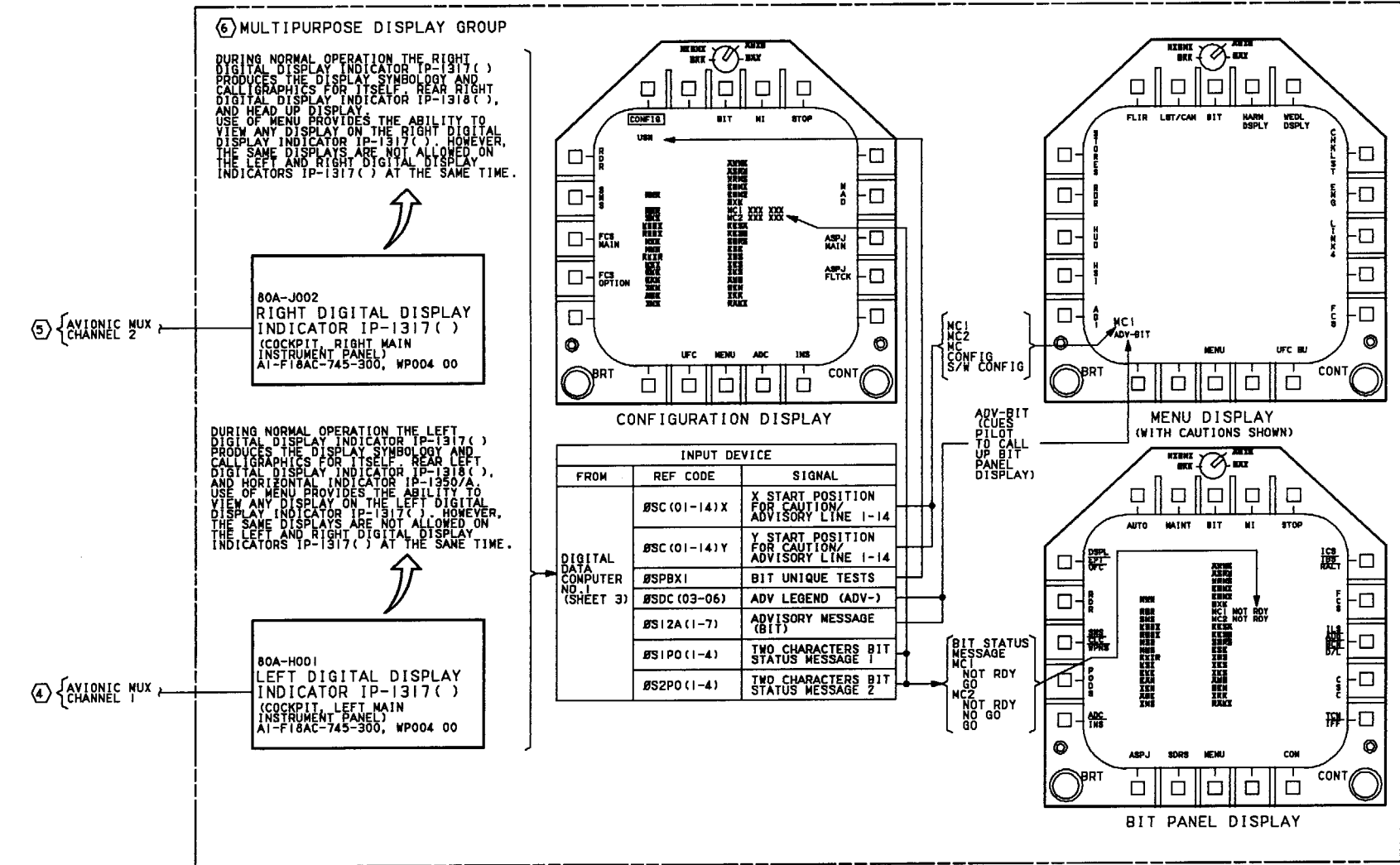


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 4)

Figure 1.

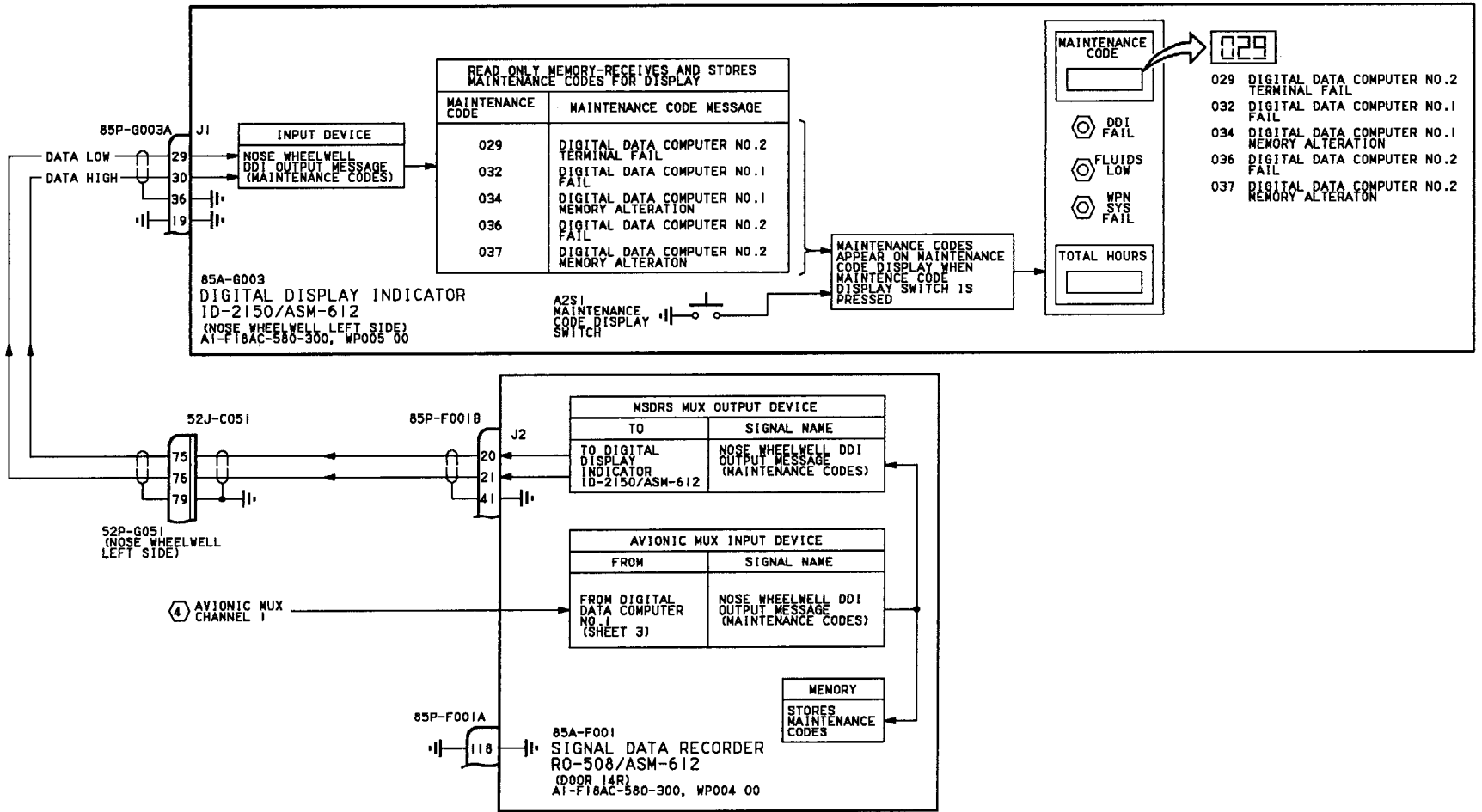


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Caution, Advisory and Maintenance Codes Schematic (Sheet 5)

Figure 1.

LEGEND

1. CONTINUITY TESTS:
- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
- (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.
3. WHEN BOTH DIGITAL DATA COMPUTER NO. 1 AND NO. 2 ARE OPERATING COMPUTER FAILURE DATA IS DETERMINED USING DIGITAL DATA COMPUTER INTERCONNECTION DATA (WP 008 00) AND AVIONIC MUX CHANNEL 3 DATA (WP 006 00). DIGITAL DATA COMPUTER NO. 1 PROVIDES ALL BIT STATUS, CAUTION, ADVISORY, AND NOSE WHEELWELL DDI CODES USING DATA COLLECTED BY BOTH COMPUTERS. DIGITAL DATA COMPUTER NO. 2 PROVIDES MC1 CAUTION AND NOSE WHEELWELL DDI CODES ONLY WHEN DIGITAL DATA COMPUTER NO. 1 IS NOT OPERATING. EACH FUNCTION BLOCK ON THIS DIAGRAM NOTES THE COMPUTER IN WHICH THAT FUNCTION IS MONITORED OR PROCESSED (DIGITAL DATA COMPUTER NO. 1 OR NO. 2).
- (4) AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 00.
- (5) AVIONIC MUX CHANNEL 2 SCHEMATIC, WP005 00.
- (6) THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD UP DISPLAY, HORIZONTAL INDICATOR IP-1350/A, ALSO ON F/A-18B; REAR LEFT DIGITAL DISPLAY INDICATOR IP-1918(), REAR RIGHT DIGITAL DISPLAY INDICATOR IP-1318(), AND REAR CENTER DIGITAL DISPLAY INDICATOR IP-1317(). FOR MULTIPURPOSE DISPLAY GROUP SCHEMATIC, REFER TO A1-F18AC-745-500, WP001 00.
- (7) FOR LOGIC DIAGRAMS RELATING TO REF CODE, REFER TO A1-F18A()-OLD-000. FOR MEMORY INSPECT ADDRESS LOCATION RELATING TO REF CODE, REFER TO A1-F18AC-FIM-100.
- (8) BUILT-IN TEST AVIONIC INTERFACE SCHEMATIC, A1-F18AC-740-500, WP022 00.
- (9) CROSS CHANNEL/MUX BUS/DISPLAYS FUNCTIONAL SCHEMATIC, A1-F18AC-570-500, WP021 01.
- (10) FAULT REPORTING INTEGRATION SCHEMATIC, WP016 00.
- (11) OPERATING STATUS SELECT AND DISPLAY SCHEMATIC, A1-F18AC-742-500, WP008 00.
- (12) INS GYRO/GB CONTROL SYSTEM, A1-F18AC-730-500, WP011 00.
- (13) 161520 THRU 161528 BEFORE F/A-18 AFC 20.
- (14) 161353 THRU 161519 AFTER F/A-18 AFC 20 BUT BEFORE F/A-18 AFC 41.
- (15) 161702 AND UP; ALSO 161353 THRU 161519 AFTER F/A-18 AFC 41 AND 181520 THRU 161528 AFTER F/A-18 AFC 20.
- (16) 161702 AND UP; ALSO 161353 THRU 161513 AFTER F/A-18 AFC 33 AND 161520 THRU 161528 AFTER F/A-18 AFC 20.
- (17) 161702 AND UP.
- (18) 161702 AND UP; ALSO 161353 THRU 161528 AFTER F/A-18 AFC 41.
- (19) 163119 AND UP; ALSO 161353 THRU 163118 AFTER F/A-18 AFC 90.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - DIGITAL DATA COMPUTER NO. 1 AND NO. 2
CAUTIONS, ADVISORY AND MAINTENANCE CODES

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A AND F/A-18B AFTER F/A-18 AFC 225 AND F/A-18 AFC 231

Reference Material

None

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Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance

Codes Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 20	-	Add Provisions for Electronic Boresight Radar (ECP MDA-F18-00050C1)	1 Jun 92	-
F/A-18 AFC 39	-	No. 1 Fuel Tank Interconnect Valve Replacement and Fuel Sequencing Modification (ECP MDA-F18-00072C1)	1 Jun 92	-
F/A-18 AFC 41	-	Throttle Sensitivity Improvement (ECP MDA-F/A-18-00054)	1 Jun 92	-
F/A-18 AFC 90	-	GFE Battery Relay Control Unit Incorporation (ECP MDA-F/A-18-00165R1)	1 Jun 92	-
F/A-18 AFC 225	-	Five (5) Avionics Multiplex Bus Upgrade, Incor- poration of (ECP MDA-F/A-18 0529)	1 Jun 02	-
F/A-18 AFC 231	-	Embedded Global Positioning System (GPS)/In- ertial Navigation System (INS) (EGI), Incorpora- tion of (ECP MDA-F/A-18 0521)	1 Jun 02	-

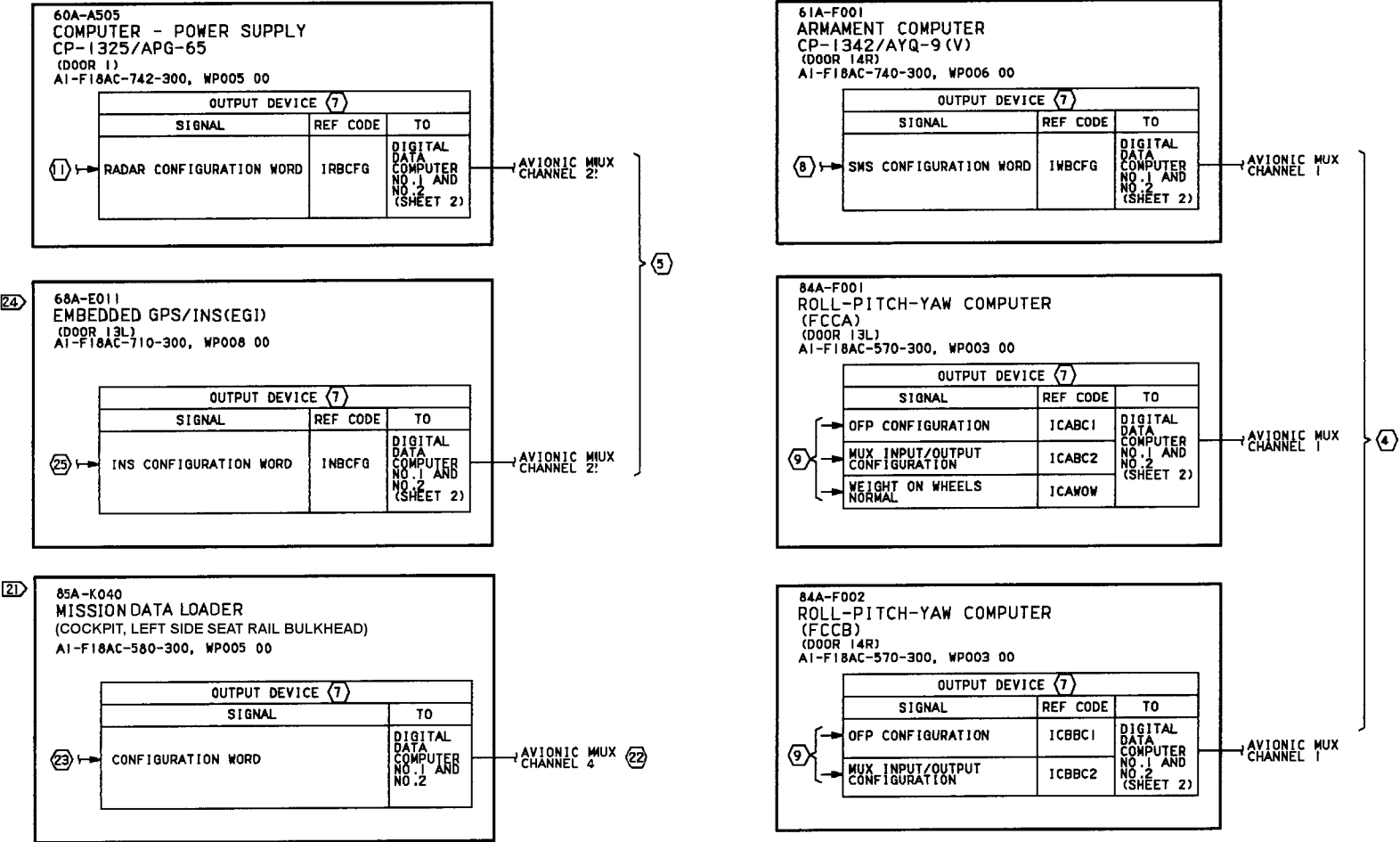


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 1)

Figure 1.

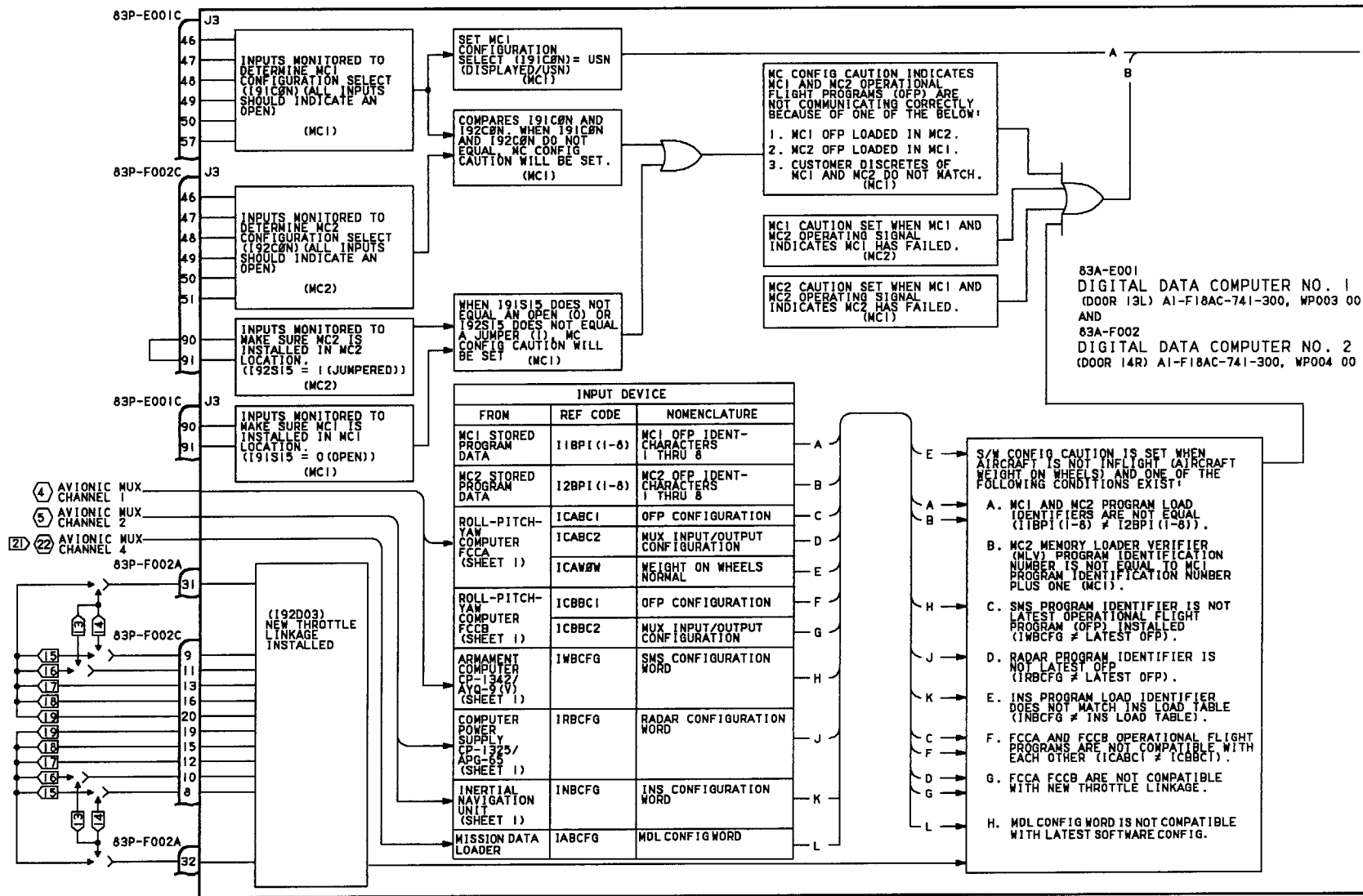


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 2)

Figure 1.

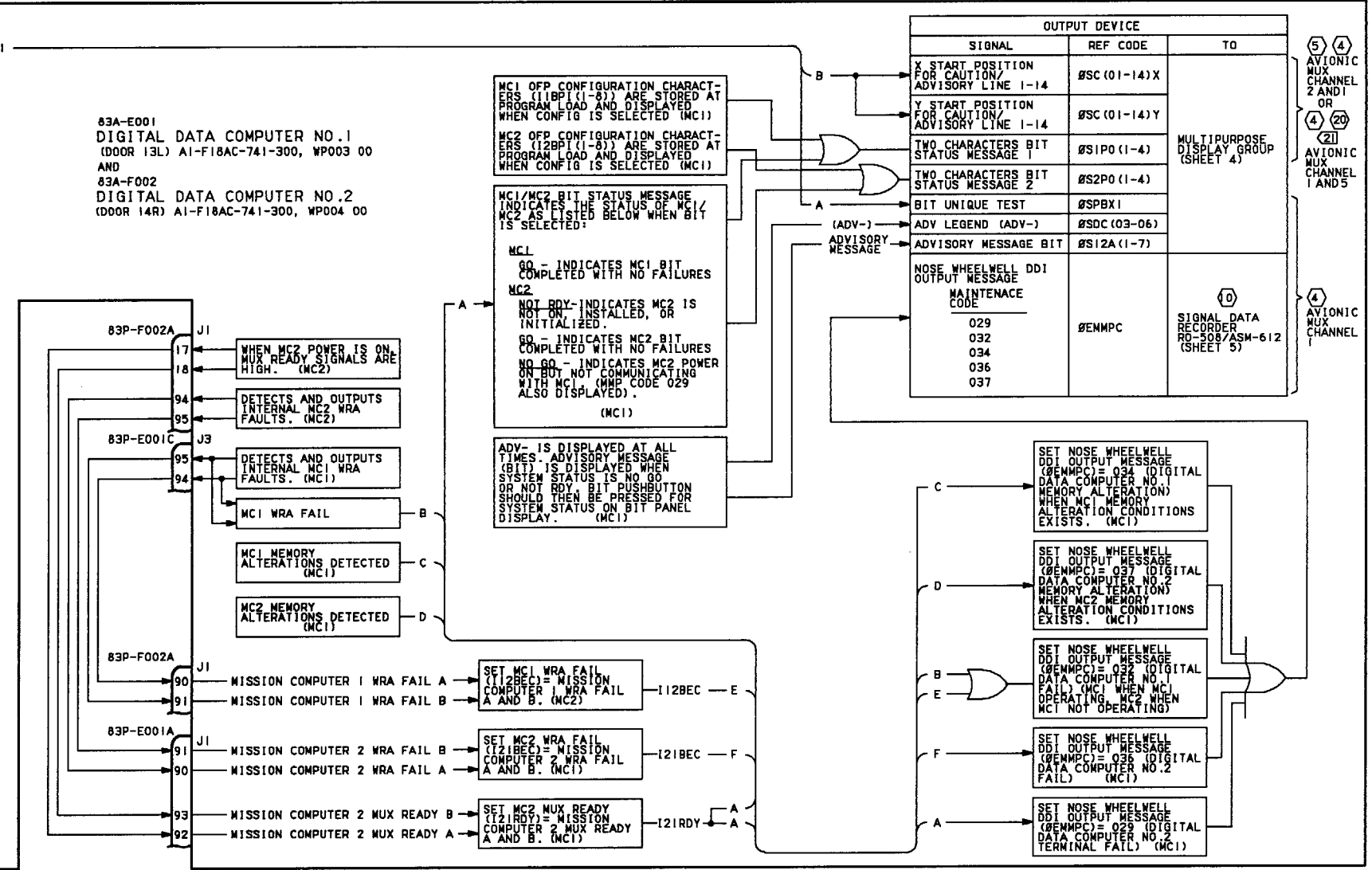


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 3)

Figure 1.

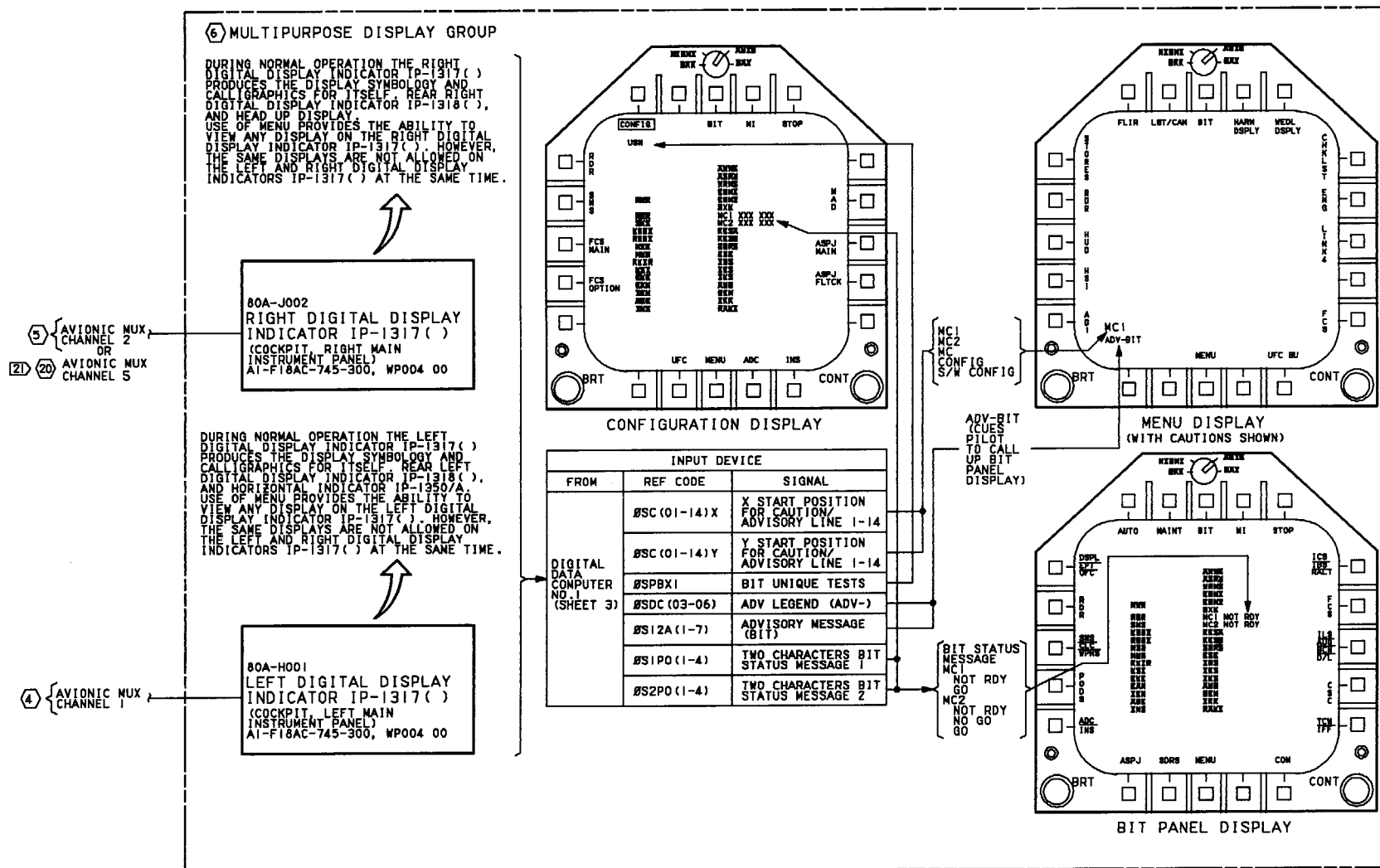


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 4)

Figure 1.

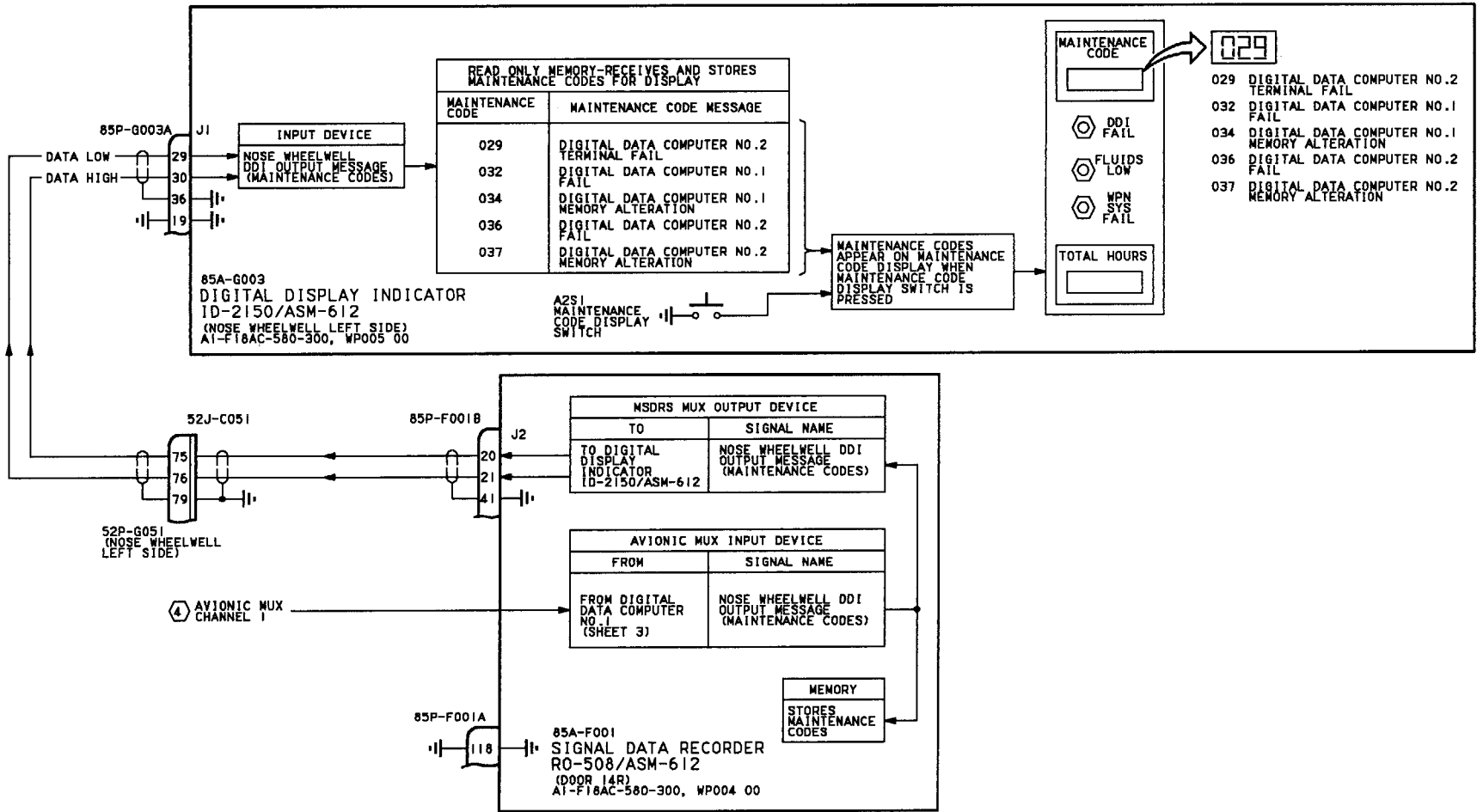


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Caution, Advisory and Maintenance Codes Schematic (Sheet 5)

Figure 1.

LEGEND

1. CONTINUITY TESTS:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
 - B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
 - C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
 - D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
 - E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
 2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.
 3. WHEN BOTH DIGITAL DATA COMPUTER NO. 1 AND NO. 2 ARE OPERATING COMPUTER FAILURE DATA IS DETERMINED USING DIGITAL DATA COMPUTER INTERCONNECTION DATA (WP 008 00) AND AVIONIC MUX CHANNEL 3 DATA (WP 006 00). DIGITAL DATA COMPUTER NO. 1 PROVIDES ALL BIT STATUS, CAUTION, ADVISORY, AND NOSE WHEELWELL DDI CODES USING DATA COLLECTED BY BOTH COMPUTERS. DIGITAL DATA COMPUTER NO. 2 PROVIDES MC1 CAUTION AND NOSE WHEELWELL DDI CODES ONLY WHEN DIGITAL DATA COMPUTER NO. 1 IS NOT OPERATING. EACH FUNCTION BLOCK ON THIS DIAGRAM NOTES THE COMPUTER IN WHICH THAT FUNCTION IS MONITORED OR PROCESSED (DIGITAL DATA COMPUTER NO. 1 OR NO. 2).
- ④ AVIONIC MUX CHANNEL 1 SCHEMATIC, WP004 00.
 - ⑤ AVIONIC MUX CHANNEL 2 SCHEMATIC, WP005 00.
 - ⑥ THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD UP DISPLAY, HORIZONTAL INDICATOR IP-1350/A, ALSO ON F/A-18B; REAR LEFT DIGITAL DISPLAY INDICATOR IP-1918(), REAR RIGHT DIGITAL DISPLAY INDICATOR IP-1318(), AND REAR CENTER DIGITAL DISPLAY INDICATOR IP-1317(). FOR MULTIPURPOSE DISPLAY GROUP SCHEMATIC, REFER TO A1-F18AC-745-500, WP001 00.
 - ⑦ FOR MEMORY INSPECT ADDRESS LOCATION RELATING TO REF CODE, REFER TO A1-F18AC-FIM-100.
 - ⑧ BUILT-IN TEST AVIONIC INTERFACE SCHEMATIC, A1-F18AC-740-500, WP022 00.
 - ⑨ CROSS CHANNEL/MUX BUS/DISPLAYS FUNCTIONAL SCHEMATIC, A1-F18AC-570-500, WP021 01.
 - ⑩ FAULT REPORTING INTEGRATION SCHEMATIC, WP016 00.
 - ⑪ OPERATING STATUS SELECT AND DISPLAY SCHEMATIC, A1-F18AC-742-500, WP008 00.
 - ⑫ INS GYRO/GB CONTROL SYSTEM, A1-F18AC-730-500, WP011 00.
 - ⑬ 161520 THRU 161528 BEFORE F/A-18 AFC 20.
 - ⑭ 161353 THRU 161519 AFTER F/A-18 AFC 20 BUT BEFORE F/A-18 AFC 41.
 - ⑮ 161702 AND UP; ALSO 161353 THRU 161519 AFTER F/A-18 AFC 41 AND 161520 THRU 161528 AFTER F/A-18 AFC 20.
 - ⑯ 161702 AND UP; ALSO 161353 THRU 161513 AFTER F/A-18 AFC 33 AND 161520 THRU 161528 AFTER F/A-18 AFC 20.
 - ⑰ 161702 AND UP.
 - ⑱ 161702 AND UP; ALSO 161353 THRU 161528 AFTER F/A-18 AFC 41.
 - ⑲ 163119 AND UP; ALSO 161353 THRU 163118 AFTER F/A-18 AFC 90.
 - ⑳ AVIONIC MUX CHANNEL 5 SCHEMATIC, WP018 00.
 - ㉑ F/A-18A AND F/A-18B AFTER F/A-18 AFC 225.
 - ㉒ AVIONIC MUX CHANNEL 4 SCHEMATIC, WP017 00.
 - ㉓ MISSION DATA LOADER FUNCTIONAL SCHEMATIC, A1-F18AC-580-500.
 - ㉔ F/A-18A AND F/A-18B AFTER F/A-18 AFC 231.
 - ㉕ TEST CONTROL SCHEMATIC, A1-F18AC-710-500, WP012 00.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - DIGITAL DATA COMPUTER NO. 1 AND NO. 2
CAUTIONS, ADVISORY AND MAINTENANCE CODES****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18
AFC 292 AND AFTER F/A-18 AFC 231 PART 2 OR 3****Reference Material**

None

Alphabetical Index**Subject****Page No.**

Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance

Codes Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	U.S. Naval Reserves A+ Avionics Upgrade, Incorporation of (ECP MDA-F/A-18 0560R1)	-	-
F/A-18 AFC 292	-	U.S. Marine Corps Reserves A+ Avionics Upgrade, Incorporation of (ECP MDA-F/A-18 0583)	-	-
F/A-18 AFC 253	-	U.S. Naval Reserves A+ Avionics Upgrade, In- corporation of (ECP MDA-F/A-18 0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	U.S. Marine Corps Reserves A+ Avionics Upgrade, Incorporation of (ECP MDA-F/A-18 0583)	1 Jan 01	-
F/A-18 AFC 231 Part 2 or Part 3	-	Embedded Global Positioning System (GPS)/In- ertial Navigation System (INS) (EGI), Incorpora- tion of (ECP MDA-F/A-18 0521)	1 Jun 02	-

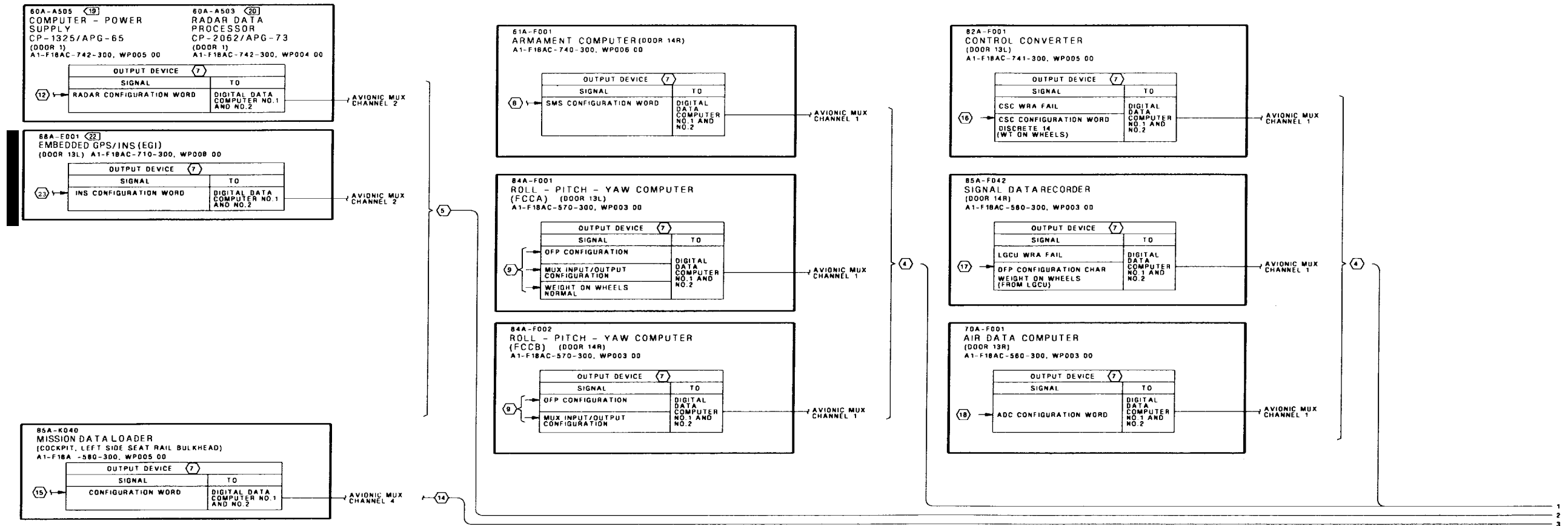


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 1)

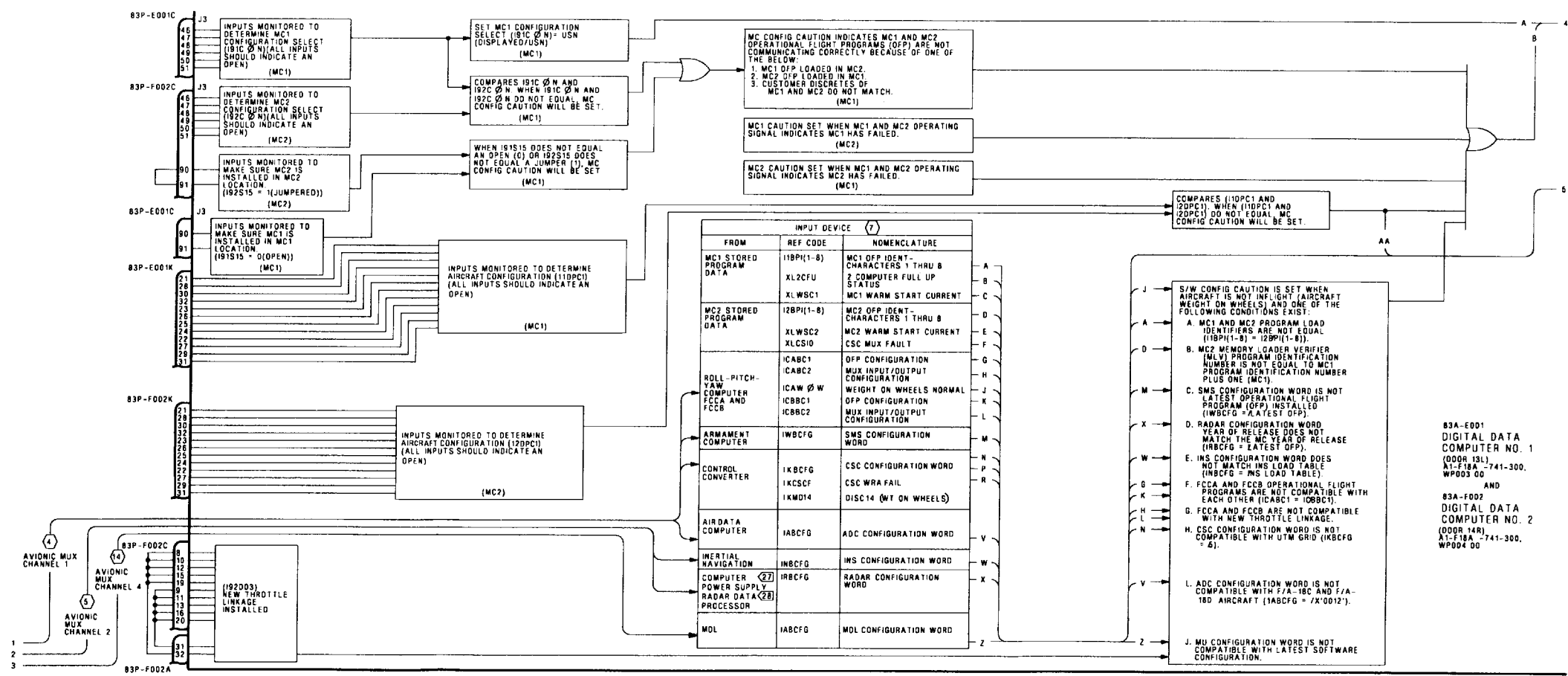


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 2)

Figure 1.

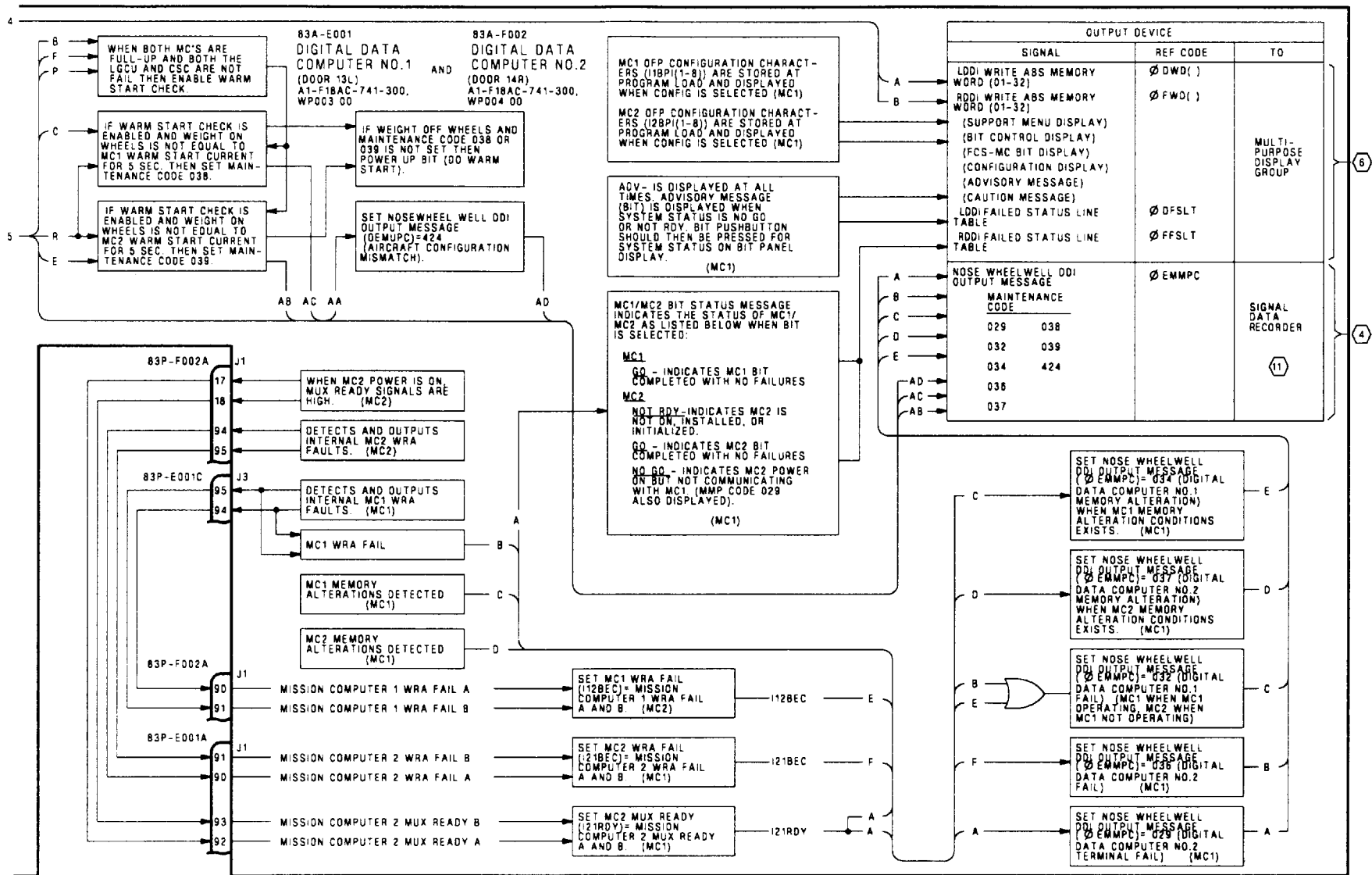


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 3)

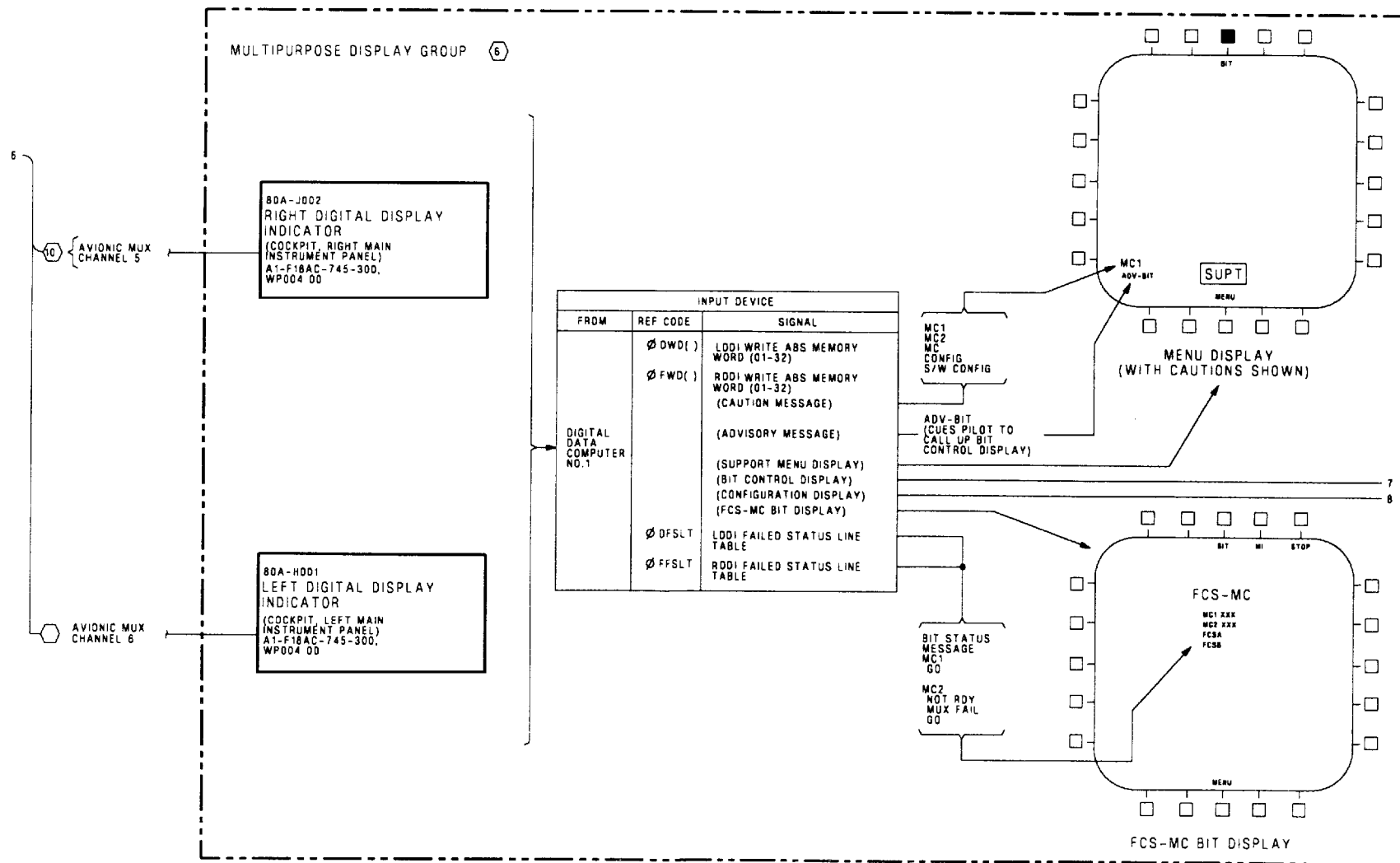


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 4)

Figure 1.

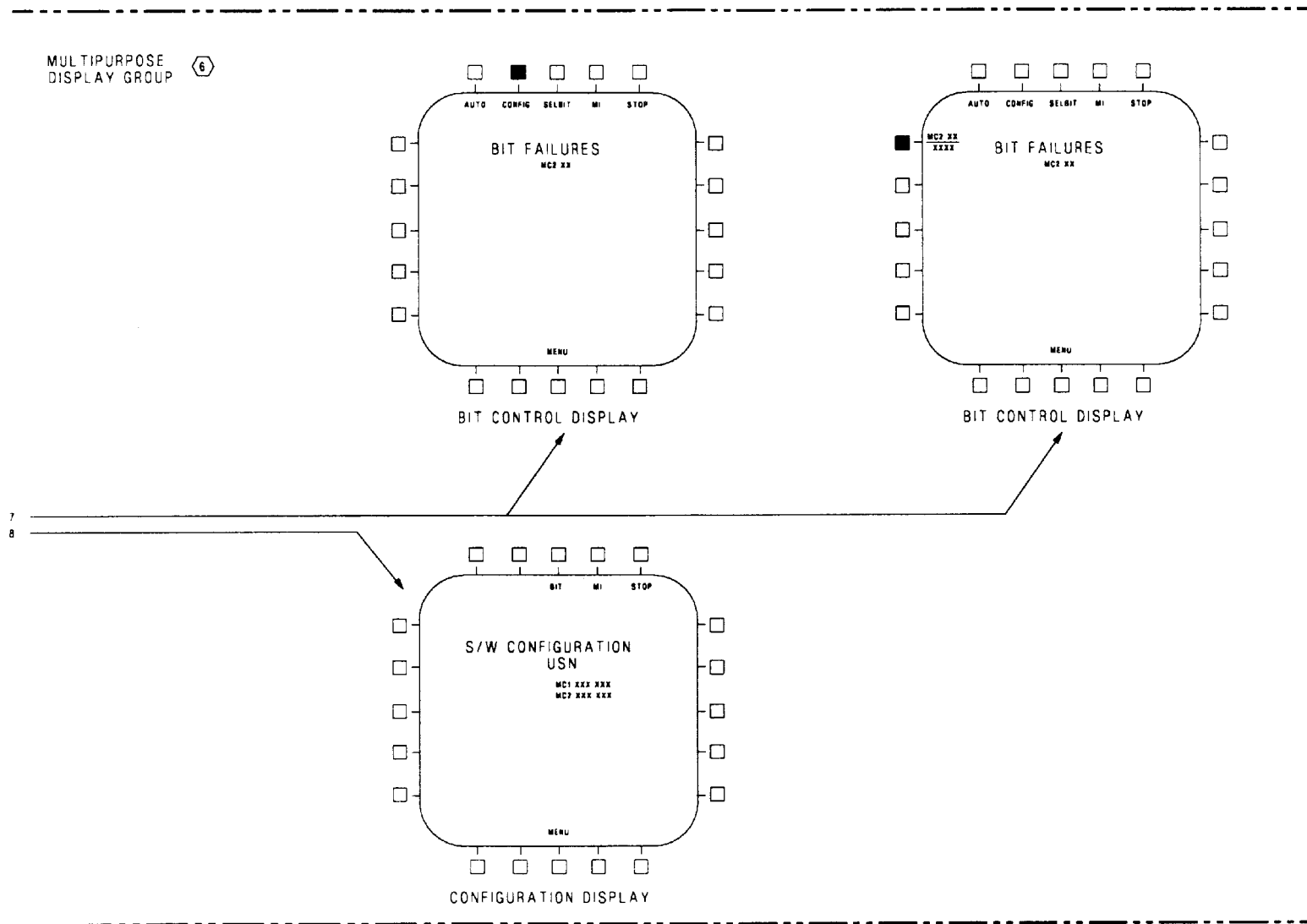


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 5)

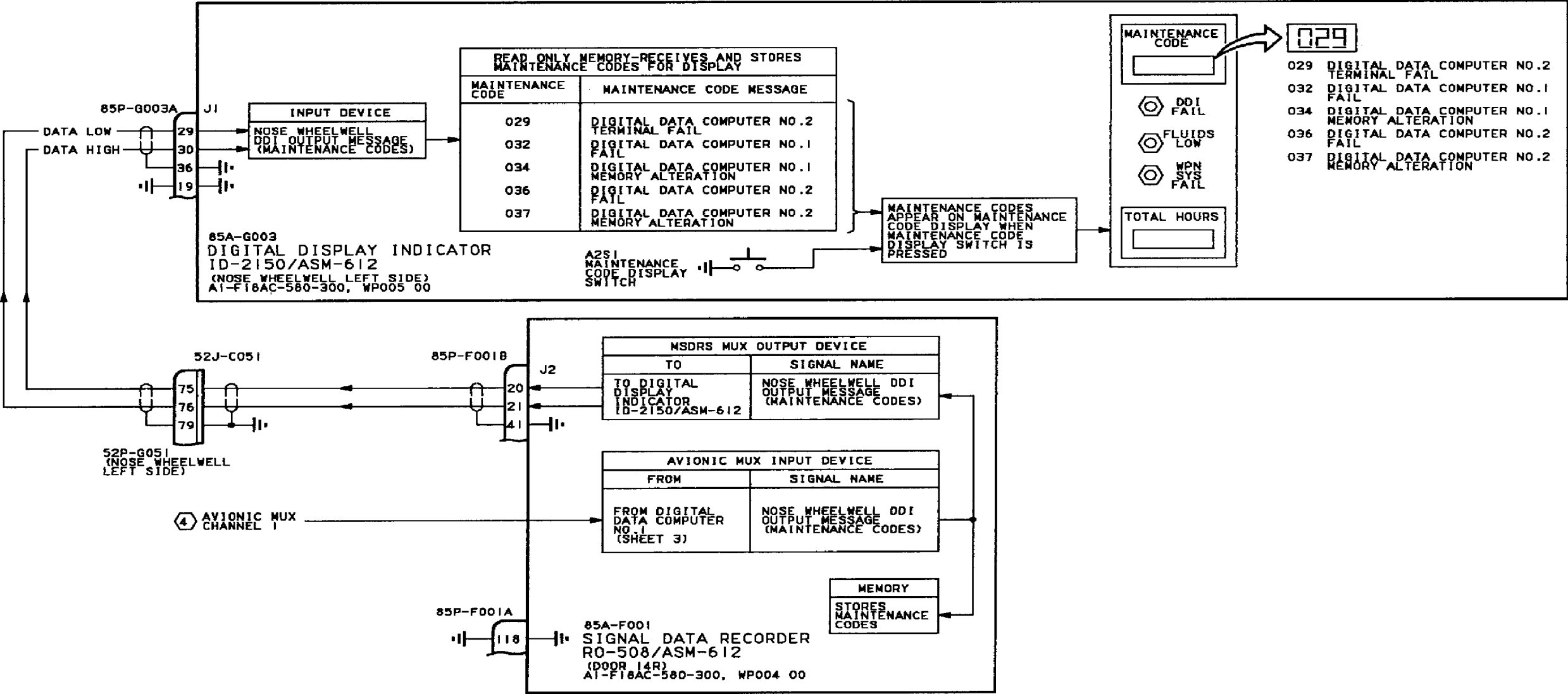


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 6)

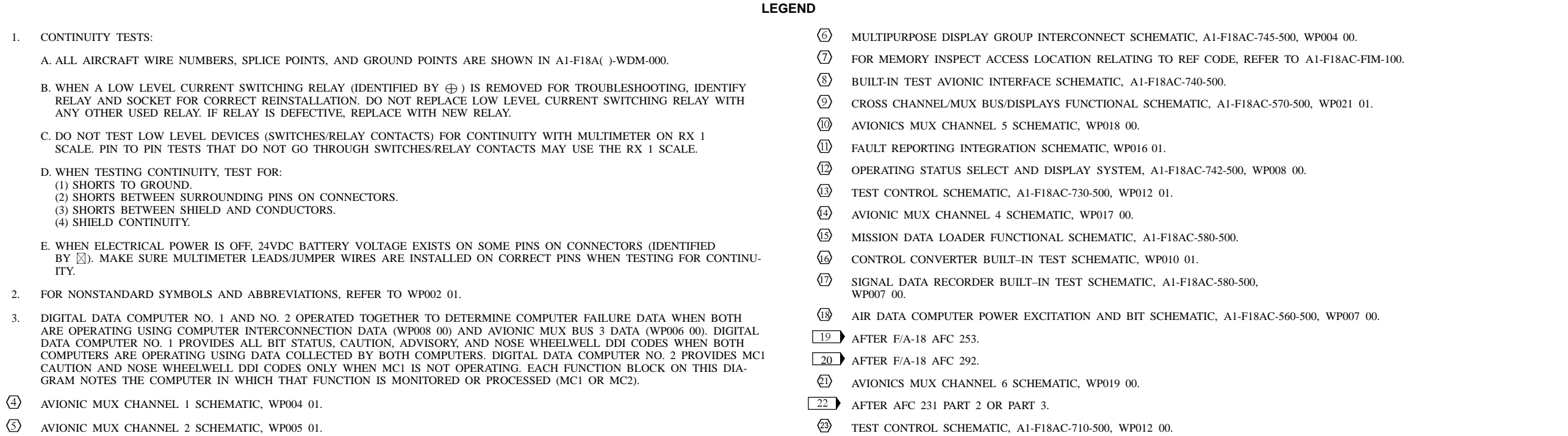


Figure 1.

Figure 1. Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (Sheet 7)

Figure 1.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - FAULT REPORTING INTEGRATION

MISSION COMPUTER SYSTEM

This WP supersedes WP016 00, dated 1 January 2002.

Title	Work Package
Schematic - Fault Reporting Integration (F/A-18A/B)	016 01
Schematic - Fault Reporting Integration (AFTER F/A-18 AFC 225)	016 02
Schematic - Fault Reporting Integration (AFTER F/A-18 AFC 253 OR F/A-18 AFC 292)	016 03

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - FAULT REPORTING INTEGRATION****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A/B****This WP supersedes WP016 01, dated 1 January 2001.**

Reference Material

None

Alphabetical Index**Subject****Page No.**

Fault Reporting Integration Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 9	-	Eliminate False Aft Cockpit Master Caution (ECP MDA-F/A-18-00085)	1 Apr 87	ECP Coverage Only

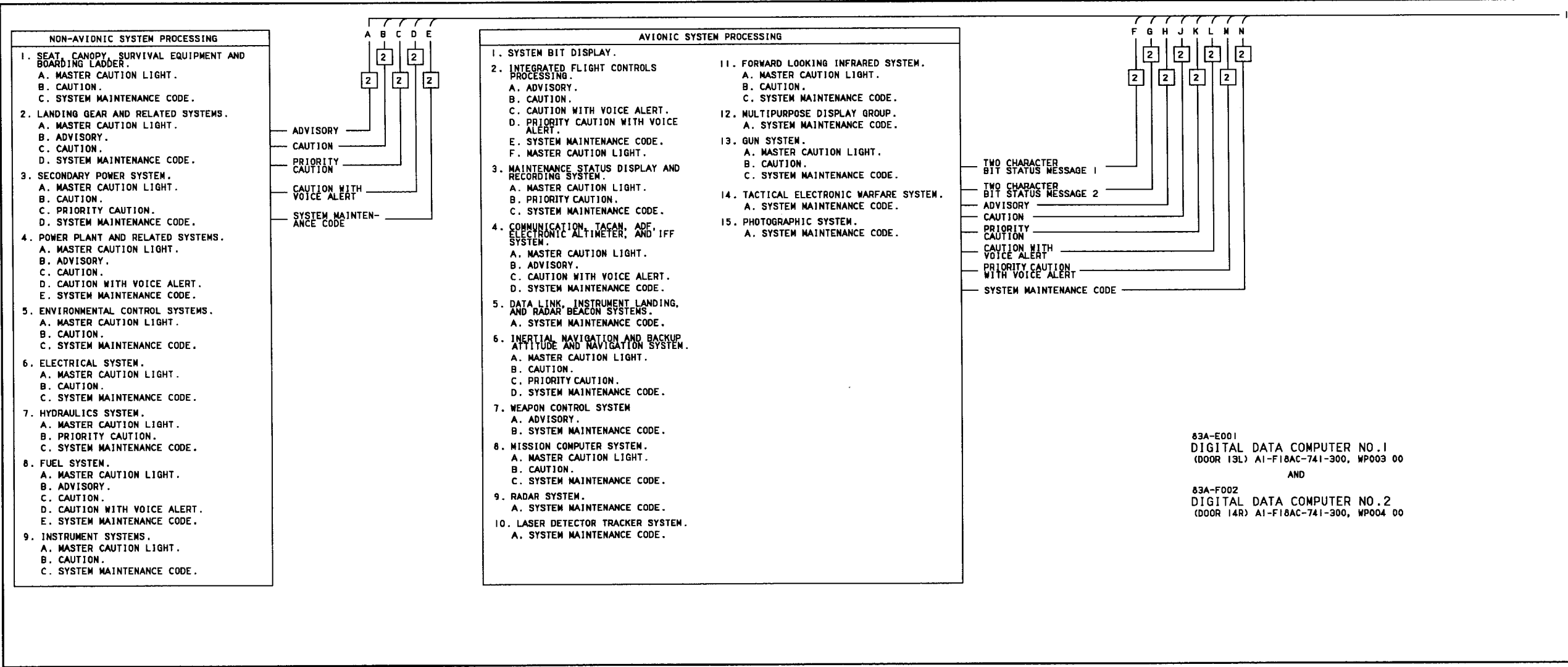


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 1)

Figure 1.

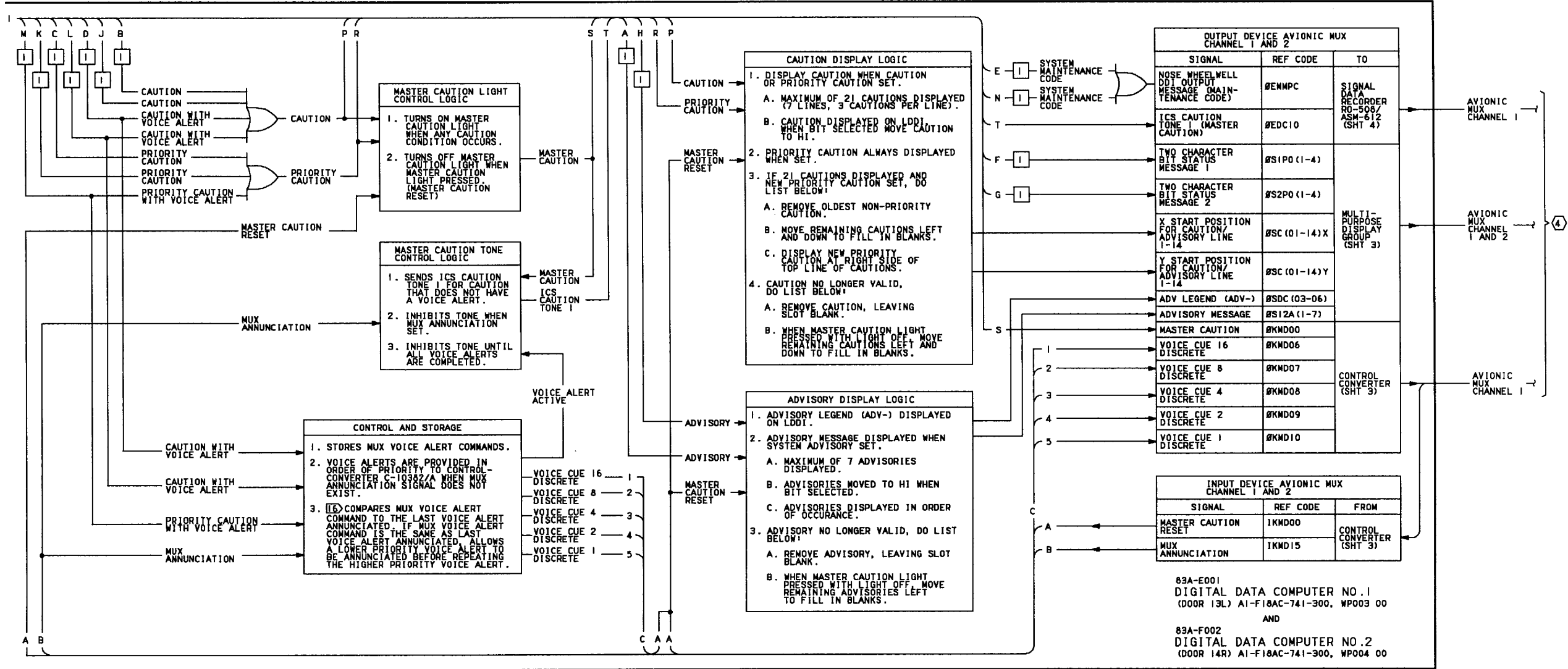


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 2)

Figure 1.

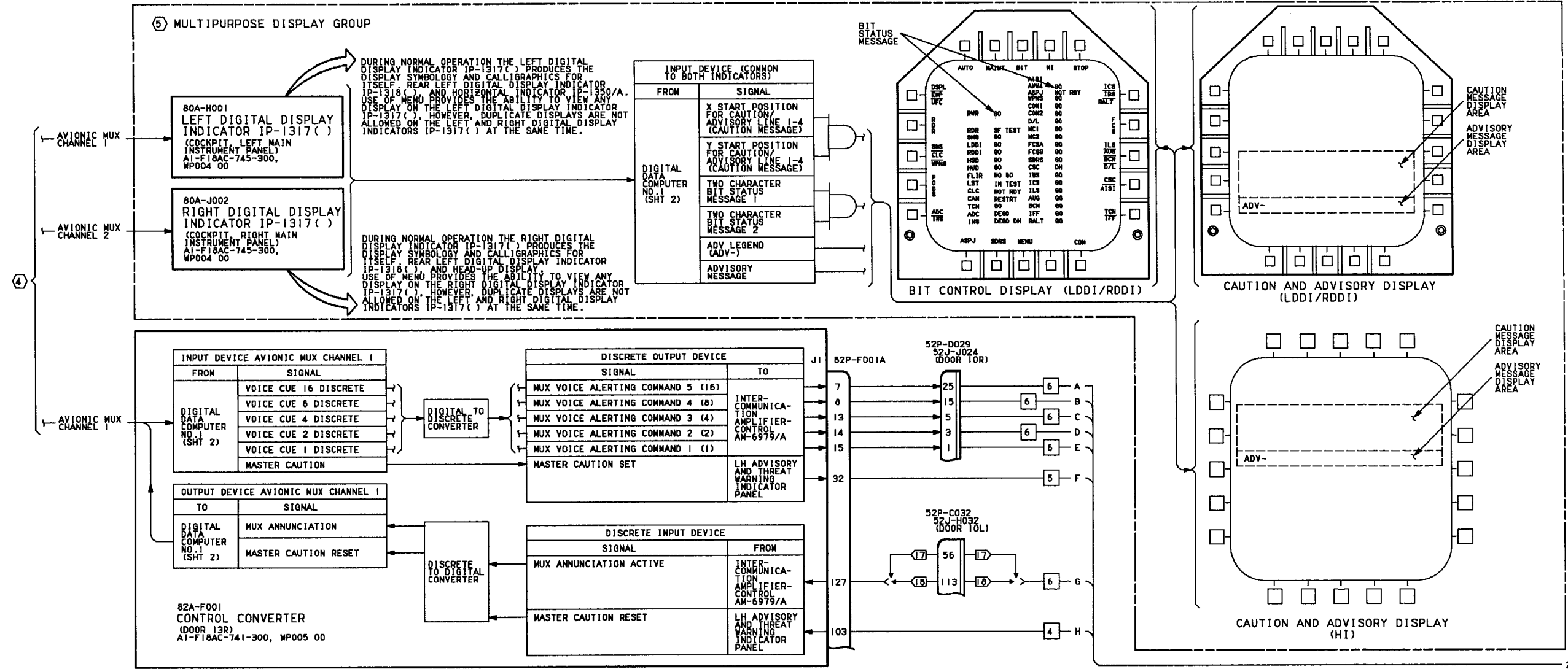


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 3)

Figure 1.

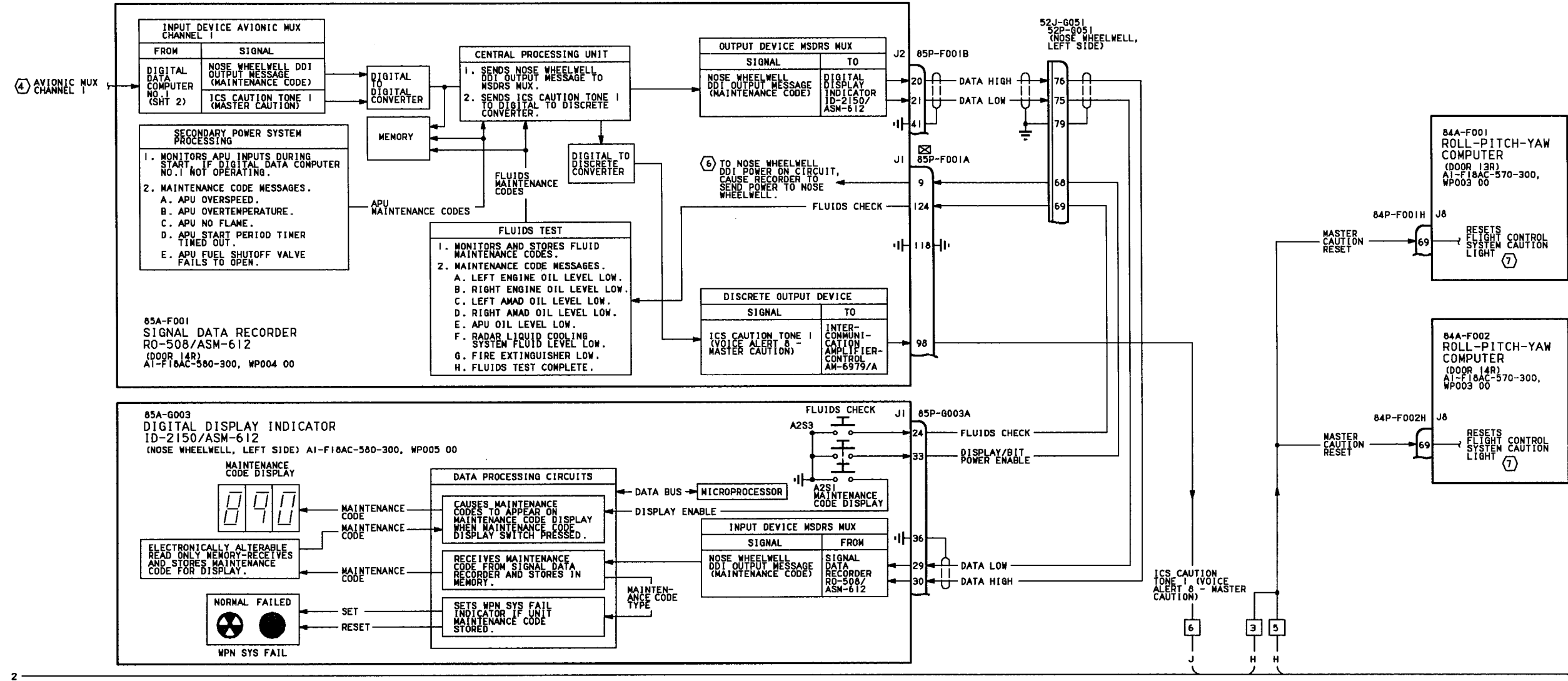


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 4)

Figure 1.

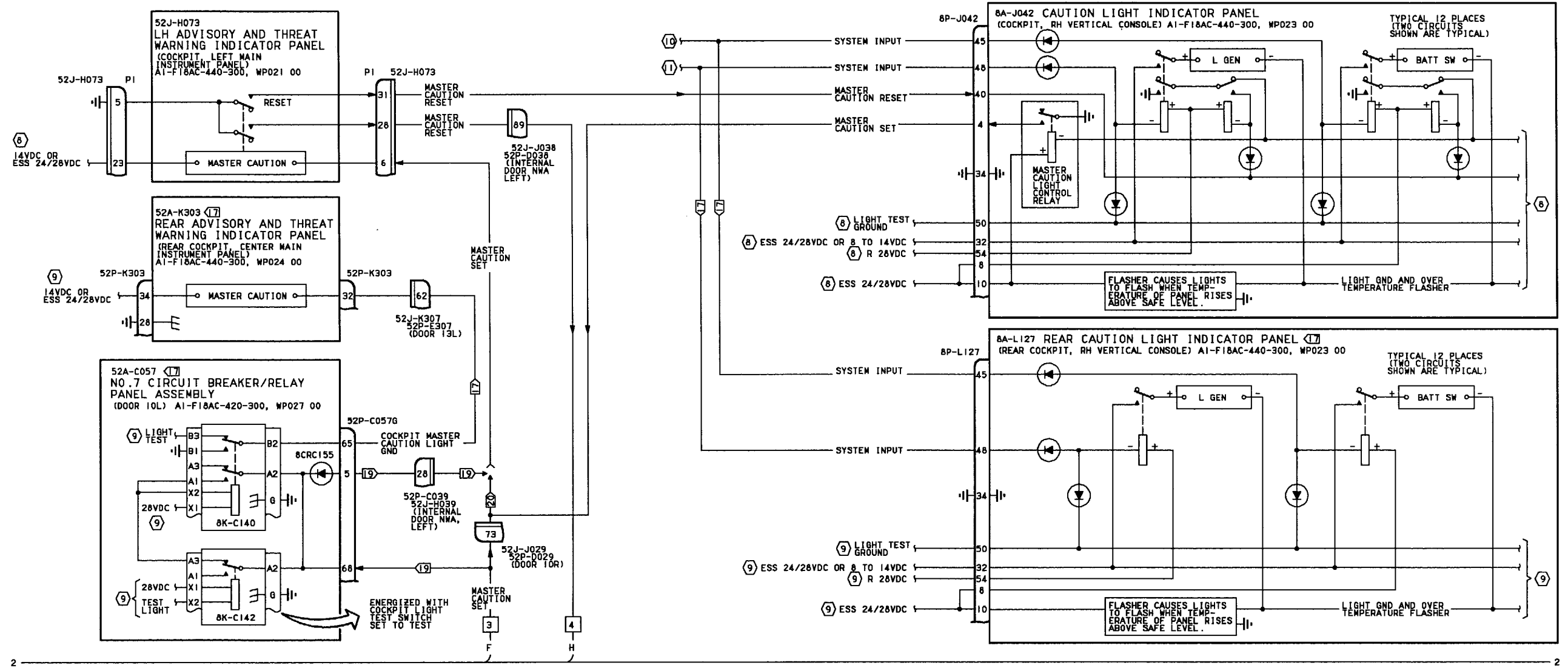


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 5)

Figure 1.

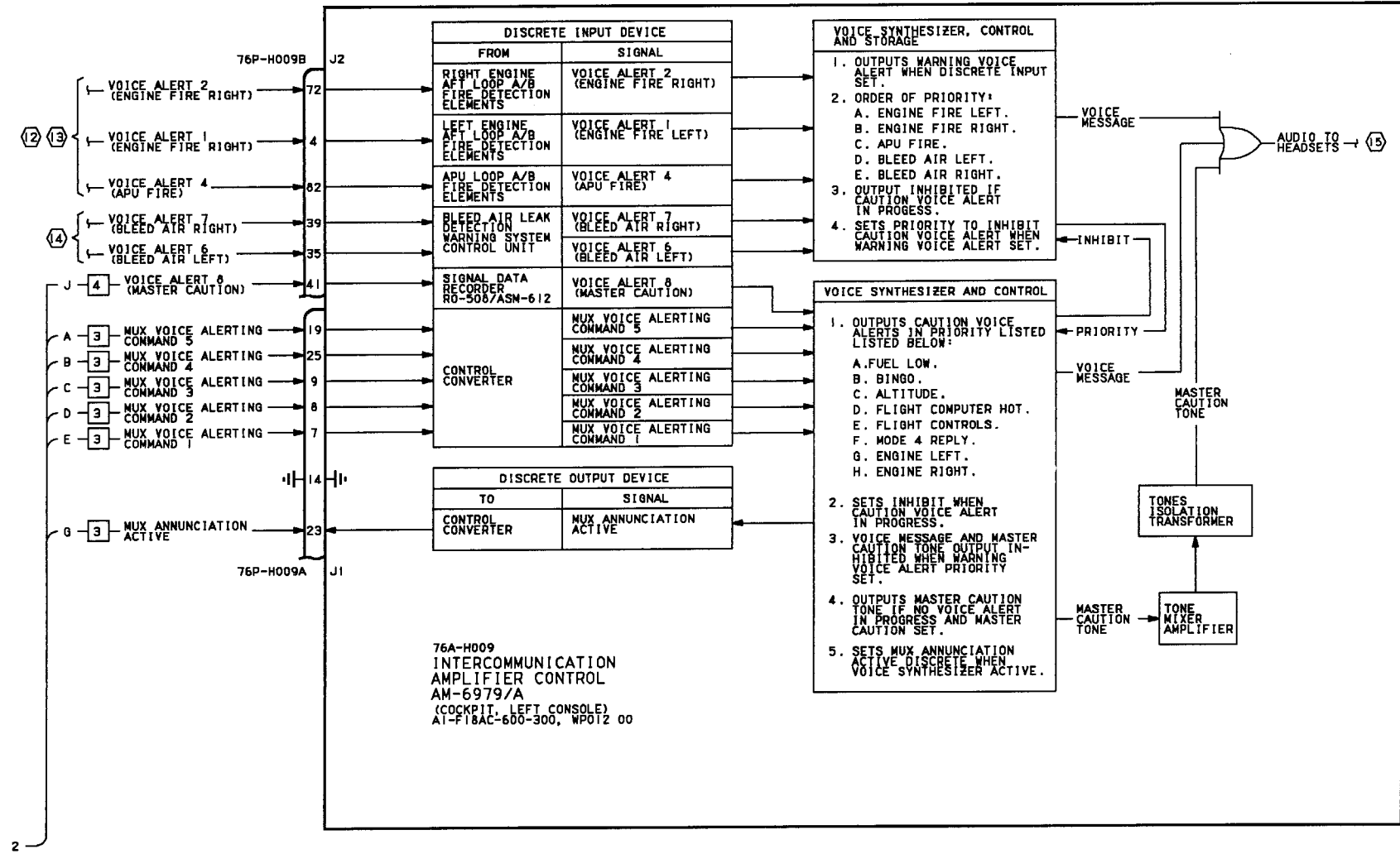


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 6)

Figure 1.

LEGEND

1. CONTINUITY TESTS:

A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.

B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY ⊕) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.

C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.

D. WHEN TESTING CONTINUITY, TEST FOR:

(1) SHORTS TO GROUND.

(2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.

(3) SHORTS BETWEEN SHIELD AND CONDUCTORS.

(4) SHIELD CONTINUITY.

E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY ☒). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.
3. LINE UNDER LETTER (S) INDICATES LOWER CASE PIN LETTER.
- ④ SEE APPLICABLE AVIONIC MUX CHANNEL SCHEMATIC, WP001 00.
- ⑤ THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD UP DISPLAY, HORIZONTAL INDICATOR IP-1350/A, ALSO ON F/A-18B; REAR LEFT DIGITAL DISPLAY INDICATOR IP-1318(), REAR RIGHT DIGITAL DISPLAY INDICATOR IP-1318(), AND REAR CENTER DIGITAL DISPLAY INDICATOR IP-1317(). FOR MULTIPURPOSE DISPLAY GROUP SCHEMATIC, REFER TO A1-F18AC-745-500, WP001 00.

⑥ POWER SCHEMATIC, A1-F18AC-580-500, WP005 00.

⑦ CAUTION AND BIT DISPLAYS SCHEMATIC, A1-F18AC-570-500, WP024 00.

⑧ COCKPIT CAUTION LIGHTS SCHEMATIC, A1-F18AC-440-500, WP006 00.

⑨ REAR COCKPIT CAUTION LIGHTS SCHEMATIC, A1-F18AC-440-500, WP007 00.

⑩ DC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.

⑪ AC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP003 00.

⑫ LOOP A FIRE DETECTION SYSTEM SCHEMATIC, A1-F18AC-240-500, WP009 00.

⑬ LOOP B FIRE DETECTION SYSTEM SCHEMATIC, A1-F18AC-240-500, WP009 00.

⑭ BLEED AIR LEAK DETECTION SYSTEM SCHEMATIC, A1-F18AC-410-500, WP006 00.

⑮ INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, A1-F18AC-600-500, WP013 00.

⑯ WITH DIGITAL DATA COMPUTER NO. 1 AND NO. 2 CONFIG/IDENT 85A AND UP, (A1-F18AC-SCM-000).

⑰ F/A-18B.

⑱ F/A-18A.

⑲ F/A-18B 161704 AND UP; ALSO 161354 THRU 161360 AFTER F/A-18 AFC 9.

⑳ F/A-18A; ALSO 161354 THRU 161360 BEFORE F/A-18 AFC 9.

Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 7)

Figure 1.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - FAULT REPORTING INTEGRATION****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AND F/A-18B AFTER F/A-18 AFC 225**

Reference Material

None

Alphabetical Index**Subject****Page No.**

Fault Reporting Integration Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 9	-	Eliminate False Aft Cockpit Master Caution (ECP MDA-F/A-18-00085)	1 Apr 87	ECP Coverage Only
F/A-18 AFC 225	-	Five (5) Avionics Multiplex Bus Upgrade, Incorporation of (ECP MDA-F/A-18 0529)	1 Jun 02	-

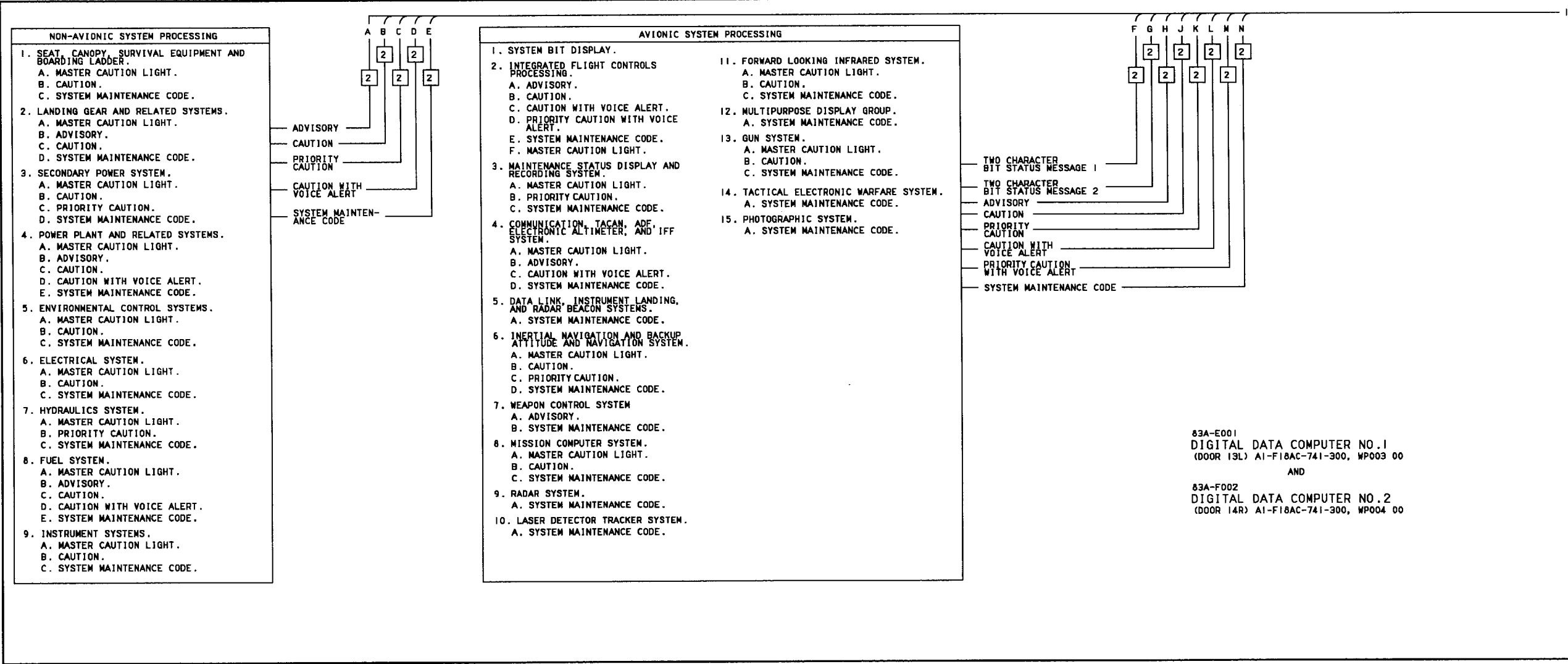


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 1)

Figure 1.

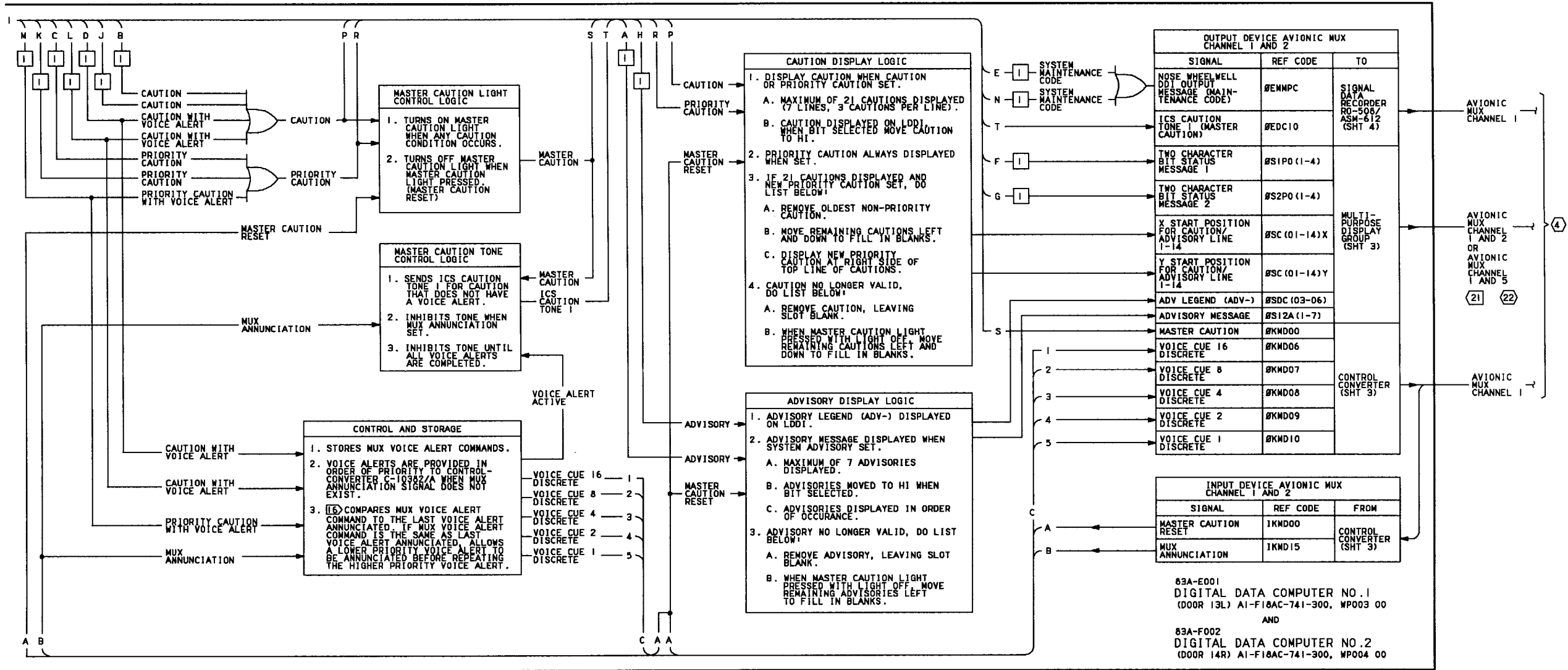


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 2)

Figure 1.

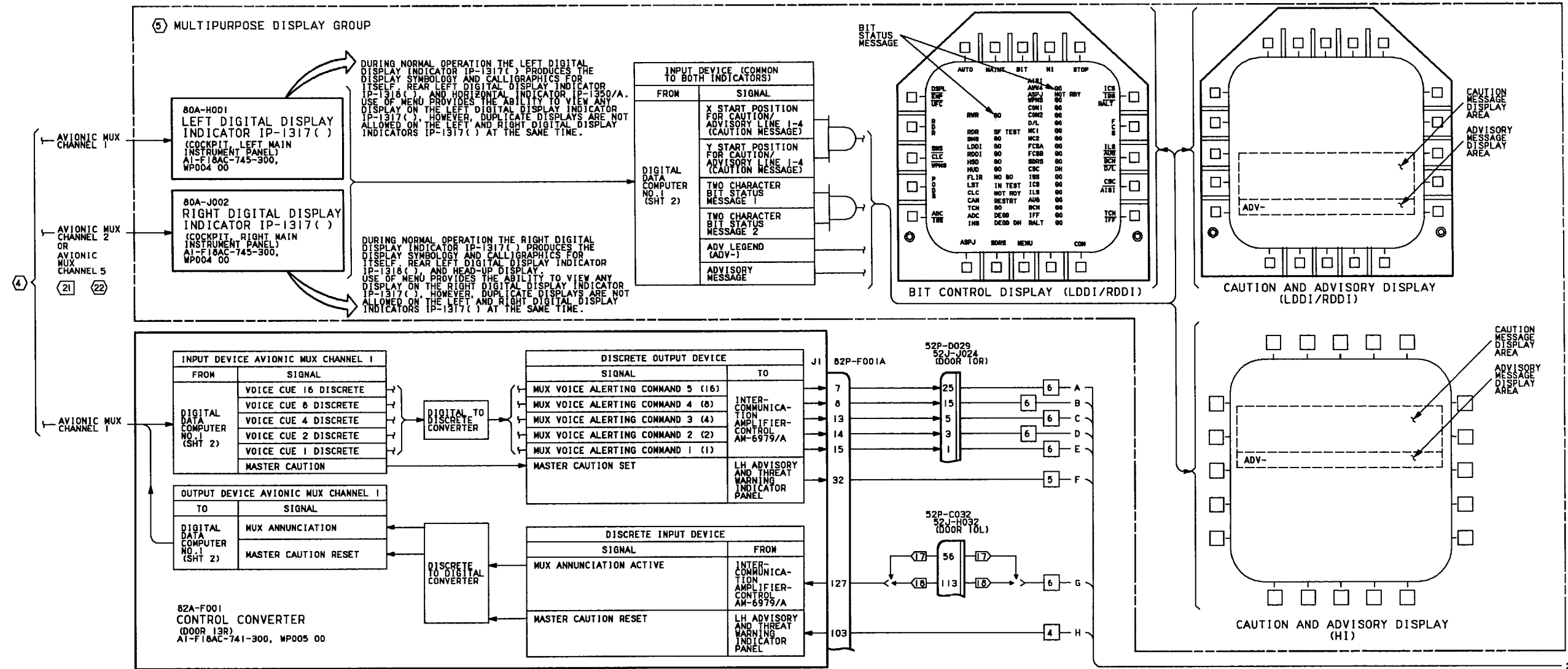


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 3)

Figure 1.

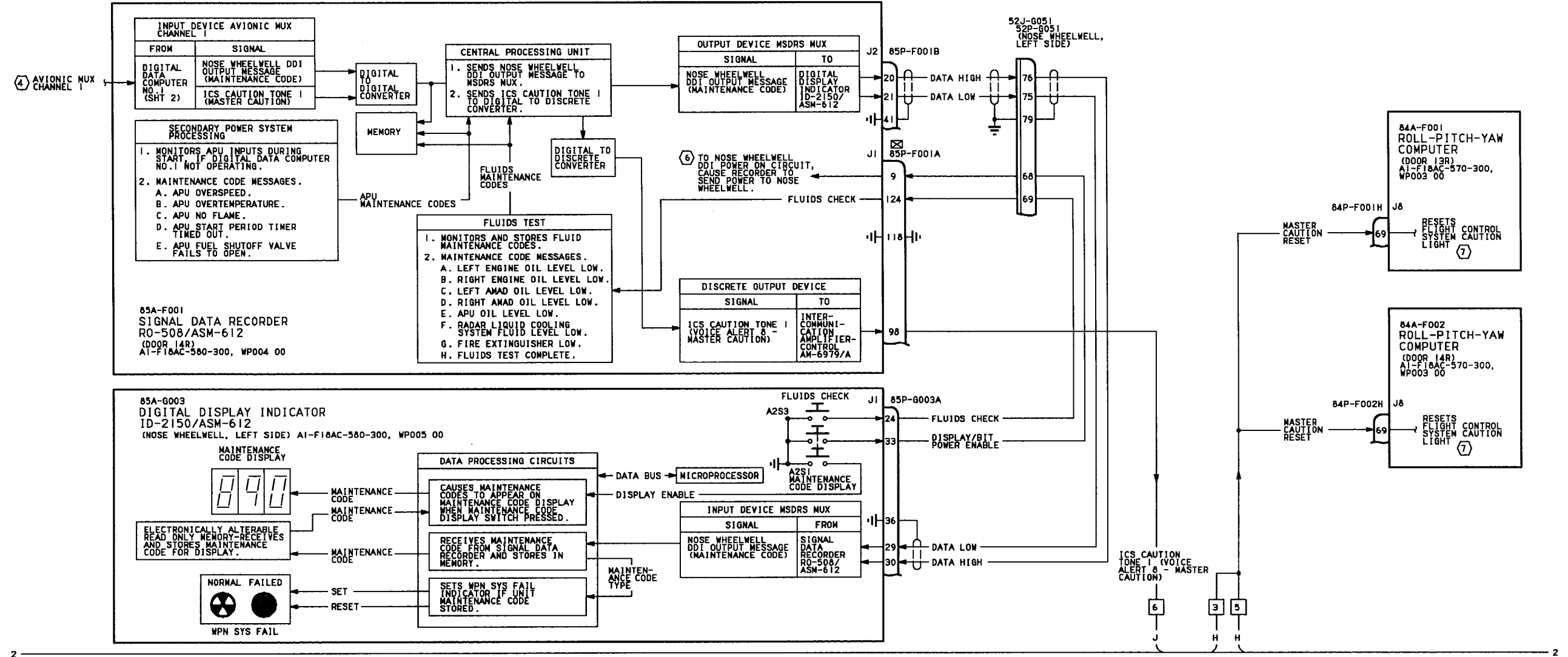


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 4)

Figure 1.

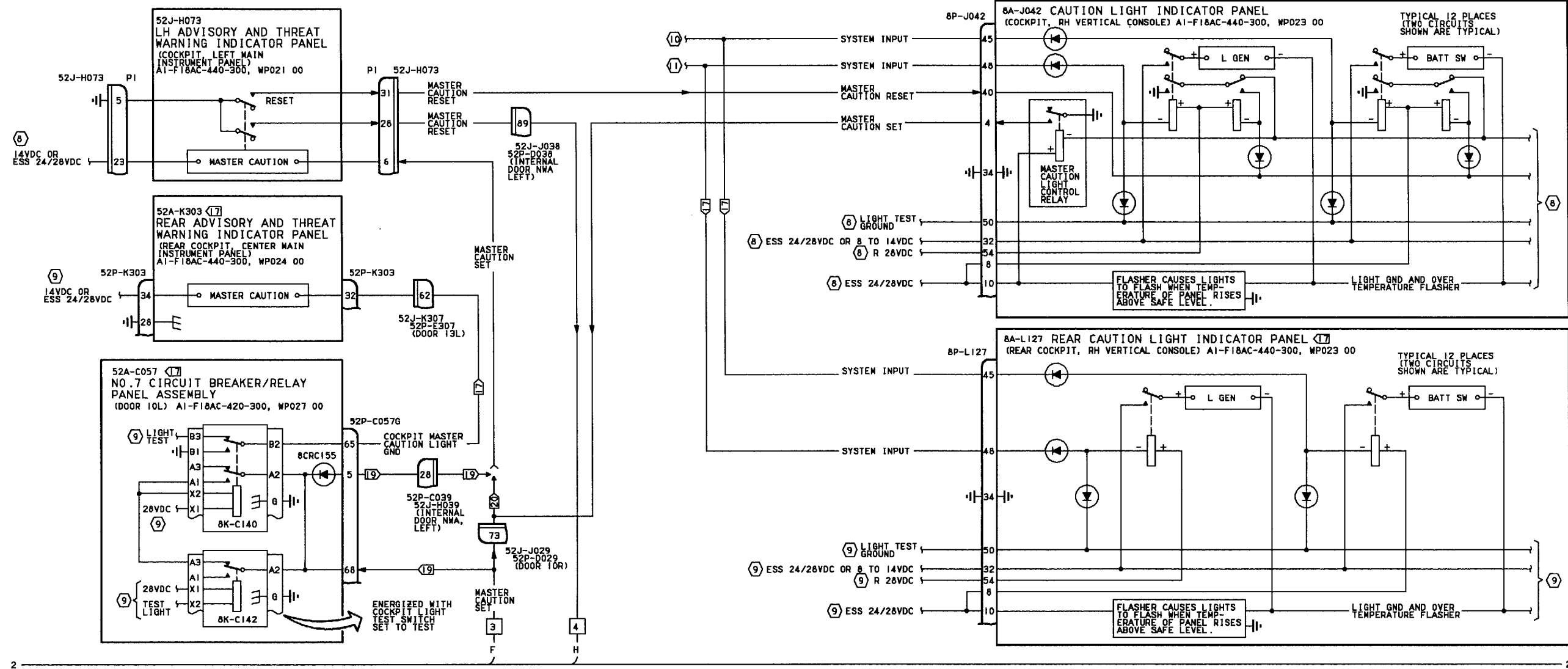


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 5)

Figure 1.

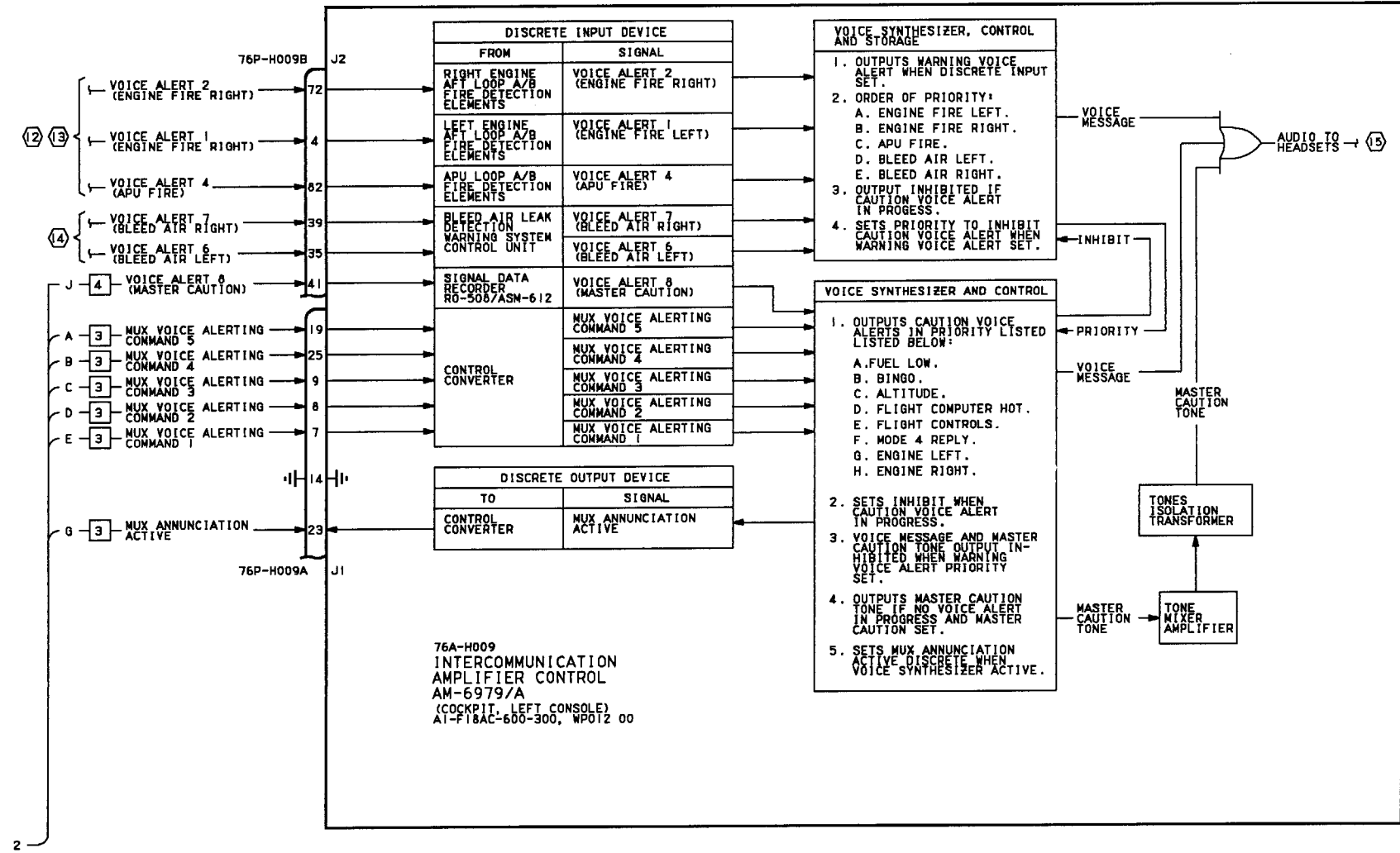


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 6)

Figure 1.

LEGEND

1.

CONTINUITY TESTS:

A.

ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.

B.

WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.

C.

DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.

D.

WHEN TESTING CONTINUITY, TEST FOR:

(1)

SHORTS TO GROUND.

(2)

SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.

(3)

SHORTS BETWEEN SHIELD AND CONDUCTORS.

(4)

SHIELD CONTINUITY.

E.

WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2.

FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.
3.

LINE UNDER LETTER (S) INDICATES LOWER CASE PIN LETTER.
- 4

SEE APPLICABLE AVIONIC MUX CHANNEL SCHEMATIC, WP001 00.
- 5

THE MULTIPURPOSE DISPLAY GROUP IS MADE UP OF THE LEFT DIGITAL DISPLAY INDICATOR IP-1317(), RIGHT DIGITAL DISPLAY INDICATOR IP-1317(), HEAD UP DISPLAY, HORIZONTAL INDICATOR IP-1350/A, ALSO ON F/A-18B; REAR LEFT DIGITAL DISPLAY INDICATOR IP-1318(), REAR RIGHT DIGITAL DISPLAY INDICATOR IP-1318(), AND REAR CENTER DIGITAL DISPLAY INDICATOR IP-1317(). FOR MULTIPURPOSE DISPLAY GROUP SCHEMATIC, REFER TO A1-F18AC-745-500, WP001 00.

6

POWER SCHEMATIC, A1-F18AC-580-500, WP005 00.

7

CAUTION AND BIT DISPLAYS SCHEMATIC, A1-F18AC-570-500, WP024 00.

8

COCKPIT CAUTION LIGHTS SCHEMATIC, A1-F18AC-440-500, WP006 00.

9

REAR COCKPIT CAUTION LIGHTS SCHEMATIC, A1-F18AC-440-500, WP007 00.

10

DC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.

11

AC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP003 00.

12

LOOP A FIRE DETECTION SYSTEM SCHEMATIC, A1-F18AC-240-500, WP009 00.

13

LOOP B FIRE DETECTION SYSTEM SCHEMATIC, A1-F18AC-240-500, WP009 00.

14

BLEED AIR LEAK DETECTION SYSTEM SCHEMATIC, A1-F18AC-410-500, WP006 00.

15

INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, A1-F18AC-600-500, WP013 00.

16

WITH DIGITAL DATA COMPUTER NO. 1 AND NO. 2 CONFIG/IDENT 85A AND UP, (A1-F18AC-SCM-000).

17

F/A-18B.

18

F/A-18A.

19

F/A-18B 161704 AND UP; ALSO 161354 THRU 161360 AFTER F/A-18 AFC 9.

20

F/A-18A; ALSO 161354 THRU 161360 BEFORE F/A-18 AFC 9.

21

F/A-18A AND F/A-18B AFTER F/A-18 AFC 225.

22

AVIONIC MUX CHANNEL 5 SCHEMATIC, WP018 00.

Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 7)

Figure 1.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - FAULT REPORTING INTEGRATION****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292**

Reference Material

None

Alphabetical Index**Subject****Page No.**

Fault Reporting Integration Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	U.S. Naval Reserves F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	U.S. Marine Corps Reserves F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jan 01	-

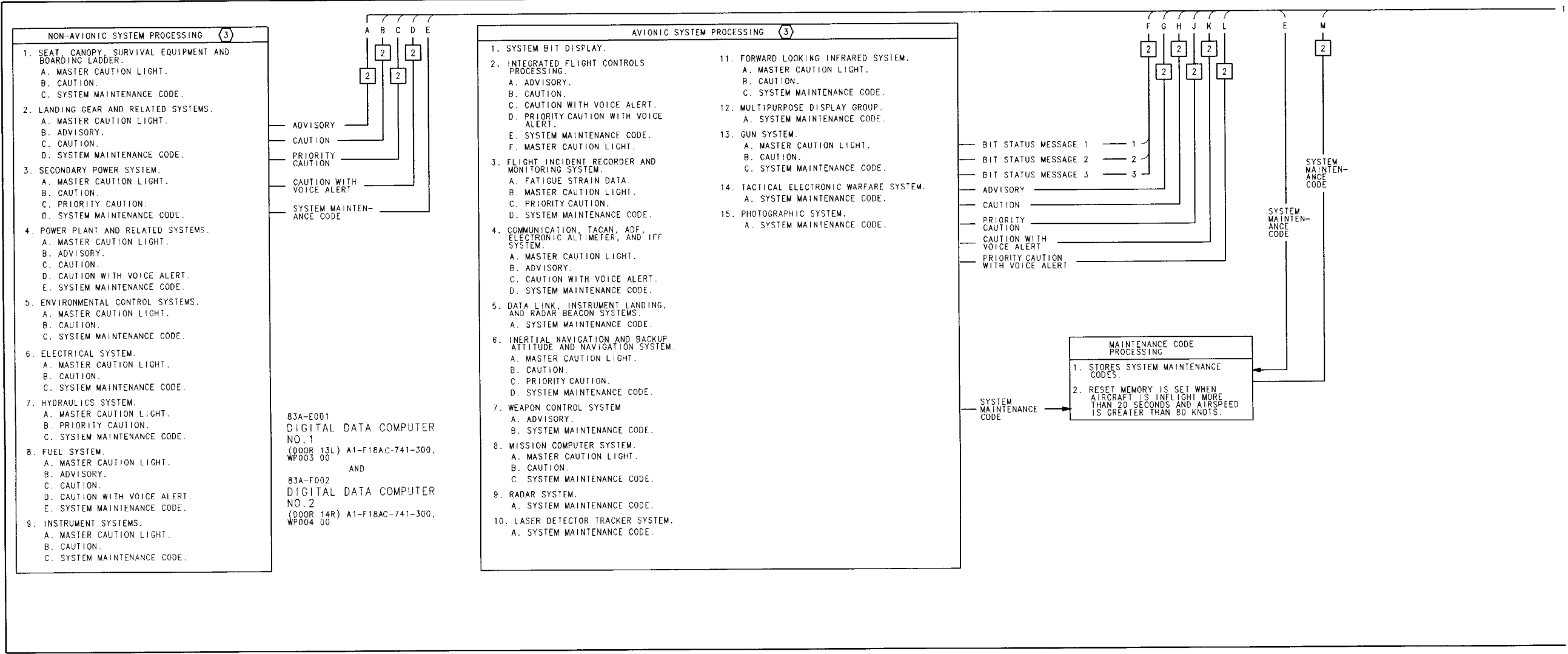


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 1)

Figure 1.

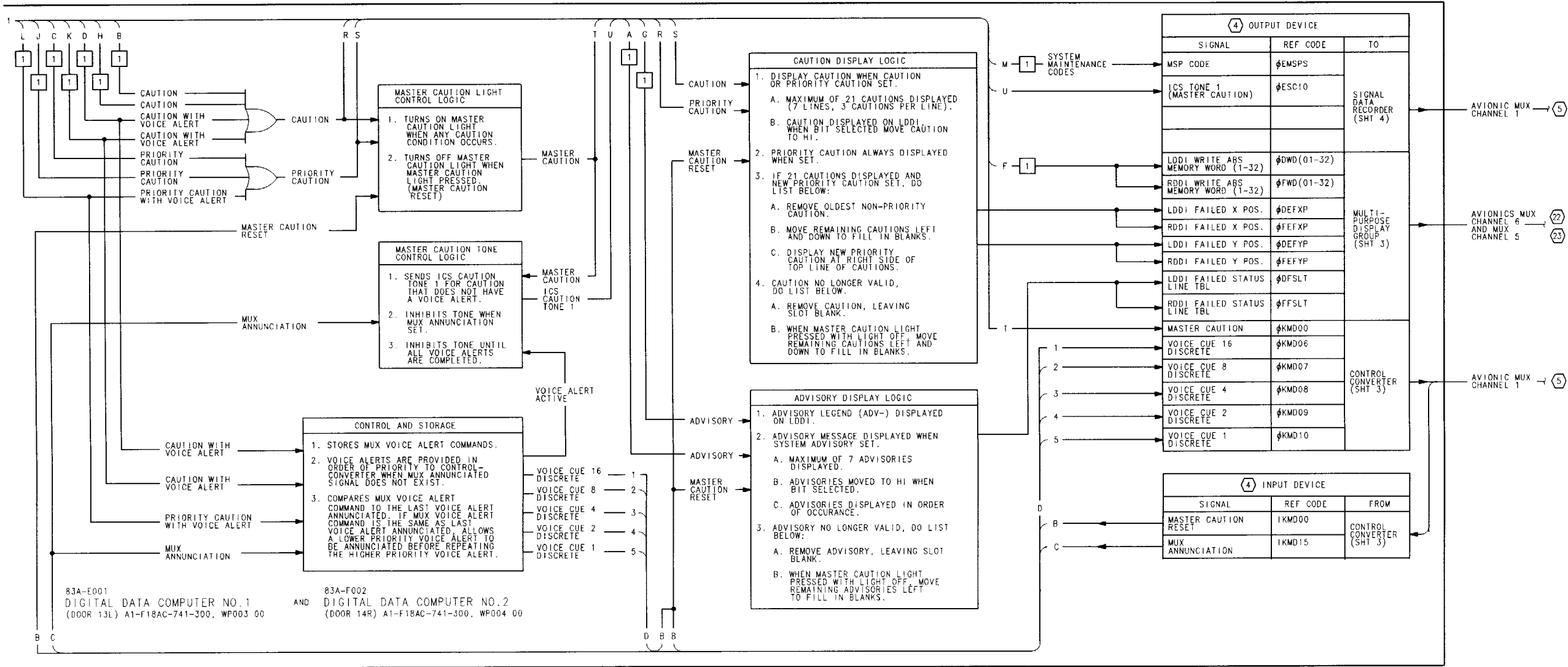


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 2)

Figure 1.



Figure 1.

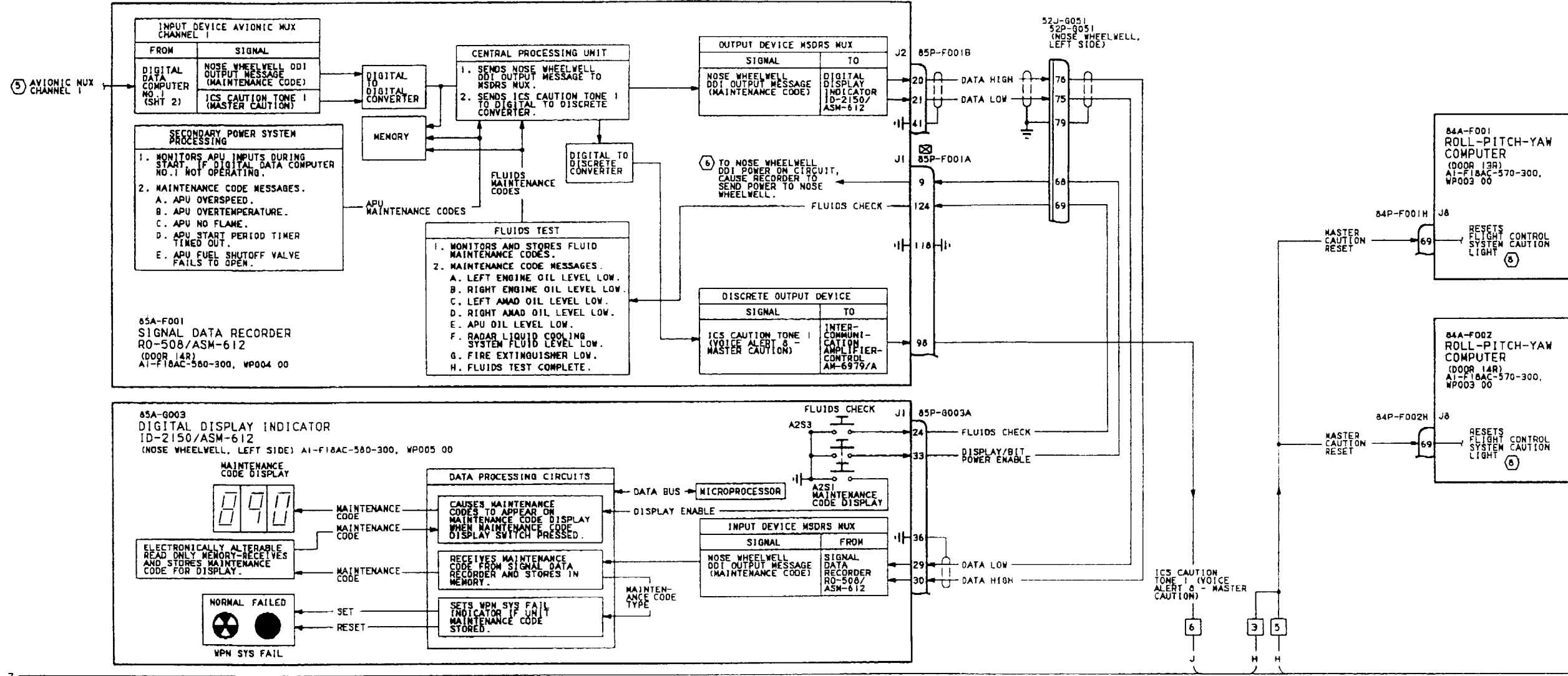


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 4)

Figure 1.

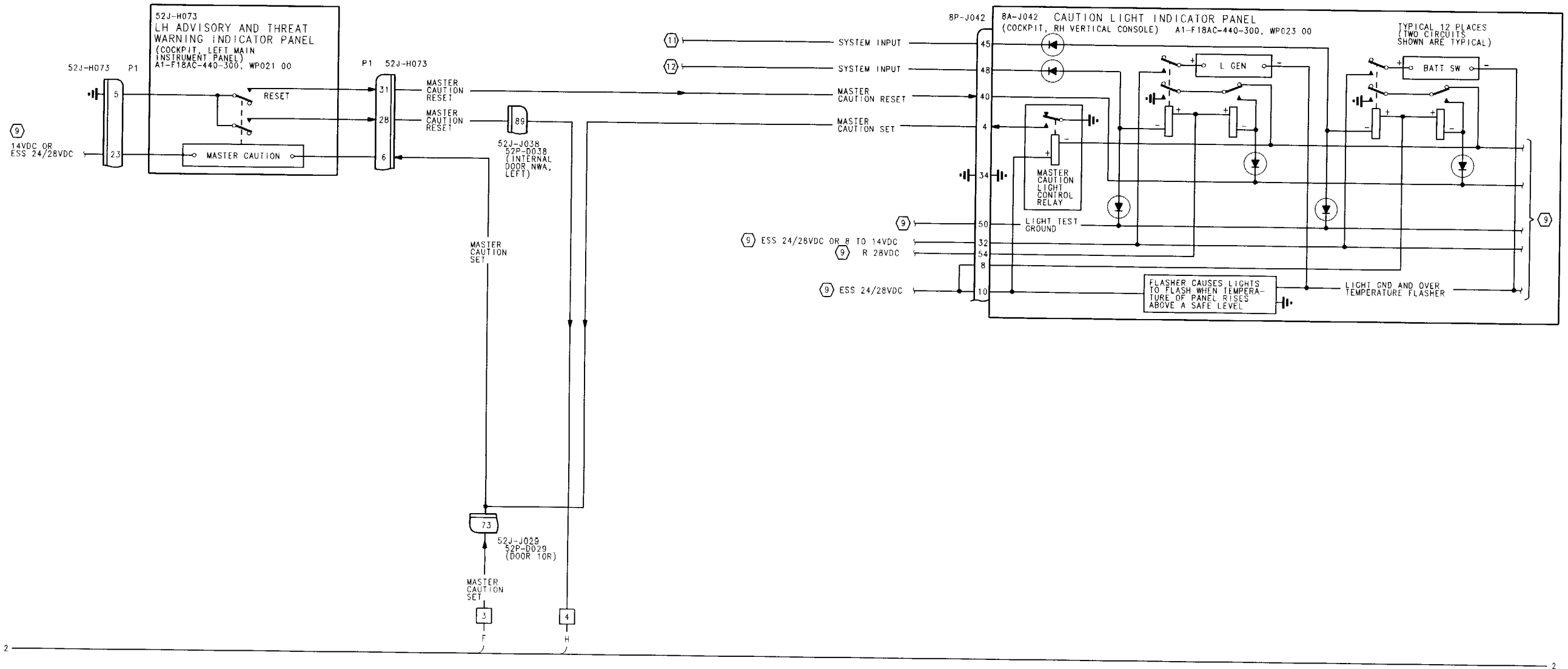


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 5)

Figure 1.

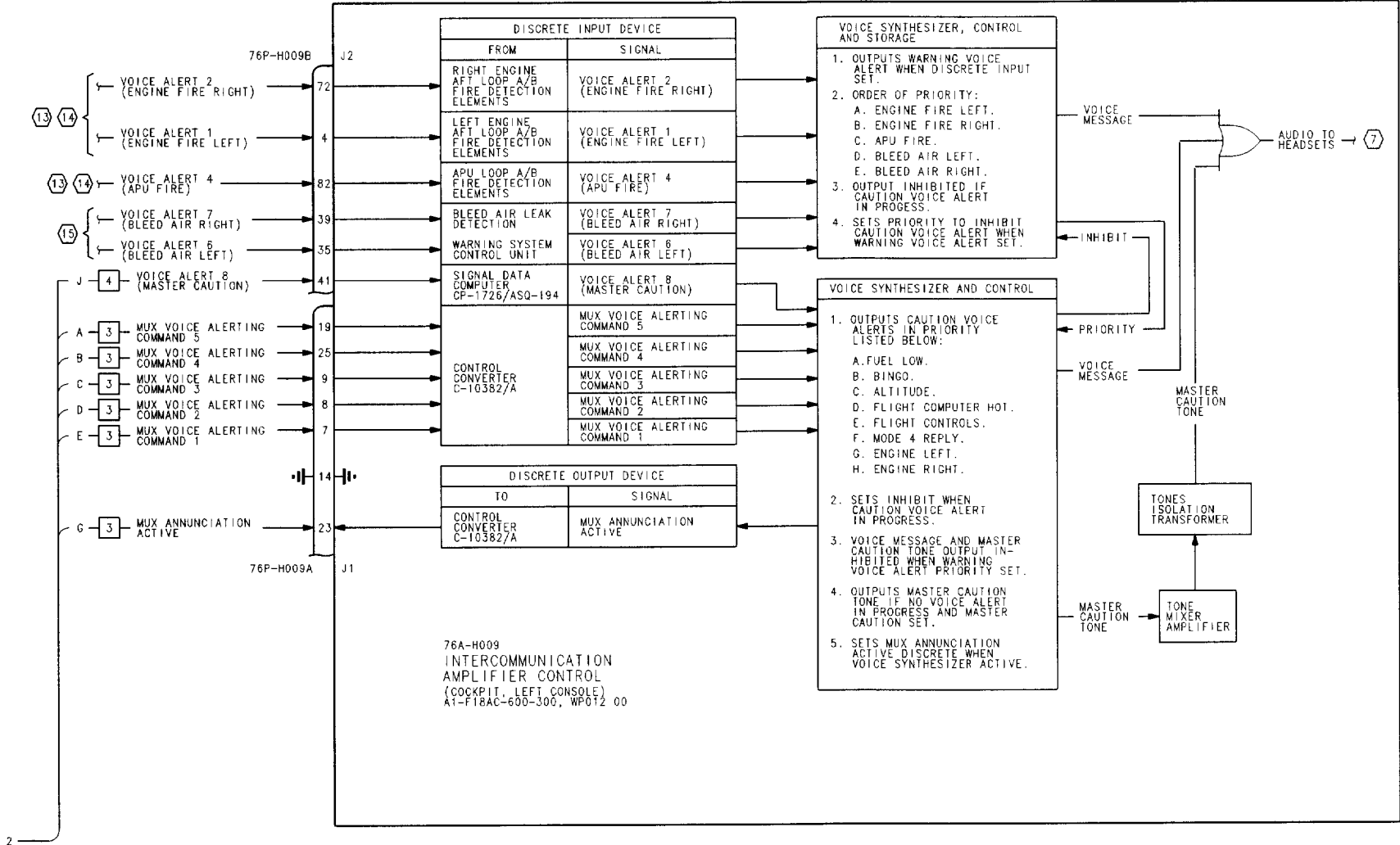


Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 6)

Figure 1.

LEGEND

1. CONTINUITY TESTS:

A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.

B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY ⊕) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.

C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.

D. WHEN TESTING CONTINUITY, TEST FOR:

(1) SHORTS TO GROUND.

(2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.

(3) SHORTS BETWEEN SHIELD AND CONDUCTORS.

(4) SHIELD CONTINUITY.

E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY ☒). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.
- ③ DIGITAL DATA COMPUTER NO. 1 AND NO. 2 OPERATE TOGETHER TO DETERMINE COMPUTER FAILURE DATA WHEN BOTH ARE OPERATING USING COMPUTER INTERCONNECTION DATA (WP008 01) AND AVIONIC MUX BUS 3 DATA (WP006 00). DIGITAL DATA COMPUTER NO. 1 PROVIDES ALL BIT STATUS, CAUTION, ADVISORY, AND NOSE WHEELWELL DDI CODES WHEN BOTH COMPUTERS ARE OPERATING USING DATA COLLECTED BY BOTH COMPUTERS. DIGITAL DATA COMPUTER NO. 2 PROVIDES MCI CAUTION AND NOSE WHEELWELL DDI CODES ONLY WHEN MCI IS NOT OPERATING. EACH FUNCTION BLOCK ON THIS DIAGRAM NOTES THE COMPUTER IN WHICH THAT FUNCTION IS MONITORED OR PERFORMED (MC1 OR MC2).
- ④ FOR MEMORY INSPECT ADDRESS LOCATION RELATING TO REF CODE, REFER TO A1-F18AC-FIM-100.

⑤ AVIONIC MUX CHANNEL 1, SCHEMATIC WP004 00.

⑥ MULTIPURPOSE DISPLAY GROUP INTERCONNECT SCHEMATIC, A1-F18AC-745-500, WP004 00.

⑦ INTERCOMMUNICATION AND AUDIO SYSTEM FUNCTIONAL SCHEMATIC, A1-F18AC-600-500, WP013 00.

⑧ CAUTION AND BIT DISPLAYS SCHEMATIC, A1-F18AC-570-500, WP024 00.

⑨ COCKPIT CAUTION LIGHTS SCHEMATIC, A1-F18AC-440-500, WP006 00.

⑩ DELETED.

⑪ DC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP004 00.

⑫ AC POWER SYSTEM SCHEMATIC, A1-F18AC-420-500, WP003 00.

⑬ LOOP A FIRE DETECTION SYSTEM SCHEMATIC, A1-F18AC-240-500, WP009 00.

⑭ LOOP B FIRE DETECTION SYSTEM SCHEMATIC, A1-F18AC-240-500, WP009 00.

⑮ BLEED AIR LEAK DETECTION SYSTEM SCHEMATIC, A1-F18AC-410-500, WP006 00.

⑯ DELETED.

⑰ DELETED.

⑱ DELETED.

⑲ DELETED.

⑳ DELETED.

㉑ DELETED.

㉒ AVIONIC MUX CHANNEL 5, SCHEMATIC WP018 01.

㉓ AVIONIC MUX CHANNEL 6, SCHEMATIC WP019 01.

Figure 1.

Figure 1. Fault Reporting Integration Schematic (Sheet 7)

Figure 1.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - AVIONIC MUX CHANNEL 4

MISSION COMPUTER SYSTEM

This WP supersedes WP017 00, dated 1 January 2001.

Title	Work Package
Schematic - Avionic MUX Channel 4 (AFTER F/A-18 AFC 225 AND F/A-18 AFC 231)	017 01
Schematic - Avionic MUX Channel 4 (AFTER F/A-18 AFC 253 OR F/A-18 AFC 292)	017 02

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - AVIONIC MUX CHANNEL 4****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AND F/A-18B AFTER F/A-18 AFC 225 AND F/A-18 AFC 231**

Reference Material

None

Alphabetical Index**Subject****Page No.**

Avionic Mux Channel 4 Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 225	-	Five (5) Avionics Multiplex Bus Upgrade, Incorporation of (ECP MDA-F/A-18 0529)	1 Jun 02	-
F/A-18 AFC 231	-	Embedded Global Positioning System (GPS)/Inertial Navigation System (INS) (EGI), Incorporation of (ECP MDA-F/A-18 0521)	1 Jun 02	-

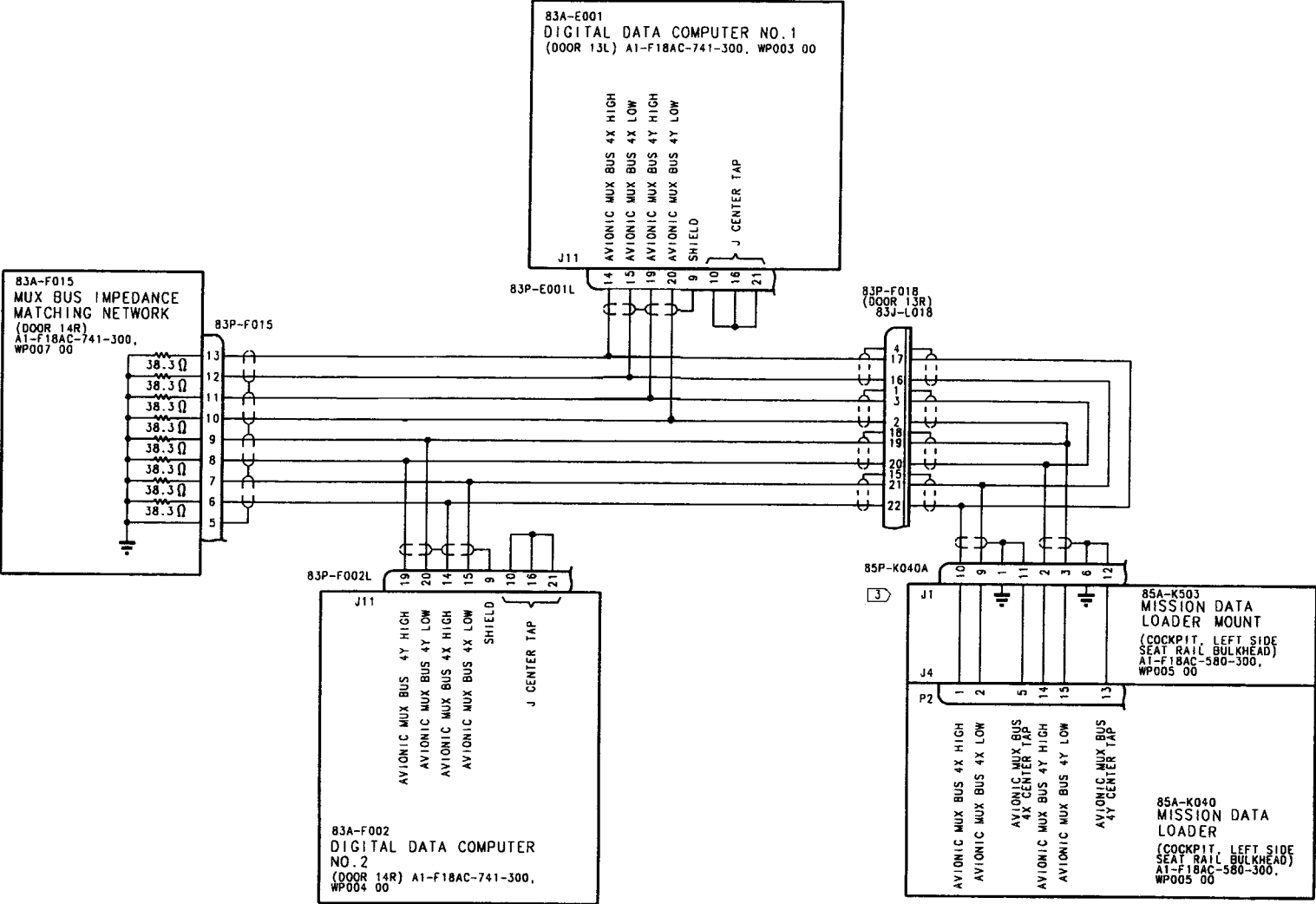


Figure 1.

Figure 1. Avionic Mux Channel 4 Schematic (Sheet 1)

Figure 1.

LEGEND**1. CONTINUITY TESTS:**

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

3 F/A-18A AND F/A-18B AFTER F/A-18 AFC 225 AND F/A-18 AFC 231.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - AVIONIC MUX CHANNEL 4****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292**

Reference Material

None

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Avionic Mux Channel 4 Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	U.S. Naval Reserves F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	U.S. Marine Corps Reserves F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jan 01	-

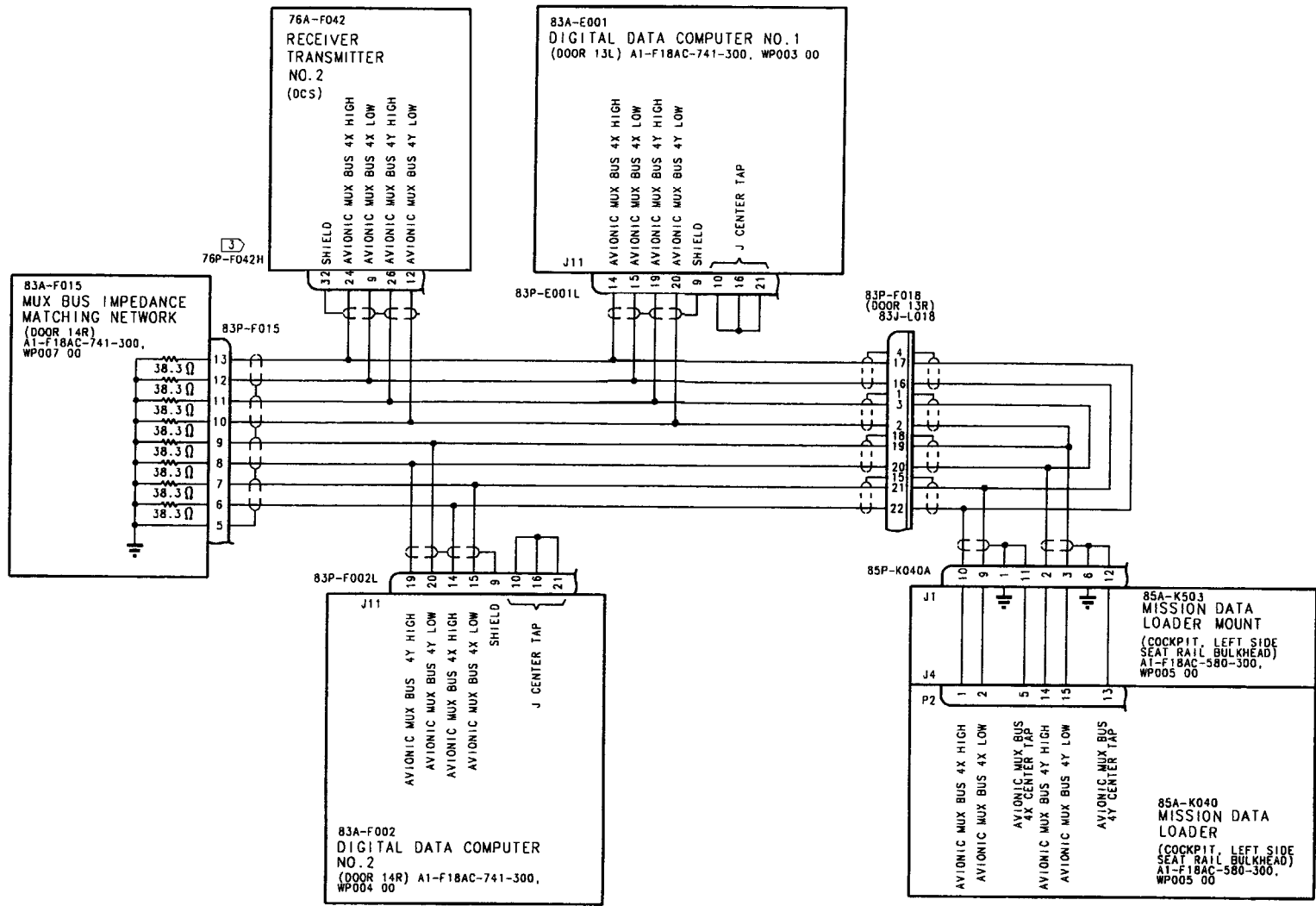


Figure 1.

Figure 1. Avionic Mux Channel 4 Schematic (Sheet 1)

Figure 1.

LEGEND**1. CONTINUITY TESTS:**

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

3  CONNECTOR STOWED WITH VHF/UHF COMM2 INSTALLED.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - AVIONIC MUX CHANNEL 5

MISSION COMPUTER SYSTEM

This WP supersedes WP018 00, dated 1 January 2001.

Title	Work Package
Schematic - Avionic MUX Channel 5 (AFTER F/A-18 AFC 225)	018 01
Schematic - Avionic MUX Channel 5 (AFTER F/A-18 AFC 253 OR F/A-18 AFC 292)	018 02

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - AVIONIC MUX CHANNEL 5****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AND F/A-18B AFTER F/A-18 AFC 225**

Reference Material

None

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Avionic Mux Channel 5 Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 225	-	Five (5) Avionics Multiplex Bus Upgrade, Incorporation of (ECP MDA-F/A-18 0529)	1 Jun 02	-

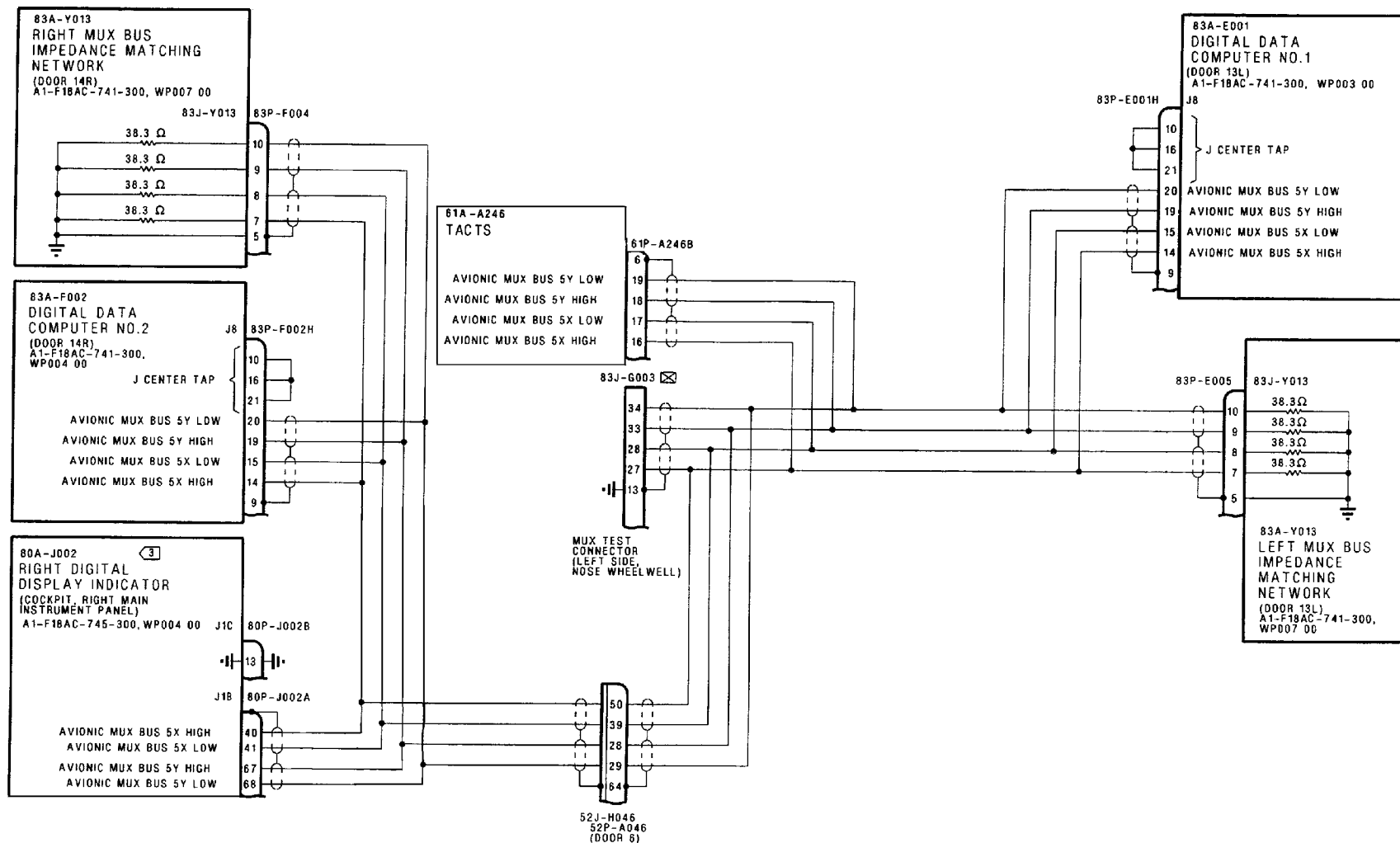


Figure 1.

Figure 1. Avionic Mux Channel 5 Schematic (Sheet 1)

Figure 1.

LEGEND**1. CONTINUITY TESTS:**

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

3 AFTER F/A-18 AFC 225.

ORGANIZATIONAL MAINTENANCE**SYSTEM SCHEMATICS****SCHEMATIC - AVIONIC MUX CHANNEL 5****MISSION COMPUTER SYSTEM****EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292**

Reference Material

None

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Avionic Mux Channel 5 Schematic, Figure 1 2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	U.S. Naval Reserves F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	U.S. Marine Corps Reserves F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jan 01	-

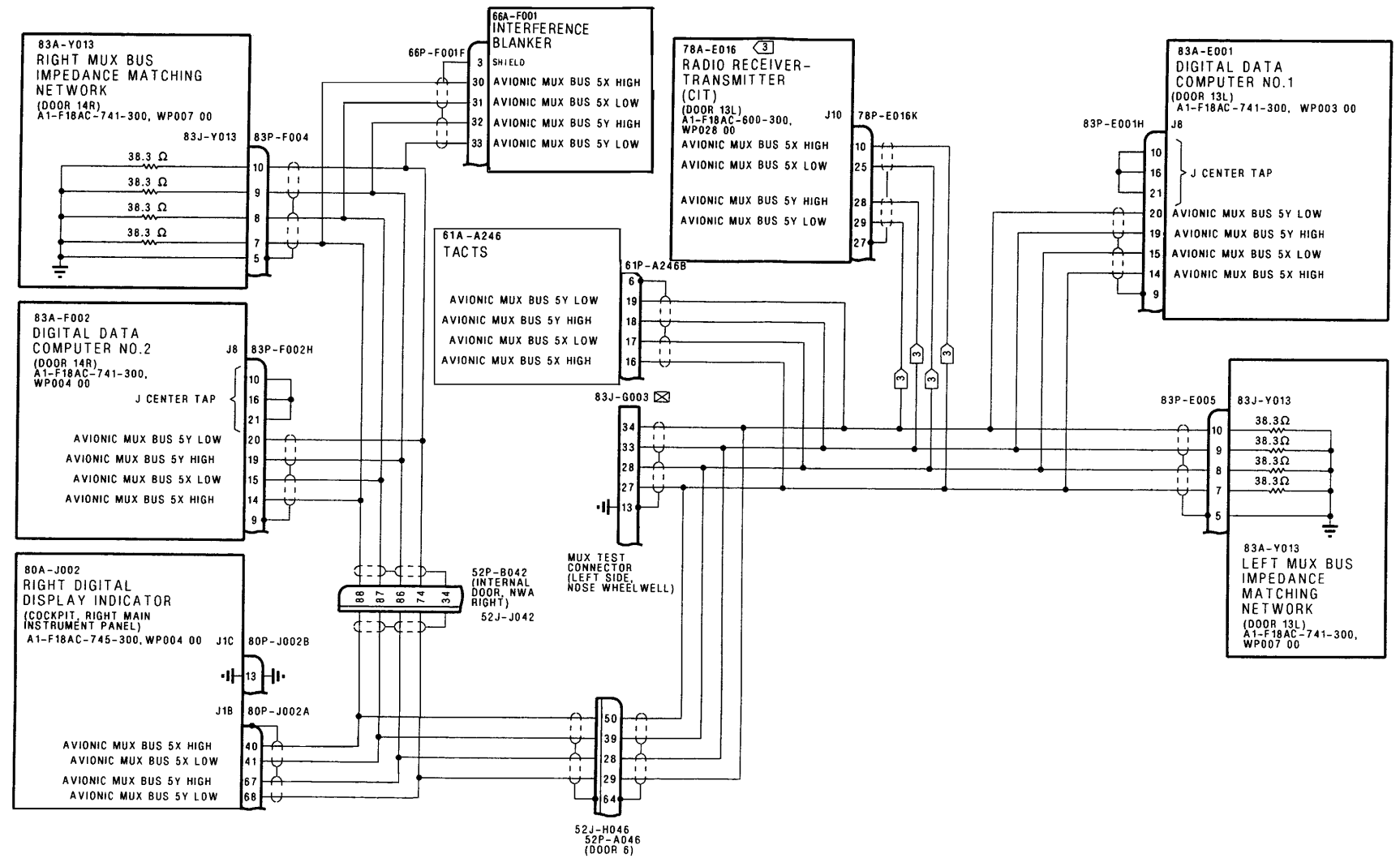


Figure 1.

Figure 1. Avionic Mux Channel 5 Schematic (Sheet 1)

Figure 1.

LEGEND**1. CONTINUITY TESTS:**

- A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
- B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
- C. DO NOT TEST LOW LEVEL DEVICES (SWITCHES/RELAY CONTACTS) FOR CONTINUITY WITH MULTIMETER ON RX 1 SCALE. PIN TO PIN TESTS THAT DO NOT GO THROUGH SWITCHES/RELAY CONTACTS MAY USE THE RX 1 SCALE.
- D. WHEN TESTING CONTINUITY, TEST FOR:
 - (1) SHORTS TO GROUND.
 - (2) SHORTS BETWEEN SURROUNDING PINS ON CONNECTORS.
 - (3) SHORTS BETWEEN SHIELD AND CONDUCTORS.
 - (4) SHIELD CONTINUITY.
- E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.

2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

3 AFTER F/A-18 AFC 292.

ORGANIZATIONAL MAINTENANCE

SYSTEM SCHEMATICS

SCHEMATIC - AVIONIC MUX CHANNEL 6

MISSION COMPUTER SYSTEM

EFFECTIVITY: F/A-18A AFTER F/A-18 AFC 253 OR F/A-18 AFC 292

Reference Material

None

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Avionic Mux Channel 6 Schematic, Figure 1	2

Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 253	-	USNR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0560R1)	1 Jan 01	-
F/A-18 AFC 292	-	USMCR F/A-18 A+ Avionics Upgrade Incorporation of, (ECP MDA-F/A-18-0583)	1 Jan 01	-

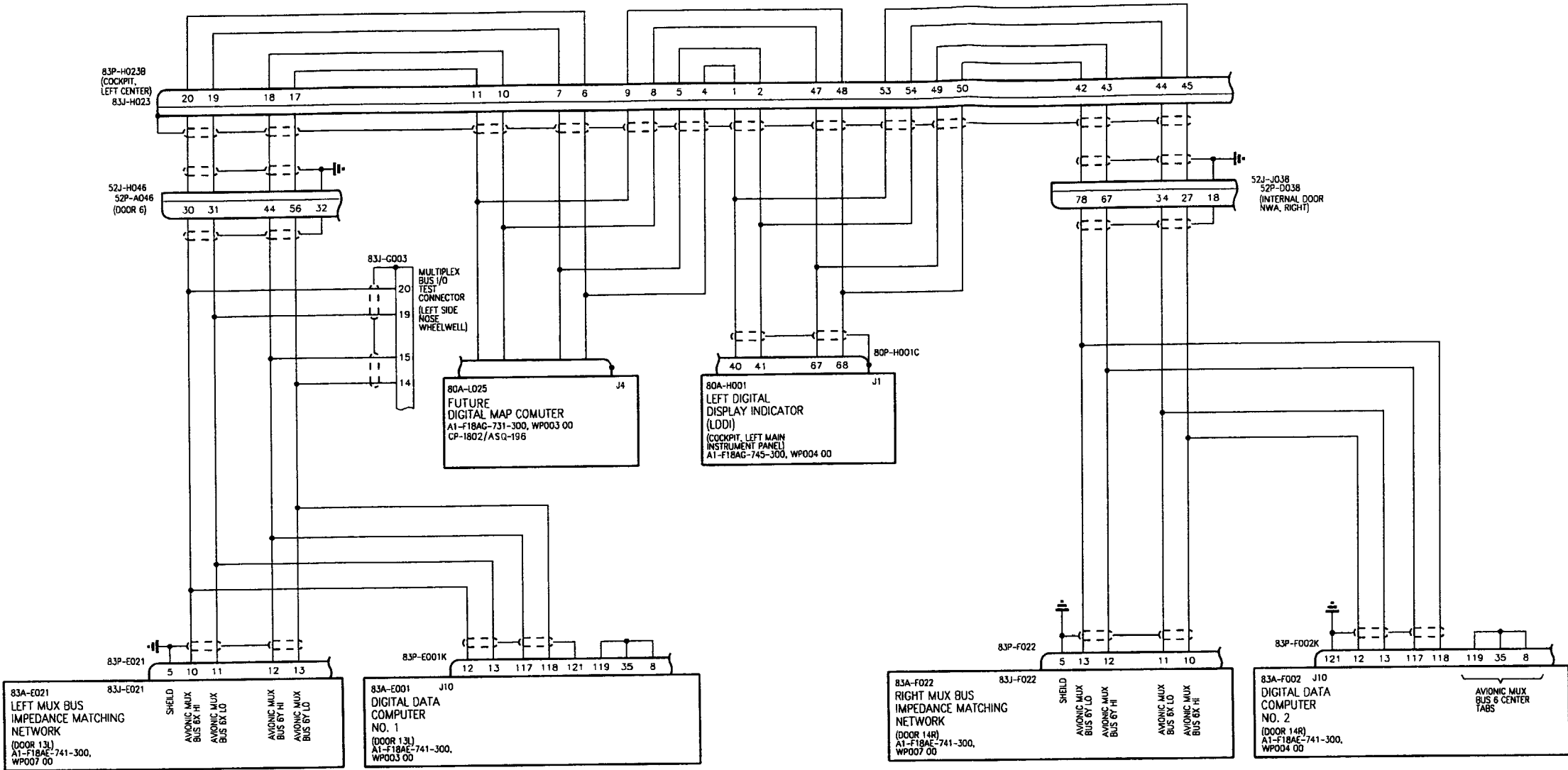


Figure 1.

Figure 1. Avionic Mux Channel 6 Schematic (Sheet 1)

Figure 1.

LEGEND

1. CONTINUITY TESTS:
 - A. ALL AIRCRAFT WIRE NUMBERS, SPLICE POINTS, AND GROUND POINTS ARE SHOWN IN A1-F18A()-WDM-000.
 - B. WHEN A LOW LEVEL CURRENT SWITCHING RELAY (IDENTIFIED BY \oplus) IS REMOVED FOR TROUBLESHOOTING, IDENTIFY RELAY AND SOCKET FOR CORRECT REINSTALLATION. DO NOT REPLACE LOW LEVEL CURRENT SWITCHING RELAY WITH ANY OTHER USED RELAY. IF RELAY IS DEFECTIVE, REPLACE WITH NEW RELAY.
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 - E. WHEN ELECTRICAL POWER IS OFF, 24VDC BATTERY VOLTAGE EXISTS ON SOME PINS ON CONNECTORS (IDENTIFIED BY \boxtimes). MAKE SURE MULTIMETER LEADS/JUMPER WIRES ARE INSTALLED ON CORRECT PINS WHEN TESTING FOR CONTINUITY.
2. FOR NONSTANDARD SYMBOLS AND ABBREVIATIONS, REFER TO WP002 01.

